NASA MEMORANDUM

TABULATED DATA FROM A PRESSURE-DISTRIBUTION INVESTIGATION AT MACH NUMBER 2.01 OF A 45° SWEPTBACK-WING AIRPLANE MODEL AT COMBINED ANGLES OF ATTACK AND SIDESLIP

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Langley Field, Va.

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TABULATED DATA FROM A PRESSURE-DISTRIBUTION INVESTIGATION AT MACH NUMBER 2.01 OF A 45° SWEPTBACK-WING AIRPLANE MODEL AT COMBINED ANGLES OF ATTACK AND SIDESLIP*

By John P. Gapcynski and Emma Jean Landrum

SUMMARY

A pressure-distribution investigation of a wing-body combination has been conducted in the Langley 4- by 4-foot supersonic pressure tunnel at a Mach number of 2.01. The model configuration consisted of an ogive-circular-cylinder body (fineness ratio of approximately 11) and a wing with 45° of sweepback at the quarter-chord line, an aspect ratio of 4, and a taper ratio of 0.2. Data were obtained on high-, mid-, and low-wing configurations and for the body and wing alone for a range of angles of attack and yaw from 0° to 15°. The tabulated pressure coefficients are presented in this report.

INTRODUCTION

As part of a general research program to determine the factors affecting the stability of supersonic aircraft at combined angles of attack and yaw, a pressure-distribution investigation of a wing-body combination has been conducted in the Langley 4- by 4-foot supersonic pressure tunnel. In addition to providing detailed pressure information to supplement the force-test results of a similar configuration (ref. 1), the results are directly applicable to the study of wing-body interference at combined angles of attack and yaw. The purpose of this report is to present the basic pressure information in tabulated form.

The model configuration consisted of an ogive-circular-cylinder body (fineness ratio of approximately 11) and a wing with 45° of sweepback at the quarter-chord line, an aspect ratio of 4, and a taper ratio

*Title, Unclassified.

of 0.2. Provision was made for changing the vertical position of the wing with respect to the body center line during the tests so that data could be obtained corresponding to high-, mid-, and low-wing configurations.

Extensive pressure data were obtained at a Mach number of 2.01 on the wing-body combination for an angle-of-attack range of 0° to 15° and angle-of-sideslip range of 0° to -15°. Wing-alone and body-alone data were also obtained for these angle ranges.

SYMBOLS

V	air speed
ρ	mass density of air
q	dynamic pressure, $\frac{1}{2}\rho V^2$
p	free-stream static pressure
p_l	local static pressure
C_p	pressure coefficient, $\frac{p_l - p}{q}$
α	angle of attack (see fig. 1), deg
β	angle of sideslip (see fig. 1), deg
θ	body polar angle, deg (see fig. 2)
x	distance from wing leading edge in chordwise direction
c	wing chord

DESCRIPTION OF MODELS AND TESTS

The details of the model configuration are shown in figure 2, and the geometric characteristics of the model are given in table I. The fuselage was an ogive-circular-cylinder configuration with a fineness ratio of approximately 11. At each of the 10 fuselage stations, shown in figure 2, there were 16 pressure orifices located $22\frac{1}{2}^\circ$ apart for a

range of 0° to 360°. The wing had 45° of sweepback at the quarter-chord line, an aspect ratio of 4, a taper ratio of 0.2, and NACA 65A004 sections in the stream direction. Provision was made for mounting the wing at two different vertical positions on the fuselage; the first position was on the fuselage center line, and the second position, 1.41 inches off the center line. On one surface of the wing, there were 144 orifices arranged symmetrically in 10 stations shown in figure 2. The chordwise distribution of orifices at each station may be seen in the data tabulation.

Pressure-coefficient data were obtained on the wing-body combination and on the body and wing alone for a range of angles of attack and sideslip. Wing-alone data were obtained by mounting the wing on a separate sting as shown in figure 2. During the tests, the model angle of attack was varied from -15° to 15° in $2\frac{1}{2}^\circ$ increments (for the midwing configuration a 5° increment was used), and the angle-of-sideslip range varied from 0° to -15° in 5° increments. Use of this angle-of-attack range in combination with the two wing positions discussed previously made it possible to obtain data for high-, mid-, and low-wing configurations for an actual angle-of-attack range from 0° to 15°. The Reynolds number based on the wing mean geometric chord was 1.98×10^6. The test procedure consisted of setting the desired angle of sideslip and obtaining data through the angle-of-attack range. In order to obtain complete wing-pressure coverage for an unsymmetrical configuration, it was necessary to repeat the tests with the wing in an inverted position.

Tunnel stagnation conditions were as follows: temperature, 100° F; dewpoint, approximately -35° F; and pressure, 14 lb/sq in. abs.

PRESENTATION OF RESULTS

The measured pressures on the model are presented in coefficient form in tables II to VI for the wing-body combination, midwing configuration; wing-body combination, high-wing configuration; wing-body combination, low-wing configuration; body alone; and wing alone, respectively.

The pressure coefficients are believed to be accurate within ±0.01. Where orifices were known to give erroneous results, the data were not tabulated. In addition, it should be noted that the pressures on the first three body stations could not be recorded during tests of the wing-body combination. Therefore, it must be assumed that these pressures

are identical to the values existing at these stations for the body-alone tests.

Langley Research Center,
National Aeronautics and Space Administration,
Langley Field, Va., August 29, 1958.

REFERENCE

1. Spearman, M. Leroy, Driver, Cornelius, and Hughes, William C.: Investigation of Aerodynamic Characteristics in Pitch and Sideslip of a 45° Sweptback-Wing Airplane Model With Various Vertical Locations of Wing and Horizontal Tail - Basic-Data Presentation, M = 2.01. NACA RM L54L06, 1955.

TABLE I

GEOMETRIC CHARACTERISTICS OF MODEL

Wing:	
Area, sq in.	144
Span, in.	24.00
Root chord, in.	10.00
Tip chord, in.	2.00
Taper ratio	0.2
Aspect ratio	4
Mean geometric chord, in.	6.89
Spanwise location of mean geometric chord, percent wing semispan	38.9
Sweep of quarter-chord line, deg	45
Airfoil section	NACA 65A004
Incidence, deg	0
Body:	
Length, in.	36.50
Diameter (max.), in.	3.33
Fineness ratio	10.96

TABLE 2

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.036	-.026	-.001	.008	-.019	-.012	-.001	.0
22.5				-.044	-.020	-.001	-.007	-.026	-.012	-.005	22.5
45.0				-.028	-.015	-.001		-.012	-.012	-.011	45.0
67.5				-.046	-.008	-.018	-.042		-.011	-.002	67.5
90.0				-.046				-.002	-.011	-.001	90.0
112.5				-.046	-.008	.000	-.021	-.002	-.009	.006	112.5
135.0				-.043	-.016	.012	-.005	-.011	-.008	-.002	135.0
157.5				-.040	-.016	.012	.004	-.020		-.002	157.5
180.0				-.039	-.016	.005	.008	-.020	-.007		180.0
202.5				-.039	-.016	.000	.009	-.020		-.009	202.5
225.0				-.037	-.016	.009	-.007	-.007	-.005	-.009	225.0
247.5				-.042	-.016	-.001	-.021	-.002	-.004	-.004	247.5
270.0				-.042				.000	-.001		270.0
292.5				-.043	-.009	-.014	-.033	.001	-.005	-.019	292.5
315.0				-.042	-.015	.002	-.016	-.008	-.007	-.018	315.0
337.5				-.040	-.015	.007	-.011	-.023		-.005	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.146	.139	.129	.040	.043	.131	.144	.159		.025
.075		.127	.107	.073	.027	.042	.087	.106	.126	.112	.075
.125	.119	.090	.077	.044	.018	.025	.055	.081	.100	.092	.125
.175	.100	.069	.054	.027	.018	.025	.024	.049	.075	.079	.175
.225	.076	.051	.030	-.001	.004	.008	.006	.032	.058	.069	.225
.275	.067	.051	.018	-.015		.008	-.004	.023	.045	.050	.275
.325	.038	.029	.005		-.004	.001	-.004	.001	.035	.050	.325
.375	.038	.011	-.004	-.025	-.004	.001	-.015	.001	.029	.038	.375
.425	.019	.001	-.004	-.023	-.004	-.004	-.018	-.011	.016		.425
.475	.019	-.006	-.020	-.032	-.015	-.011	-.023	-.025	.008	.014	.475
.550	.001	-.019		-.046	-.024	-.019	-.042	-.036	-.001	.003	.550
.650	-.018	-.043	-.059	-.058		-.027	-.048	-.068	-.034	-.018	.650
.750	-.026	-.049	-.065	-.061	-.035	-.027	-.058		-.040	-.030	.750
.800	-.035									-.035	.300
.850		-.054	-.075	-.061	-.035	-.024	-.070	-.049	-.052		.850
.900			-.067	-.061			-.056	-.054			.900
.950					-.035	-.024					.950
LOWER SURFACE											

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 2.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.032	-.014	.004	.040	.006	.000	.001	.0
22.5				-.032	-.014	.036	.028	.007	.000	-.002	22.5
45.0				-.018	-.015	.036		.006	-.004	-.007	45.0
67.5				-.039	-.007	.026	.000		-.009	-.001	67.5
90.0				-.043				-.001	-.009	-.007	90.0
112.5				-.048	-.013	-.040	-.056	-.002	-.016	.006	112.5
135.0				-.048	-.018	-.021	-.037	-.018	-.016	-.004	135.0
157.5				-.047	-.018	-.008	-.033	-.042		-.004	157.5
180.0				-.047	-.018	-.008	-.028	-.046	-.015		180.0
202.5				-.047	-.018	-.008	-.028	-.040		-.012	202.5
225.0				-.046	-.028	-.028	-.042	-.030	-.015	-.007	225.0
247.5				-.046	-.016	-.043	-.056	-.006	-.015	-.007	247.5
270.0				-.043				.000	-.015		270.0
292.5				-.043	-.013	.030	.002	.000	-.005	-.018	292.5
315.0				-.037	-.013	.039	.018	.000	.001	-.007	315.0
337.5				-.033	-.013	.030	.037	.000		-.007	337.5

x/C	C_p AT WING STATION										x/C
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.002	-.004	.018	-.069	-.055	.020	.027	.051		.025
.075		.014	.000	-.014	-.045	-.040	-.005	.014	.032	.020	.075
.125	.019	.000	-.014	-.036	-.045	-.040	-.025	.002	.014	.008	.125
.175	.007	-.015	-.027	-.046	-.045	-.040	-.051	-.024	-.006	-.004	.175
.225	-.007	-.035	-.045	-.075	-.057	-.043	-.068	-.037	-.019	-.010	.225
.275	-.012	-.025	-.054	-.088		-.043	-.080	-.048	-.027	-.023	.275
.325	-.035	-.048	-.065		-.057	-.043	-.079	-.059	-.034	-.023	.325
.375	-.035	-.057	-.076	-.089	-.057	-.043	-.083	-.065	-.039	-.033	.375
.425	-.050	-.064	-.076	-.083	-.059	-.043	-.083	-.075	-.049		.425
.475	-.051	-.074	-.090	-.093	-.059	-.054	-.082	-.087	-.057	-.052	.475
.550	-.062	-.084		-.099	-.069	-.054	-.095	-.096	-.064	-.061	.550
.650	-.082	-.107	-.117	-.109		-.061	-.095	-.115	-.092	-.078	.650
.750	-.083	-.107	-.126	-.104	-.067	-.061	-.095		-.089	-.083	.750
.800	-.089									-.081	.800
.850		-.107	-.126	-.099	-.067	-.061	-.095	-.103	-.097		.850
.900			-.117	-.099			-.090	-.098			.900
.950					-.067	-.056					.950
LOWER SURFACE											
.025		.257	.246	.230	.107	.130	.231	.250	.267		.025
.075		.207	.177	.151	.089	.102	.157	.193	.216	.204	.075
.125	.186	.157	.134	.115	.082	.076	.125	.150	.179	.181	.125
.175	.158	.127	.114	.092	.073	.076	.090	.118	.148	.159	.175
.225	.133	.107	.089	.071	.059	.067	.067	.098	.130	.149	.225
.275	.124	.100	.071	.054		.058	.055	.080	.111	.126	.275
.325	.105	.076	.055		.043	.054	.046	.062	.097	.116	.325
.375	.099	.064	.042	.032	.033	.049	.036	.052	.088	.104	.375
.425	.074	.052	.042	.035	.033	.040	.027	.036	.074		.425
.475	.071	.039	.025	.019	.013	.031	.021	.023	.064	.078	.475
.550	.051	.018		.006	.007	.019	-.002	.012	.049	.061	.550
.650	.031	-.002	-.008	-.014		.010	-.006	-.017	.019	.038	.650
.750	.013	-.008	-.018	-.021	-.011	.007	-.019		.010	.023	.750
.800	.007									.015	.800
.850		-.019	-.025	-.025	-.011	.004	-.027	-.004	-.004		.850
.900			-.025	-.025			-.024	-.007			.900
.950					-.011	.004					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 5.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.023	-.009	.006	.076	.047	.012	.002	.0
22.5				-.023	-.009	.046	.064	.040	.007	-.009	22.5
45.0				-.021	-.021	.076		.020	-.004	-.016	45.0
67.5				-.044	-.030	.066	.028		-.015	-.026	67.5
90.0				-.054				-.006	-.021	-.030	90.0
112.5				-.064	-.037	-.075	-.077	-.016	-.034	-.014	112.5
135.0				-.064	-.025	-.048	-.066	-.032	-.028	-.011	135.0
157.5				-.044	-.018	-.030	-.064	-.060		-.012	157.5
180.0				-.053	-.016	-.020	-.063	-.075	-.025		180.0
202.5				-.050	-.016	-.034	-.063	-.066		-.009	202.5
225.0				-.055	-.026	-.054	-.077	-.041	-.029	-.016	225.0
247.5				-.056	-.039	-.075	-.084	-.022	-.026	-.027	247.5
270.0				-.053				-.009	-.021		270.0
292.5				-.047	-.039	.068	.039	.002	-.012	-.028	292.5
315.0				-.035	-.016	.077	.047	.019	-.007	-.018	315.0
337.5				-.023	-.015	.054	.063	.035		-.007	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.100	-.096	-.048	-.138	-.136	-.071	-.063	-.044		.025
.075		-.067	-.080	-.082	-.119	-.112	-.083	-.068	-.049	-.055	.075
.125	-.057	-.077	-.086	-.098	-.093	-.106	-.099	-.074	-.058	-.062	.125
.175	-.069	-.084	-.094	-.105	-.083	-.093	-.117	-.094	-.068	-.068	.175
.225	-.069	-.096	-.106	-.120	-.092	-.100	-.130	-.102	-.076	-.071	.225
.275	-.079	-.086	-.113	-.138		-.093	-.139	-.104	-.086	-.081	.275
.325	-.092	-.105	-.124		-.090	-.093	-.136	-.115	-.092	-.078	.325
.375	-.092	-.111	-.124	-.137	-.090	-.093	-.148	-.115	-.089	-.085	.375
.425	-.112	-.120	-.124	-.127	-.088	-.093	-.132	-.121	-.100		.425
.475	-.106	-.125	-.134	-.138	-.088	-.093	-.132	-.134	-.106	-.100	.475
.550	-.115	-.132		-.138	-.088	-.093	-.145	-.138	-.110	-.109	.550
.650	-.125	-.149	-.157	-.138		-.093	-.145	-.158	-.132	-.121	.650
.750	-.115	-.152	-.151	-.138	-.088	-.093	-.145		-.137	-.106	.750
.800	-.113									-.097	.800
.850		-.145	-.149	-.138	-.088	-.088	-.145	-.124	-.132		.850
.900			-.138	-.138			-.137	-.123			.900
.950					-.083	-.077					.950
LOWER SURFACE											
.025		.356	.340	.317	.199	.230	.325	.342	.361		.025
.075		.289	.258	.232	.173	.171	.234	.263	.299	.287	.075
.125	.270	.240	.203	.190	.149	.143	.195	.219	.257	.258	.125
.175	.239	.208	.182	.162	.133	.143	.155	.182	.224	.233	.175
.225	.211	.182	.149	.131	.115	.120	.126	.159	.200	.221	.225
.275	.198	.169	.132	.117		.109	.108	.142	.180	.196	.275
.325	.177	.139	.114		.105	.106	.102	.119	.161	.185	.325
.375	.165	.130	.105	.090	.096	.099	.090	.111	.148	.171	.375
.425	.145	.117	.096	.080	.088	.086	.081	.093	.132		.425
.475	.132	.102	.083	.067	.076	.077	.069	.076	.122	.137	.475
.550	.109	.081		.052	.058	.058	.044	.058	.106	.122	.550
.650	.090	.056	.038	.021		.046	.039	.027	.075	.097	.650
.750	.071	.043	.020	.021	.038	.040	.018		.063	.082	.750
.800	.065									.073	.800
.850		.037	.020	.021	.031	.040	.015	.045	.049		.850
.900			.020	.021			.015	.041			.900
.950					.031	.031					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 7.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.006	.009	.008	.124	.085	.037	.009	.0
22.5				-.006	.005	.073	.107	.077	.026	.004	22.5
45.0				-.005	-.022	.112		.042	.002	-.016	45.0
67.5				-.059	-.048	.101	.081		-.020	-.032	67.5
90.0				-.077				-.005	-.033	-.042	90.0
112.5				-.083	-.054	-.097	-.101	-.021	-.044	-.021	112.5
135.0				-.083	-.032	-.080	-.090	-.030	-.032	-.011	135.0
157.5				-.057	-.026	-.054	-.078	-.068		-.006	157.5
180.0				-.056	-.019	-.034	-.070	-.087	-.025		180.0
202.5				-.056	-.022	-.054	-.078	-.062		-.006	202.5
225.0				-.066	-.035	-.078	-.103	-.041	-.037	-.016	225.0
247.5				-.084	-.060	-.096	-.105	-.035	-.044	-.034	247.5
270.0				-.070				-.009	-.027		270.0
292.5				-.059	-.062	.118	.082	.015	-.012	-.037	292.5
315.0				-.033	-.023	.118	.100	.040	.011	-.007	315.0
337.5				-.001	.002	.087	.109	.070		.001	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.163	-.162	-.108	-.193	-.194	-.150	-.143	-.128		.025
.075		-.134	-.143	-.143	-.188	-.183	-.150	-.137	-.117	-.127	.075
.125	-.121	-.136	-.148	-.150	-.155	-.171	-.156	-.137	-.121	-.127	.125
.175	-.121	-.140	-.153	-.150	-.127	-.133	-.171	-.146	-.126	-.127	.175
.225	-.121	-.146	-.153	-.164	-.138	-.136	-.184	-.153	-.132	-.127	.225
.275	-.129	-.136	-.159	-.182		-.130	-.184	-.153	-.136	-.134	.275
.325	-.150	-.153	-.165		-.121	-.125	-.178	-.161	-.140	-.127	.325
.375	-.140	-.158	-.174	-.180	-.119	-.117	-.187	-.161	-.137	-.132	.375
.425	-.152	-.162	-.165	-.170	-.113	-.117	-.178	-.168	-.143		.425
.475	-.151	-.167	-.178	-.176	-.112	-.117	-.178	-.182	-.148	-.144	.475
.550	-.151	-.167		-.176	-.112	-.117	-.183	-.182	-.151	-.148	.550
.650	-.157	-.178	-.186	-.183		-.111	-.182	-.193	-.168	-.147	.650
.750	-.145	-.169	-.171	-.176	-.111	-.111	-.180		-.151	-.134	.750
.800	-.145									-.133	.800
.850		-.165	-.170	-.176	-.111	-.107	-.180	-.151	-.146		.850
.900			-.165	-.176			-.176	-.152			.900
.950					-.111	-.100					.950
LOWER SURFACE											
.025		.438	.424	.394	.284	.328	.419	.434	.440		.025
.075		.359	.327	.303	.233	.256	.328	.350	.373	.363	.075
.125	.340	.306	.278	.258	.205	.214	.274	.297	.328	.332	.125
.175	.309	.276	.246	.226	.193	.199	.233	.263	.294	.303	.175
.225	.278	.246	.217	.195	.174	.194	.205	.232	.267	.292	.225
.275	.268	.231	.198	.180		.186	.186	.209	.247	.267	.275
.325	.238	.202	.176		.162	.170	.168	.193	.226	.251	.325
.375	.221	.189	.164	.150	.145	.162	.158	.180	.210	.232	.375
.425	.202	.173	.149	.136	.140	.150	.145	.162	.194		.425
.475	.190	.163	.138	.124	.130	.142	.132	.150	.183	.199	.475
.550	.168	.129		.100	.104	.119	.107	.123	.161	.179	.550
.650	.142	.098	.087	.080		.093	.099	.093	.127	.154	.650
.750	.126	.096	.074	.064	.093	.087	.067		.115	.134	.750
.800	.119									.126	.800
.850		.083	.074	.064	.074	.087	.067	.097	.100		.850
.900			.074	.064			.067	.091			.900
.950					.074	.077					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 10.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.012	.032	.026	.163	.125	.062	.035	.0
22.5				.021	.019	.091	.146	.110	.047	.014	22.5
45.0				-.002	-.028	.139		.067	.011	-.021	45.0
67.5				-.069	-.080	.139	.126		-.030	-.062	67.5
90.0				-.115				-.020	-.048	-.074	90.0
112.5				-.119	-.099	-.128	-.146	-.050	-.060	-.040	112.5
135.0				-.094	-.050	-.113	-.137	-.039	-.037	-.018	135.0
157.5				-.062	-.040	-.097	-.105	-.074		-.013	157.5
180.0				-.061	-.019	-.044	-.091	-.116	-.018		180.0
202.5				-.067	-.040	-.098	-.106	-.083		-.014	202.5
225.0				-.088	-.055	-.126	-.153	-.047	-.048	-.029	225.0
247.5				-.119	-.096	-.130	-.143	-.048	-.064	-.054	247.5
270.0				-.108				-.022	-.050		270.0
292.5				-.066	-.081	.135	.128	.015	-.022	-.071	292.5
315.0				-.027	-.025	.146	.129	.064	.006	-.023	315.0
337.5				.013	.012	.108	.139	.106		.001	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.231	-.227	-.168	-.252	-.262	-.218	-.209	-.192		.025
.075		-.193	-.199	-.202	-.250	-.240	-.214	-.199	-.179	-.188	.075
.125	-.184	-.193	-.200	-.202	-.241	-.231	-.213	-.190	-.176	-.186	.125
.175	-.183	-.194	-.202	-.202	-.197	-.205	-.213	-.197	-.178	-.184	.175
.225	-.183	-.201	-.209	-.215	-.184	-.194	-.225	-.203	-.180	-.183	.225
.275	-.183	-.185	-.209	-.224		-.152	-.225	-.203	-.181	-.184	.275
.325	-.201	-.206	-.216		-.145	-.142	-.216	-.219	-.181	-.176	.325
.375	-.187	-.205	-.215	-.219	-.134	-.132	-.222	-.213	-.179	-.177	.375
.425	-.200	-.205	-.212	-.209	-.128	-.132	-.214	-.226	-.187		.425
.475	-.188	-.206	-.222	-.216	-.128	-.132	-.214	-.226	-.190	-.186	.475
.550	-.185	-.210		-.216	-.128	-.132	-.221	-.226	-.187	-.173	.550
.650	-.183	-.201	-.209	-.216		-.125	-.221	-.226	-.183	-.170	.650
.750	-.177	-.197	-.202	-.213	-.140	-.125	-.218		-.173	-.167	.750
.800	-.181									-.166	.800
.850		-.196	-.202	-.209	-.128	-.117	-.216	-.182	-.175		.850
.900			-.201	-.216			-.212	-.186			.900
.950					-.126	-.114					.950
LOWER SURFACE											
.025		.509	.496	.463	.373	.400	.500	.505	.513		.025
.075		.438	.393	.370	.289	.320	.396	.417	.445	.432	.075
.125	.415	.387	.351	.323	.261	.271	.340	.369	.400	.403	.125
.175	.383	.350	.317	.287	.255	.259	.293	.327	.361	.373	.175
.225	.348	.321	.289	.258	.240	.254	.264	.292	.334	.358	.225
.275	.333	.293	.262	.238		.243	.245	.275	.309	.330	.275
.325	.309	.268	.243		.219	.233	.232	.256	.289	.310	.325
.375	.289	.257	.225	.205	.205	.218	.215	.237	.270	.295	.375
.425	.269	.238	.209	.194	.192	.204	.193	.219	.256		.425
.475	.254	.220	.198	.180	.184	.188	.193	.201	.242	.260	.475
.550	.226	.187		.154	.152	.170	.158	.175	.217	.237	.550
.650	.195	.170	.139	.123		.146	.145	.146	.178	.213	.650
.750	.186	.157	.123	.111	.148	.131	.115		.162	.195	.750
.800	.179									.185	.800
.850		.148	.118	.107	.120	.130	.114	.145	.152		.850
.900			.115	.107			.114	.134			.900
.950					.108	.123					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 12.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.047	.060	.059	.206	.167	.088	.049	.0
22.5				.047	.042	.098	.191	.143	.063	.032	22.5
45.0				.007	-.018	.178		.085	.016	-.015	45.0
67.5				-.070	-.085	.142	.176		-.043	-.081	67.5
90.0				-.136				-.009	-.069	-.125	90.0
112.5				-.155	-.118	-.153	-.164	-.084	-.076	-.060	112.5
135.0				-.109	-.067	-.138	-.174	-.018	-.049	-.027	135.0
157.5				-.075	-.067	-.138	-.137	-.098		-.018	157.5
180.0				-.056	-.009	-.043	-.114	-.135	-.023		180.0
202.5				-.073	-.075	-.135	-.114	-.095		-.019	202.5
225.0				-.098	-.082	-.149	-.192	-.041	-.057	-.035	225.0
247.5				-.155	-.125	-.158	-.160	-.082	-.076	-.064	247.5
270.0				-.136				-.018	-.076		270.0
292.5				-.078	-.090	.173	.171	.032	-.043	-.102	292.5
315.0				-.014	-.019	.189	.171	.088	.016	-.006	315.0
337.5				.035	.036	.133	.196	.142		.020	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.268	-.261	-.187	-.278	-.281	-.252	-.251	-.233		.025
.075		-.233	-.234	-.236	-.281	-.281	-.253	-.237	-.219	-.227	.075
.125	-.219	-.234	-.234	-.236	-.281	-.281	-.253	-.225	-.213	-.225	.125
.175	-.220	-.234	-.234	-.232	-.256	-.252	-.256	-.233	-.213	-.222	.175
.225	-.220	-.234	-.234	-.237	-.244	-.258	-.256	-.242	-.213	-.219	.225
.275	-.214	-.214	-.234	-.244		-.212	-.256	-.234	-.214	-.218	.275
.325	-.212	-.231	-.234		-.177	-.178	-.234	-.249	-.213	-.195	.325
.375	-.202	-.231	-.234	-.234	-.156	-.163	-.243	-.242	-.209	-.191	.375
.425	-.202	-.230	-.234	-.234	-.146	-.163	-.242	-.253	-.213		.425
.475	-.202	-.221	-.246	-.244	-.146	-.153	-.240	-.245	-.209	-.192	.475
.550	-.202	-.214		-.243	-.146	-.149	-.246	-.245	-.199	-.186	.550
.650	-.202	-.214	-.228	-.239		-.139	-.246	-.245	-.196	-.186	.650
.750	-.202	-.214	-.225	-.239	-.144	-.139	-.238		-.193	-.185	.750
.800	-.199									-.183	.800
.850		-.214	-.225	-.232	-.144	-.139	-.238	-.204	-.199		.850
.900			-.218	-.242			-.240	-.205			.900
.950					-.144	-.139					.950
LOWER SURFACE											
.025		.568	.555	.531	.429	.464	.557	.566	.579		.025
.075		.495	.461	.445	.326	.371	.456	.485	.511	.499	.075
.125	.467	.439	.405	.391	.304	.330	.406	.425	.466	.468	.125
.175	.445	.402	.377	.350	.304	.320	.351	.389	.432	.438	.175
.225	.405	.376	.348	.320	.288	.305	.327	.357	.400	.423	.225
.275	.391	.352	.320	.298		.300	.305	.335	.372	.395	.275
.325	.367	.324	.295		.275	.291	.287	.311	.352	.375	.325
.375	.345	.314	.285	.262	.260	.275	.275	.296	.333	.357	.375
.425	.326	.294	.268	.254	.252	.260	.254	.273	.317		.425
.475	.310	.279	.248	.239	.243	.249	.244	.258	.301	.319	.475
.550	.276	.243		.207	.217	.224	.206	.232	.276	.296	.550
.650	.255	.214	.186	.180		.193	.199	.189	.238	.271	.650
.750	.239	.201	.171	.164	.204	.176	.163		.222	.248	.750
.800	.238									.242	.800
.850		.191	.175	.164	.179	.168	.160	.194	.205		.850
.900			.175	.164			.160	.185			.900
.950					.164	.162					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 15.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.081	.101	.097	.266	.212	.123	.081	.0
22.5				.087	.076	.132	.241	.188	.094	.052	22.5
45.0				.032	.001	.226		.128	.027	-.016	45.0
67.5				-.071	-.078	.160	.197		-.068	-.118	67.5
90.0				-.152				-.026	-.126	-.174	90.0
112.5				-.193	-.135	-.188	-.195	-.131	-.101	-.069	112.5
135.0				-.116	-.123	-.195	-.229	-.020	-.067	-.060	135.0
157.5				-.078	-.091	-.167	-.129	-.082		-.041	157.5
180.0				-.050	.001	-.055	-.118	-.142	-.015		180.0
202.5				-.081	-.101	-.158	-.132	-.092		-.036	202.5
225.0				-.108	-.123	-.205	-.231	-.018	-.061	-.060	225.0
247.5				-.195	-.135	-.184	-.192	-.144	-.089	-.081	247.5
270.0				-.150				-.042	-.121		270.0
292.5				-.070	-.083	.179	.193	.025	-.068	-.135	292.5
315.0				.015	.002	.228	.220	.125	.027	-.012	315.0
337.5				.084	.068	.125	.243	.183		.041	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.275	-.275	-.206		-.294	-.281	-.272	-.258		.025
.075			-.250	-.258	-.296	-.294	-.275	-.263	-.248	-.230	.075
.125	-.232	-.246		-.258	-.296	-.284	-.268	-.249	-.241	-.230	.125
.175	-.225	-.246	-.249	-.251	-.278	-.270	-.268	-.256	-.241	-.229	.175
.225	-.224	-.246	-.249	-.251	-.253	-.265	-.265	-.256	-.238	-.227	.225
.275	-.222	-.246	-.250	-.251		-.218	-.262	-.256	-.234	-.227	.275
.325	-.221	-.244	-.250		-.189	-.197	-.251	-.269	-.227	-.212	.325
.375	-.218	-.247	-.249	-.253	-.174	-.182	-.259	-.263	-.217	-.211	.375
.425	-.218	-.244	-.241		-.157	-.176	-.253	-.258	-.219		.425
.475	-.215	-.244	-.241	-.251	-.157	-.168	-.250	-.257	-.218	-.215	.475
.550	-.215	-.234		-.251	-.156	-.158	-.256	-.249	-.211	-.210	.550
.650	-.215	-.234	-.250	-.251		-.151	-.255	-.249	-.216	-.213	.650
.750	-.207	-.246	-.243	-.251	-.146	-.151	-.255		-.212	-.212	.750
.800	-.214									-.210	.800
.850		-.243	-.236	-.251	-.162	-.161	-.253	-.216	-.221		.850
.900			-.222	-.263			-.251	-.211			.900
.950					-.178	-.167					.950
LOWER SURFACE											
.025		.635	.624	.592	.477	.547	.619	.623	.633		.025
.075		.567	.535	.502	.375	.429	.520	.551	.571	.557	.075
.125	.541	.522	.493	.455	.338	.385	.470	.497	.531	.528	.125
.175	.511	.488	.450	.408	.338	.362	.420	.444	.492	.497	.175
.225	.479	.460	.426	.385	.338	.357	.397	.437	.460	.483	.225
.275	.461	.424	.398	.364		.349	.371	.400	.432	.454	.275
.325	.438	.407	.375		.332	.342	.356	.379	.411	.433	.325
.375	.416	.398	.363	.330	.320	.330	.343	.361	.389	.415	.375
.425	.400	.375	.351	.316	.306	.318	.314	.337	.372		.425
.475	.379	.355	.329	.300	.301	.306	.305	.325	.357	.378	.475
.550	.350	.312		.267	.276	.283	.270	.296	.328	.349	.550
.650	.324	.294	.244	.236		.252	.261	.251	.284	.324	.650
.750	.317	.276	.232	.219	.273	.237	.219		.278	.308	.750
.800	.320									.298	.800
.850		.267	.232	.218	.237	.229	.216	.242	.261		.850
.900			.235	.221			.214	.235			.900
.950					.211	.218					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.059	-.065	-.040	-.033	-.062	-.058	-.054	.0
22.5				-.072	-.061	-.043	-.044	-.051	-.054	-.051	22.5
45.0				-.035	-.042	-.034	-.047	-.029	-.040	-.045	45.0
67.5				-.042	-.012	-.029	-.049		-.022	-.027	67.5
90.0				-.033				-.010	-.015	-.029	90.0
112.5				-.037	-.010	-.020	-.049	-.015	-.022	-.014	112.5
135.0				-.050	-.041	-.030	-.044	-.026	-.036	-.029	135.0
157.5				-.065	-.061	-.042	-.040	-.045		-.048	157.5
180.0				-.071	-.069	-.037	-.037	-.061	-.049		180.0
202.5				-.086	-.066	-.019	-.036	-.055	-.049	-.042	202.5
225.0				-.082	-.052	-.009	-.035	-.047	-.044	-.022	225.0
247.5				-.077	-.023	-.019	-.042	-.040	-.044	-.037	247.5
270.0				-.070				-.028	-.035		270.0
292.5				-.075	-.022	-.017	-.049	-.037	-.042	-.038	292.5
315.0				-.082	-.048	-.019	-.045	-.047	-.044	-.040	315.0
337.5				-.083	-.058	-.008	-.037	-.058	-.044	-.045	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.165	.153	.131	.025	.019	.058	.066	.084		.025
.075		.116	.107	.080	.025	.019	.033	.044	.057	.040	.075
.125	.101	.089	.069	.040	.007	.005	.006	.012	.035	.029	.125
.175	.078	.058	.044	.014	.000	-.007	-.014	-.008	.012	.014	.175
.225	.052	.044	.022	-.004	-.007	-.014	-.030	-.022	-.004	.006	.225
.275	.045	.019	.004	-.018	-.007	-.014	-.037	-.032	-.015	-.005	.275
.325	.025	.006	-.009	-.031	-.017	-.014	-.041	-.043	-.025	-.013	.325
.375	.008	-.008	-.022	-.038	-.022	-.025	-.044	-.051	-.034		.375
.425	-.004	-.011		-.049	-.025	-.025	-.050	-.057	-.045		.425
.475	-.012	-.025	-.041	-.056	-.028	-.030	-.051	-.065	-.053	-.048	.475
.550	-.028	-.045	-.063	-.066	-.038	-.038	-.067	-.076	-.065	-.059	.550
.650	-.049	-.064	-.088	-.090		-.054	-.069	-.090	-.081	-.071	.650
.750	-.064	-.077	-.095	-.095	-.057	-.054	-.082	-.083	-.083	-.084	.750
.800	-.064									-.086	.800
.850		-.082	-.095		-.056	-.054	-.082	-.080	-.096		.850
.900			-.099	-.095			-.075	-.081			.900
.950					-.056	-.054					.950
LOWER SURFACE											

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 5.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.016	-.015	-.010	.057	.036	.035		.0
22.5				.008	.013	.052	.051	.042		.008	22.5
45.0				.014	.017	.085	.044	.049	.024	.016	45.0
67.5				-.002	.039	.097	.049	.034	.009	.023	67.5
90.0				-.029				.009	-.003	-.014	90.0
112.5				-.064	-.054	-.102	-.079		-.025	-.027	112.5
135.0				-.067	-.071	-.081	-.080	-.046	-.046	-.039	135.0
157.5				-.087	-.056	-.053	-.075	-.074	-.045	-.029	157.5
180.0				-.066	-.032	-.041	-.068	-.086	-.030	-.013	180.0
202.5				-.051	-.014	-.032	-.074	-.079	-.028	-.020	202.5
225.0				-.049	-.012	-.041	-.078	-.046	-.025	-.022	225.0
247.5				-.060	-.014	-.071	-.092	-.005	-.017	-.012	247.5
270.0				-.080				.006	-.017		270.0
292.5				-.087	-.058	.043	.050	.005	-.017	-.025	292.5
315.0				-.076	-.075	.067	.057	.012	-.020	-.041	315.0
337.5				-.046	-.048	.028	.061	.016	-.015	-.031	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.039	-.038	-.032	-.120	-.101	-.126	-.123	-.120		.025
.075		-.019	-.035	-.048	-.106	-.062	-.123	-.114	-.110	-.126	.075
.125	-.012	-.035	-.046	-.066	-.106	-.064	-.123	-.102	-.110	-.126	.125
.175	-.020	-.049	-.064	-.073	-.090	-.058	-.132	-.119	-.115	-.126	.175
.225	-.037	-.059	-.074	-.101	-.101	-.070	-.132	-.133	-.123	-.123	.225
.275	-.045	-.047	-.086	-.119	-.092	-.070	-.128	-.123	-.128	-.131	.275
.325	-.065	-.077	-.095	-.128	-.099	-.070	-.110	-.143	-.128	-.118	.325
.375	-.066	-.085	-.106	-.139	-.099	-.070	-.121	-.126	-.123		.375
.425	-.085	-.095		-.131	-.090	-.077	-.113	-.138	-.131		.425
.475	-.075	-.104	-.118	-.148	-.101	-.077	-.113	-.154	-.136	-.140	.475
.550	-.087	-.104	-.118	-.163	-.101	-.077	-.123	-.147	-.136	-.137	.550
.650	-.104	-.133	-.134	-.161		-.081	-.123	-.161	-.156	-.153	.650
.750	-.104	-.137	-.145	-.144	-.105	-.081	-.119	-.137	-.148	-.138	.750
.800	-.104									-.130	.800
.850		-.137	-.145	-.139	-.094	-.081	-.108	-.126	-.140		.850
.900			-.130	-.148			-.100	-.126			.900
.950					-.090	-.081					.950
LOWER SURFACE											
.025		.433	.419	.379	.240	.170	.279	.287	.292		.025
.075		.357	.324	.294	.211	.146	.210	.224	.239	.229	.075
.125	.319	.300	.264	.236	.178	.133	.172	.186	.207	.206	.125
.175	.287	.258	.227	.195	.167	.133	.139	.158	.180	.186	.175
.225	.253	.230	.200	.170	.148	.114	.122	.135	.159	.177	.225
.275	.241	.212	.177	.147	.141	.114	.110	.124	.143	.153	.275
.325	.216	.183	.154	.132	.128	.110	.102	.106	.126	.143	.325
.375	.201	.171	.142	.118	.119	.101	.087	.097	.115		.375
.425	.178	.154		.111	.112	.094	.078	.087	.104		.425
.475	.167	.137	.118	.094	.104	.087	.066	.074	.093	.101	.475
.550	.137	.112	.093	.079	.084	.075	.049	.059	.078	.088	.550
.650	.114	.088	.064	.047		.054	.047	.033	.048	.071	.650
.750	.101	.073	.052	.037	.048	.047	.027	.034	.038	.056	.750
.800	.086									.051	.800
.850		.065	.040	.037	.048	.047	.027	.037	.028		.850
.900			.040	.037			.027	.034			.900
.950					.048	.047					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 10.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.006	.032	.024	.132	.102	.043	.014	.0
22.5				.059	.056	.140	.100	.109	.061	.036	22.5
45.0				.063	.045	.180	.131	.100	.049	.023	45.0
67.5				.013	.044	.211	.144		.016	-.009	67.5
90.0				-.049				.006	-.021	-.058	90.0
112.5				-.104	-.130	-.149	-.115	-.028	-.057	-.064	112.5
135.0				-.138	-.126	-.105	-.114	-.061	-.074	-.045	135.0
157.5				-.088	-.058	-.089	-.116	-.093		-.021	157.5
180.0				-.065	-.051	-.101	-.146	-.138	-.014		180.0
202.5				-.049	-.005	-.037	-.096	-.101	-.037	-.021	202.5
225.0				-.060	-.037	-.081	-.105	-.044	-.027	-.021	225.0
247.5				-.085	-.046	-.143	-.146	-.014	-.025	-.030	247.5
270.0				-.123				.007	-.025		270.0
292.5				-.121	-.153	.050	.095	.009	-.032	-.074	292.5
315.0				-.081	-.096	.097	.130	.036	-.030	-.052	315.0
337.5				-.020	-.034	.002	.134	.076	.012	-.031	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.187	-.185	-.172	-.230	-.225	-.247	-.246	-.245		.025
.075		-.150	-.163	-.172	-.230	-.167	-.234	-.233	-.232	-.240	.075
.125	-.139	-.150	-.163	-.172	-.231	-.142	-.230	-.219	-.224	-.236	.125
.175	-.139	-.155	-.167	-.172	-.211	-.137	-.225	-.227	-.223	-.230	.175
.225	-.148	-.155	-.172	-.192	-.210	-.137	-.225	-.230	-.221	-.226	.225
.275	-.148	-.155	-.177	-.206	-.192	-.131	-.221	-.230	-.221	-.218	.275
.325	-.159	-.167	-.185	-.214	-.192	-.121	-.207	-.230	-.221	-.201	.325
.375	-.159	-.174	-.192	-.219	-.188	-.118	-.207	-.230	-.214		.375
.425	-.168	-.179		-.219	-.186	-.114	-.207	-.226	-.215		.425
.475	-.164	-.179	-.199	-.228	-.181	-.110	-.207	-.227	-.207	-.195	.475
.550	-.171	-.184	-.199	-.239	-.183	-.106	-.218	-.220	-.196	-.193	.550
.650	-.171	-.206	-.210	-.228		-.106	-.204	-.219	-.203	-.199	.650
.750	-.163	-.200	-.193	-.212	-.139	-.107	-.197	-.208	-.201	-.196	.750
.800	-.163									-.194	.800
.850		-.185	-.193	-.211	-.128	-.117	-.190	-.199	-.211		.850
.900			-.185	-.211			-.174	-.187			.900
.950					-.128	-.115					.950
LOWER SURFACE											
.025		.624	.594	.542	.398	.377	.432	.423	.406		.025
.075		.520	.487	.442	.335	.268	.342	.353	.353	.341	.075
.125	.483	.458	.417	.374	.306	.229	.301	.315	.320	.320	.125
.175	.449	.407	.375	.334	.289	.230	.259	.279	.292	.298	.175
.225	.406	.377	.340	.297	.268	.221	.233	.257	.267	.281	.225
.275	.397	.355	.315	.275	.255	.225	.220	.240	.242	.259	.275
.325	.364	.328	.291	.253	.243	.214	.210	.223	.225	.242	.325
.375	.342	.307	.274	.236	.227	.207	.193	.210	.214		.375
.425	.317	.287		.229	.213	.193	.182	.197	.198		.425
.475	.300	.269	.237	.213	.206	.182	.175	.180	.184	.198	.475
.550	.274	.231	.212	.185	.179	.166	.144	.166	.166	.181	.550
.650	.242	.205	.171	.139		.139	.144	.122	.134	.160	.650
.750	.224	.187	.159	.129	.136	.128	.121	.116	.126	.145	.750
.800	.208									.138	.800
.850		.180	.146	.129	.126	.128	.111	.112	.116		.850
.900			.146	.129			.120	.108			.900
.950					.123	.128					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 15.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.069	.090	.109	.203	.181	.093	.074	.0
22.5				.130	.118	.207	.191	.185	.116	.090	22.5
45.0				.111	.078	.261	.220	.162	.079	.052	45.0
67.5				.019	.043	.318	.255		.008	-.029	67.5
90.0				-.079				-.005	-.086	-.131	90.0
112.5				-.171	-.199	-.179	-.173	-.057	-.138	-.097	112.5
135.0				-.218	-.133	-.164	-.174	-.116	-.129	-.074	135.0
157.5				-.107	-.147	-.212	-.211	-.109		-.047	157.5
180.0				-.086	-.109	-.145	-.155	-.169	-.035		180.0
202.5				-.060	-.012	-.059	-.123	-.079	-.056	-.059	202.5
225.0				-.093	-.135	-.179	-.151	-.069	-.061	-.038	225.0
247.5				-.133	-.130	-.215	-.191	-.039	-.037	-.059	247.5
270.0				-.201				-.005	-.041		270.0
292.5				-.155	-.153	.132	.174	-.001	-.081	-.146	292.5
315.0				-.067	-.085	.148	.217	.065	-.024	-.071	315.0
337.5				.027	.009	.029	.206	.142	.050	.003	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.268	-.265	-.245	-.289	-.278	-.281	-.294	-.277		.025
.075		-.234	-.244	-.245	-.280	-.233	-.281	-.294	-.271	-.251	.075
.125	-.229	-.234	-.234	-.245	-.280	-.192	-.281	-.291	-.270	-.250	.125
.175	-.225	-.234	-.234	-.238	-.269	-.171	-.281	-.298	-.270	-.248	.175
.225	-.221	-.234	-.234	-.252	-.279	-.165	-.291	-.298	-.269	-.247	.225
.275	-.219	-.214	-.234	-.258	-.262	-.164	-.284	-.298	-.266	-.243	.275
.325	-.219	-.237	-.246	-.262	-.271	-.164	-.277	-.295	-.265	-.233	.325
.375	-.206	-.234	-.246	-.262	-.257	-.164	-.277	-.293	-.258		.375
.425	-.206	-.231		-.262	-.237	-.164	-.268	-.293	-.260		.425
.475	-.200	-.227	-.253	-.262	-.218	-.164	-.274	-.287	-.260	-.236	.475
.550	-.201	-.217	-.225	-.262	-.184	-.164	-.279	-.280	-.253	-.239	.550
.650	-.201	-.217	-.234	-.262		-.201	-.274	-.291	-.265	-.247	.650
.750	-.201	-.217	-.228	-.244	-.166	-.233	-.273	-.258	-.253	-.253	.750
.800	-.201										.800
.850		-.217	-.235	-.244	-.161	-.220	-.273	-.230	-.248		.850
.900			-.235	-.244			-.274	-.209			.900
.950					-.161	-.202					.950
LOWER SURFACE											
.025		.729	.722	.678	.533	.403	.497	.513	.509		.025
.075		.637	.618	.575	.445	.301	.446	.456	.470	.459	.075
.125	.606	.582	.551	.508	.413	.223	.405	.415	.443	.437	.125
.175	.576	.539	.504	.458	.399	.205	.367	.386	.416	.416	.175
.225	.535	.502	.475	.423	.386	.223	.345	.362	.391	.401	.225
.275	.517	.475	.442	.397	.373	.239	.325	.338	.368	.380	.275
.325	.490	.449	.413	.373	.357	.250	.307	.318	.343	.364	.325
.375	.464	.429	.389	.359	.339	.250	.293	.300	.332		.375
.425	.439	.407		.336	.323	.250	.273	.286	.315		.425
.475	.415	.383	.350	.323	.311	.238	.260	.266	.298	.313	.475
.550	.387	.337	.312	.287	.283	.226	.225	.236	.276	.289	.550
.650	.353	.307	.272	.246		.200	.225	.206	.238	.272	.650
.750	.337	.291	.254	.230	.220	.199	.203	.217	.233	.267	.750
.800	.329									.261	.800
.850		.281	.246	.227	.224	.199	.201	.212	.227		.850
.900			.244	.225			.206	.213			.900
.950					.218	.199					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.112	-.116	-.076	-.056	-.070	-.063	-.060	.0
22.5				-.090	-.090	-.091	-.078	-.058	-.075	-.065	22.5
45.0				-.013	-.027	-.032	-.055	-.018	-.035	-.034	45.0
67.5				.014	.029	-.005	-.016		.011	.013	67.5
90.0				.041				.023	.025	.018	90.0
112.5				.040	.039	.027	.000	.015	.012	.029	112.5
135.0				.001	-.004	-.007	-.037	-.006	-.014	-.012	135.0
157.5				-.028	-.053	-.078	-.070	-.042		-.047	157.5
180.0				-.084	-.111	-.077	-.047	-.064	-.062		180.0
202.5				-.106	-.098	-.021	-.016	-.044		-.029	202.5
225.0				-.099	-.050	-.011	-.035	-.035	-.034	-.037	225.0
247.5				-.071	-.018	-.033	-.051	-.056	-.050	-.035	247.5
270.0				-.056				-.015	-.039		270.0
292.5				-.055	-.013	-.023	-.063	-.063	-.054	-.041	292.5
315.0				-.082	-.040	-.026	-.053	-.047	-.037	-.034	315.0
337.5				-.105	-.068	-.016	-.040	-.055	-.033	-.041	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.251	.235	.215	.087	.014	.087	.079	.080		.025
.075	.218	.210	.183	.165	.098	.014	.063	.057	.057	.040	.075
.125	.196	.166	.141	.119	.063	.014	.043	.044	.039	.032	.125
.175	.170	.139		.094	.062	.014	.021	.020	.022	.024	.175
.225	.147	.109	.093	.056	.034	.021	.014	.005	.009	.014	.225
.275	.135	.109	.074	.030	.037	.021	.005	.005	.004	-.001	.275
.325	.097	.083	.059	.019	.027	.021	.012	-.007	-.002	.005	.325
.375	.097	.071	.045	.007	.027	.021	.001	-.007	-.001	-.003	.375
.425	.074	.058		.007	.027		.001	-.017	-.015		.425
.475	.074	.049	.021	-.014	.015	.009	.001	-.032	-.023		.475
.550	.058	.033	.012	-.034	.007	.009	-.026	-.032	-.025	-.032	.550
.650	.032	.002	-.014	-.052		-.002	-.026	-.046	-.047	-.046	.650
.750	.019	-.015	-.027	-.047	-.011	-.009	-.026	-.043	-.047	-.060	.750
.800	.002									-.056	.800
.850		-.015	-.050	-.047	-.013	-.012	-.038	-.035	-.054		.850
.900			-.039	-.058			-.026	-.036			.900
.950					-.014	-.020					.950
LOWER SURFACE											

TABLE 2. Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 5.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.081	-.062	-.056	-.039	-.027	-.054	-.061	.0
22.5				.002	-.007	.020	-.028	-.007	-.019	-.023	22.5
45.0				.063	.043	.083	.040	.013	.015	.014	45.0
67.5				.060	.088	.121	.075		.028	.027	67.5
90.0				.037				.025	.016	.012	90.0
112.5				-.012	-.020	-.068	-.081	.004	-.015	-.014	112.5
135.0				-.075	-.092	-.116	-.093	-.035	-.051	-.050	135.0
157.5				-.111	-.140	-.128	-.091	-.083		-.050	157.5
180.0				-.123	-.081	-.053	-.081	-.095	-.039		180.0
202.5				-.086	-.051	-.060	-.086	-.084		-.018	202.5
225.0				-.060	-.040	-.064	-.112	-.082	-.046	-.027	225.0
247.5				-.050	-.002	-.041	-.067	-.041	-.043	-.034	247.5
270.0				-.064				.001	-.036		270.0
292.5				-.088	-.030	-.036	-.030	-.032	-.040	-.033	292.5
315.0				-.133	-.121	.008	.008	-.021	-.033	-.028	315.0
337.5				-.112	-.127	.009	.014	-.015	-.036	-.070	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		.019	.023	.004	-.081	.013	-.155	-.155	-.166		.025
.075	.042	.028	.018	-.009	-.063	-.025	-.133	-.142	-.150	-.168	.075
.125	.031	.013	-.004	-.034	-.077	-.042	-.122	-.138	-.144	-.167	.125
.175	.021	-.006		-.049	-.068	-.049	-.116	-.143	-.150	-.163	.175
.225	.006	-.020	-.039	-.074	-.068	-.057	-.109	-.143	-.149	-.162	.225
.275	-.005	-.027	-.051	-.093		-.057	-.106	-.136	-.152	-.167	.275
.325	-.026	-.044	-.063	-.102	-.068	-.057	-.103	-.143	-.149	-.157	.325
.375	-.034	-.053	-.072	-.113	-.071	-.057	-.102	-.130	-.144	-.157	.375
.425	-.045	-.060		-.119	-.075		-.101	-.130	-.153		.425
.475	-.049	-.068	-.090	-.129	-.075	-.047	-.091	-.132	-.157		.475
.550	-.064	-.084	-.101	-.148	-.084	-.047	-.091	-.132	-.155	-.165	.550
.650	-.079	-.104	-.127	-.162		-.056	-.091	-.132	-.155	-.155	.650
.750	-.079	-.113	-.133	-.146	-.095	-.062	-.091	-.121	-.141	-.149	.750
.800	-.074									-.145	.800
.850		-.106	-.143	-.136	-.095	-.064	-.089	-.112	-.137		.050
.900			-.122	-.138			-.089	-.110			.900
.950					-.095	-.069					.950
LOWER SURFACE											
.025		.491	.476	.435	.274	.002	.262	.245	.249		.025
.075	.420	.408	.375	.339	.248	.020	.203	.200	.200	.194	.075
.125	.369	.347	.310	.276	.203	.027	.166	.171	.177	.174	.125
.175	.337	.300		.241	.194	.039	.136	.145	.151	.156	.175
.225	.296	.270	.241	.204	.177	.060	.122	.129	.138	.146	.225
.275	.283	.251	.217	.177	.162	.070	.108	.117	.121	.128	.275
.325	.249	.224	.192	.158	.151	.082	.103	.103	.110	.120	.325
.375	.243	.208	.176	.143	.143	.081	.091	.100	.104	.107	.375
.425	.218	.194		.136	.135		.082	.085	.092		.425
.475	.208	.176	.149	.117	.127	.089	.082	.071	.082		.475
.550	.178	.151	.121	.093	.114	.091	.060	.062	.071	.071	.550
.650	.147	.117	.087	.060		.079	.058	.039	.044	.055	.650
.750	.123	.104	.074	.053	.075	.070	.050	.046	.040	.045	.750
.800	.103									.043	.800
.850		.091	.059	.053	.075	.071	.050	.046	.035		.850
.900			.059	.049			.053	.049			.900
.950					.075	.058					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 10.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.033	.001	.032	-.018	.026	.001	-.027	.0
22.5				.071	.062	.106	.068	.029	.047	.027	22.5
45.0				.107	.090	.187	.137	.070	.056	.042	45.0
67.5				.070	.113	.234	.166		.028	.023	67.5
90.0				.004				.005	-.019	-.039	90.0
112.5				-.072	-.097	-.144	-.116	-.018	-.067	-.096	112.5
135.0				-.149	-.179	-.134	-.110	-.075	-.102	-.103	135.0
157.5				-.181	-.137	-.119	-.124	-.112		-.047	157.5
180.0				-.111	-.109	-.133	-.159	-.152	-.053		180.0
202.5				-.085	-.084	-.149	-.182	-.186		-.019	202.5
225.0				-.057	-.002	-.057	-.114	-.070	-.091	-.046	225.0
247.5				-.084	-.083	-.127	-.121	-.070	-.036	-.053	247.5
270.0				-.105				.000	-.034		270.0
292.5				-.180	-.110	-.098	.076	-.004	-.029	-.004	292.5
315.0				-.154	-.182	.062	.082	.026	-.032	-.076	315.0
337.5				-.083	-.104	-.085	.054	.037	-.043	-.074	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.138	-.129	-.131	-.192	-.069	-.248	-.269	-.264		.025
.075	-.097	-.095	-.113	-.131	-.178	-.097	-.230	-.246	-.258	-.247	.075
.125	-.095	-.103	-.113	-.141	-.187	-.107	-.230	-.233	-.247	-.245	.125
.175	-.093	-.109		-.147	-.161	-.102	-.230	-.244	-.251	-.244	.175
.225	-.101	-.116	-.132	-.161	-.167	-.110	-.227	-.243	-.244	-.244	.225
.275	-.101	-.112	-.139	-.179	-.150	-.089	-.225	-.225	-.242	-.244	.275
.325	-.119	-.129	-.151	-.186	-.151	-.088	-.201	-.241	-.239	-.231	.325
.375	-.119	-.135	-.158	-.192	-.151	-.084	-.201	-.228	-.230	-.228	.375
.425	-.132	-.141		-.192	-.151		-.187	-.228	-.234		.425
.475	-.132	-.148	-.164	-.205	-.151	-.084	-.187	-.228	-.232		.475
.550	-.140	-.154	-.171	-.214	-.151	-.085	-.187	-.223	-.219	-.225	.550
.650	-.140	-.173	-.191	-.220		-.085	-.179	-.233	-.225	-.228	.650
.750	-.127	-.178	-.197	-.198	-.166	-.085	-.160	-.216	-.219	-.226	.750
.800	-.127									-.224	.800
.850		-.154	-.190	-.196	-.150	-.090	-.136	-.206	-.222		.850
.900			-.176	-.197			-.133	-.194			.900
.950					-.135	-.098					.950
LOWER SURFACE											
.025		.694	.673	.613	.453	-.034	.392	.377	.363		.025
.075	.607	.586	.547	.501	.398	-.026	.326	.326	.322	.306	.075
.125	.545	.515	.467	.429	.345	-.026	.277	.292	.298	.284	.125
.175	.505	.466		.379	.327	-.001	.254	.263	.265	.266	.175
.225	.461	.429	.381	.344	.302	.056	.234	.243	.249	.256	.225
.275	.443	.402	.357	.313	.289	.091	.218	.229	.232	.238	.275
.325	.413	.365	.326	.298	.276	.117	.211	.215	.216	.228	.325
.375	.391	.344	.307	.275	.258	.128	.197	.207	.206	.216	.375
.425	.365	.326		.261	.248		.185	.193	.193		.425
.475	.347	.307	.269	.245	.238	.145	.179	.177	.182		.475
.550	.315	.270	.236	.216	.216	.139	.152	.162	.168	.173	.550
.650	.275	.235	.204	.174		.120	.152	.135	.135	.154	.650
.750	.243	.212	.185	.156	.162	.120	.145	.140	.136	.142	.750
.800	.224									.139	.800
.850		.203	.174	.156	.164	.120	.145	.145	.132		.850
.900			.174	.148			.145	.141			.900
.950					.164	.102					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 15.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.046	.084	.130	.077	.099	.057	.037	.0
22.5				.161	.146	.226	.167	.116	.107	.092	22.5
45.0				.170	.141	.295	.252	.146	.110	.081	45.0
67.5				.091	.147	.361	.277		.037	.011	67.5
90.0				-.014				-.007	-.056	-.091	90.0
112.5				-.120	-.148	-.180	-.149	-.055	-.135	-.138	112.5
135.0				-.209	-.194	-.146	-.149	-.124	-.160	-.035	135.0
157.5				-.180	-.140	-.158	-.166	-.125		-.053	157.5
180.0				-.156	-.197	-.198	-.174	-.135	-.054		180.0
202.5				-.093	-.084	-.121	-.172	-.151		-.046	202.5
225.0				-.088	-.071	-.130	-.161	-.090	-.140	-.068	225.0
247.5				-.152	-.172	-.197	-.162	-.118	-.042	-.014	247.5
270.0				-.166				.000	-.030		270.0
292.5				-.207	-.165	-.039	.168	-.001	-.040	.035	292.5
315.0				-.127	-.148	.092	.149	.043	-.061	-.096	315.0
337.5				-.022	-.043	-.047	.095	.071	-.025	-.041	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.242	-.228	-.212	-.241	-.075	-.276	-.291	-.285		.025
.075	-.200	-.196	-.197	-.200	-.224	-.094	-.276	-.291	-.278	-.251	.075
.125	-.189	-.197	-.197	-.200	-.242	-.121	-.276	-.279	-.280	-.251	.125
.175	-.189	-.197		-.203	-.229	-.130	-.286	-.289	-.277	-.249	.175
.225	-.189	-.196	-.196	-.215	-.238	-.162	-.286	-.289	-.277	-.248	.225
.275	-.189	-.179	-.200	-.227	-.215	-.206	-.276	-.278	-.275	-.242	.275
.325	-.202	-.194	-.203	-.227	-.227	-.211	-.264	-.289	-.273	-.238	.325
.375	-.185	-.203	-.211	-.234	-.216	-.187	-.264	-.288	-.267	-.235	.375
.425	-.200	-.203		-.234	-.209		-.256	-.287	-.274		.425
.475	-.192	-.203	-.213	-.240	-.217	-.177	-.256	-.283	-.272		.475
.550	-.192	-.203	-.217	-.250	-.209	-.161	-.261	-.278	-.266	-.241	.550
.650	-.184	-.211	-.212	-.241		-.185	-.255	-.286	-.260	-.249	.650
.750	-.176	-.202	-.203	-.223	-.186	-.200	-.248	-.259	-.249	-.254	.750
.800	-.178									-.251	.800
.850		-.194	-.203	-.224	-.158	-.183	-.248	-.244	-.241		.850
.900			-.200	-.224			-.230	-.224			.900
.950					-.149	-.171					.950
LOWER SURFACE											
.025		.835	.821	.762	.594	-.019	.432	.450	.441		.025
.075	.758	.729	.695	.649	.528	-.021	.399	.410	.416	.398	.075
.125	.695	.667	.621	.569	.477	-.043	.361	.379	.398	.383	.125
.175	.656	.611		.515	.454	-.031	.340	.356	.374	.368	.175
.225	.610	.575	.528	.479	.429	-.011	.324	.341	.350	.362	.225
.275	.587	.538	.491	.448	.413	.027	.307	.324	.334	.344	.275
.325	.554	.511	.461	.421	.391	.069	.293	.307	.319	.331	.325
.375	.521	.481	.441	.403	.379	.093	.280	.294	.305	.319	.375
.425	.499	.460		.384	.354		.268	.274	.290		.425
.475	.475	.437	.397	.365	.355	.142	.257	.262	.281		.475
.550	.441	.385	.357	.328	.326	.157	.230	.241	.258	.272	.550
.650	.398	.356	.315	.279		.157	.242	.229	.239	.268	.650
.750	.365	.340	.293	.260	.269	.182	.224	.236	.240	.261	.750
.800	.344									.258	.800
.850		.319	.286	.260	.262	.182	.231	.238	.234		.850
.900			.276	.260			.231	.237			.900
.950					.260	.182					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.169	-.174	-.021	-.049	-.106	-.119	-.084	.0
22.5				-.081	-.105	-.121	-.123	-.085	-.105	-.123	22.5
45.0				.027	-.004	-.012	-.046	-.032	-.028	-.033	45.0
67.5				.085	.091	.060	.027		.037	.046	67.5
90.0				.113				.075	.051	.054	90.0
112.5				.098	.102	.075	.042	.057	.047	.068	112.5
135.0				.025	.018	.001	-.021	-.011	-.007	-.008	135.0
157.5				-.049	-.076	-.105	-.113	-.071		-.109	157.5
180.0				-.138	-.171	-.077	-.046	-.104	-.126		180.0
202.5				-.189	-.128	-.061	-.078	-.078	-.064		202.5
225.0				-.126	-.137	-.160	-.102	-.065	-.047	-.069	225.0
247.5				-.092	-.098	-.093	-.112	-.079	-.063	-.056	247.5
270.0				-.060				-.058	-.064		270.0
292.5				-.076	-.081	-.062	-.109	-.106	-.058	-.054	292.5
315.0				-.111	-.143	-.172	-.116	-.076	-.053	-.053	315.0
337.5				-.188	-.125	-.075	-.090	-.086	-.070	-.068	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.303	.298	.264	.141	-.002	.070	.050	.046		.025
.075	.191	.250	.230	.203		.001	.045	.030	.029	.013	.075
.125	.180	.211	.188	.151	.112	-.002	.031	.015	.019	.003	.125
.175	.163	.177	.158	.113	.096	-.008	.014	.005	.010	-.006	.175
.225	.146	.153	.134	.087	.085	.006	.008	-.007	.001	-.011	.225
.275	.131	.132	.110	.064	.082	.001	.002	-.008	-.007	-.015	.275
.325	.115	.118	.095	.045	.071	-.005	.002	-.014	-.010	-.021	.325
.375	.106	.101	.081	.032		-.009	-.009	-.019	-.017		.375
.425	.091	.089	.072	.023	.064		-.009	-.024	-.022		.425
.475	.082	.078	.057	.008	.059	-.021	-.014	-.027	-.027	-.041	.475
.550	.062	.051	.037	-.013	.050	-.031	-.033	-.042	-.039	-.047	.550
.650	.036	.025	-.013	-.038		-.050	-.028	-.051	-.047	-.053	.650
.750	.024	.011	-.027		.024	-.056	-.037	-.035	-.050	-.057	.750
.800	.019									-.062	.800
.850		.006	-.037	-.039	.028	-.062	-.037	-.039	-.050		.850
.900			-.034	-.042			-.036	-.035			.900
.950					.030	-.066					.950
LOWER SURFACE											

TABLE 2. Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 5.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.147	-.112	-.030	-.106	-.101	-.125	-.116	.0
22.5				-.007	-.021	-.019	-.053	-.068	-.077	-.065	22.5
45.0				.092	.068	.089	.053	.027	.008	.011	45.0
67.5				.117	.119	.159	.116		.050	.051	67.5
90.0				.103				.065	.048	.046	90.0
112.5				.044	.028	-.020	-.039	.034	.015	.000	112.5
135.0				-.055	-.071	-.097	-.099	-.040	-.050	-.060	135.0
157.5				-.126	-.153	-.123	-.085	-.106		-.129	157.5
180.0				-.196	-.147	-.094	-.097	-.124	-.104		180.0
202.5				-.169	-.147	-.167	-.155	-.111	-.070		202.5
225.0				-.110	-.145	-.184	-.181	-.095	-.050	-.040	225.0
247.5				-.070	-.021	-.056	-.110	-.102	-.078	-.049	247.5
270.0				-.084				-.043	-.079		270.0
292.5				-.110	-.152	-.161	-.139	-.060	-.069	-.076	292.5
315.0				-.208	-.136	-.075	-.113	-.056	-.061	-.071	315.0
337.5				-.188	-.186	-.069	-.036	-.060	-.067	-.076	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.090	.091	.081	-.014	.147	-.148	-.181	-.197		.025
.075	.044	.099	.077	.057		.062	-.133	-.172	-.182	-.200	.075
.125	.037	.071	.050	.024	.002	.023	-.127	-.134	-.180	-.197	.125
.175	.025	.048	.026	.007	.004	-.004	-.127	-.152	-.173	-.193	.175
.225	.015	.031	.011	-.023	-.006	-.011	-.090	-.153	-.166	-.192	.225
.275	.013	.031	-.004	-.052	-.001	-.038	-.070	-.141	-.164	-.195	.275
.325	-.007	.006	-.018	-.064	-.011	-.034	-.063	-.144	-.160	-.164	.325
.375	-.004	-.008	-.030	-.072		-.043	-.082	-.113	-.144		.375
.425	-.020	-.018	-.025	-.074	-.012		-.076	-.101	-.150		.425
.475	-.012	.000	-.046	-.093	-.018	-.059	-.069	-.094	-.154	-.175	.475
.550	-.029	-.039	-.056	-.110	-.025	-.074	-.091	-.101	-.147	-.167	.550
.650	-.044	-.065	-.088	-.132		-.091	-.100	-.120	-.157	-.173	.650
.750	-.042	-.077	-.088		-.043	-.099	-.088	-.102	-.129	-.175	.750
.800	-.040									-.172	.800
.850		-.070	-.107	-.095	-.038	-.103	-.075	-.099	-.114		.850
.900			-.084	-.105			-.087	-.098			.900
.950					-.034	-.110					.950
LOWER SURFACE											
.025		.559	.548	.488	.327	-.233	.235	.209	.205		.025
.075	.434	.465	.438	.386		-.158	.183	.163	.170	.155	.075
.125	.393	.400	.368	.317	.247	-.116	.152	.135	.150	.137	.125
.175	.356	.351	.318	.265	.227	-.094	.133	.118	.131	.124	.175
.225	.321	.318	.285	.234	.210	-.086	.115	.100	.116	.115	.225
.275	.303	.290	.261	.206	.202	-.069	.102	.091	.105	.103	.275
.325	.277	.269	.236	.185	.189	-.046	.095	.081	.095	.094	.325
.375	.259	.251	.219	.168		-.030	.083	.076	.088		.375
.425	.241	.231	.203	.152	.169		.076	.067	.078		.425
.475	.225	.214	.182	.137	.163	-.021	.077	.058	.072	.066	.475
.550	.195	.176	.149	.112	.146	-.018	.057	.051	.059	.055	.550
.650	.158	.147	.100	.075		-.014	.065	.032	.044	.045	.650
.750	.132	.130	.078		.113	-.008	.043	.056	.040	.036	.750
.800	.125									.035	.800
.850		.121	.070	.057	.111	-.008	.037	.053	.039		.850
.900			.067	.058			.043	.048			.900
.950					.109	-.026					.950

TABLE 2, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 10.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.098	-.026	-.021	-.058	-.089	-.081	-.086	.0
22.5				.077	.071	.102	.056	.022	.000	.016	22.5
45.0				.165	.138	.220	.159	.100	.072	.067	45.0
67.5				.151	.197	.280	.217		.075	.072	67.5
90.0				.084				.051	.035	.022	90.0
112.5				-.007	-.029	-.088	-.092	.014	-.043	-.049	112.5
135.0				-.119	-.137	-.162	-.113	-.068	-.100	-.141	135.0
157.5				-.173	-.178	-.114	-.112	-.138		-.063	157.5
180.0				-.194	-.147	-.114	-.138	-.121	-.099		180.0
202.5				-.175	-.181	-.175	-.157	-.114	-.056		202.5
225.0				-.096	-.079	-.132	-.166	-.130	-.060	-.030	225.0
247.5				-.093	-.063	-.092	-.128	-.109	-.089	-.048	247.5
270.0				-.132				-.005	-.090		270.0
292.5				-.148	-.138	-.088	-.147	-.092	-.076	-.091	292.5
315.0				-.217	-.194	-.056	-.013	-.046	-.070	-.104	315.0
337.5				-.145	-.171	-.150	-.005	-.012	-.063	-.158	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.070	-.059	-.058	-.118	.152	-.216	-.231	-.243		.025
.075	-.058	-.024	-.043	-.055		.099	-.233	-.255	-.241	-.234	.075
.125	-.052	-.037	-.052	-.069	-.078	.071	-.221	-.196	-.242	-.236	.125
.175	-.061	-.047	-.064	-.075	-.066	-.032	-.217	-.196	-.238	-.236	.175
.225	-.065	-.062	-.075	-.101	-.078	-.095	-.220	-.243	-.236	-.236	.225
.275	-.070	-.046	-.084	-.123	-.069	-.096	-.221	-.241	-.239	-.236	.275
.325	-.091	-.076	-.094	-.131	-.080	-.085	-.163	-.179	-.236	-.215	.325
.375	-.081	-.085	-.103	-.139		-.078	-.193	-.180	-.219		.375
.425	-.104	-.095	-.095	-.132	-.072		-.171	-.231	-.222		.425
.475	-.085	-.064	-.114	-.153	-.083	-.083	-.151	-.184	-.228	-.219	.475
.550	-.097	-.101	-.116	-.166	-.087	-.090	-.163	-.210	-.215	-.212	.550
.650	-.087	-.128	-.139	-.178		-.110	-.148	-.196	-.228	-.222	.650
.750	-.078	-.132	-.161		-.096	-.122	-.151	-.191	-.204	-.227	.750
.800	-.080									-.228	.800
.850		-.108	-.152	-.139	-.095	-.135	-.122	-.175	-.186		.850
.900			-.125	-.153			-.148	-.177			.900
.950					-.090	-.140					.950
LOWER SURFACE											
.025		.799	.779	.694	.523	-.254	.344	.314	.301		.025
.075	.677	.671	.634	.568		-.209	.292	.274	.273	.256	.075
.125	.608	.594	.551	.485	.405	-.124	.261	.247	.255	.243	.125
.175	.563	.539	.487	.432	.375	-.082	.235	.227	.237	.225	.175
.225	.520	.499	.456	.394	.354	-.070	.217	.213	.219	.218	.225
.275	.502	.463	.418	.361	.339	-.072	.208	.200	.208	.206	.275
.325	.467	.437	.390	.333	.323	-.083	.196	.185	.198	.196	.325
.375	.440	.409	.365	.316		-.072	.184	.178	.193		.375
.425	.417	.391	.348	.296	.292		.173	.172	.183		.425
.475	.393	.392	.325	.276	.285	-.036	.173	.162	.175	.165	.475
.550	.346	.320	.284	.242	.261	-.005	.163	.145	.157	.147	.550
.650	.298	.284	.228	.206		.014	.173	.131	.139	.141	.650
.750	.258	.267	.204		.215	.030	.140	.146	.133	.141	.750
.800	.245									.137	.800
.850		.255	.202	.183	.214	.043	.141	.141	.127		.850
.900			.201	.181			.143	.137			.900
.950					.213	.043					.950

TABLE 2, Concluded

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

MIDWING CONFIGURATION

$\alpha = 15.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.027	.068	-.014	.040	-.004	-.034	-.007	.0
22.5				.189	.168	.224	.169	.106	.082	.097	22.5
45.0				.240	.210	.340	.272	.169	.133	.119	45.0
67.5				.181	.237	.405	.331		.089	.085	67.5
90.0				.072				.015	.000	-.011	90.0
112.5				-.064	-.088	-.140	-.131	-.025	-.110	-.118	112.5
135.0				-.182	-.200	-.164	-.138	-.085	-.159	-.117	135.0
157.5				-.207	-.175	-.169	-.143	-.152		-.109	157.5
180.0				-.181	-.174	-.168	-.174	-.150	-.109		180.0
202.5				-.204	-.214	-.186	-.181	-.127	-.067		202.5
225.0				-.108	-.100	-.118	-.172	-.133	-.093	-.065	225.0
247.5				-.197	-.187	-.185	-.162	-.126	-.120	-.091	247.5
270.0				-.179				.002	-.093		270.0
292.5				-.193	-.157	-.054	-.076	-.036	-.070	-.125	292.5
315.0				-.183	-.209	-.019	.070	-.014	-.083	-.159	315.0
337.5				-.071	-.096	-.081	.028	.002	-.063	-.129	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.219	-.202	-.191	-.214	-.081	-.284	-.273	-.278		.025
.075	-.196	-.169	-.173	-.173		-.100	-.295	-.297	-.281	-.259	.075
.125	-.183	-.165	-.164	-.179	-.215	-.115	-.291	-.254	-.281	-.259	.125
.175	-.181	-.164	-.169	-.178	-.183	-.126	-.290	-.249	-.281	-.256	.175
.225	-.181	-.169	-.171	-.191	-.185	-.156	-.271	-.279	-.280	-.253	.225
.275	-.179	-.153	-.175	-.209	-.160	-.176	-.285	-.276	-.282	-.253	.275
.325	-.195	-.171	-.183	-.214	-.167	-.191	-.214	-.241	-.281	-.242	.325
.375	-.184	-.177	-.190	-.216		-.189	-.283	-.235	-.272		.375
.425	-.197	-.183	-.181	-.213	-.157		-.281	-.273	-.273		.425
.475	-.176	-.166	-.197	-.227	-.163	-.204	-.251	-.245	-.278	-.244	.475
.550	-.166	-.188	-.197	-.238	-.165	-.188	-.266	-.296	-.273	-.247	.550
.650	-.162	-.204	-.202	-.230		-.181	-.255	-.253	-.274	-.255	.650
.750	-.162	-.190	-.200		-.175	-.169	-.258	-.267	-.259	-.265	.750
.800	-.165									-.267	.800
.850		-.176	-.198	-.204	-.167	-.163	-.217	-.249	-.260		.850
.900			-.186	-.214			-.233	-.240			.900
.950					-.158	-.169					.950
LOWER SURFACE											
.025		.962	.942	.868	.680	-.166	.361	.357	.362		.025
.075	.850	.842	.794	.733		-.173	.341	.343	.359	.337	.075
.125	.779	.756	.711	.641	.553	-.093	.318	.318	.349	.325	.125
.175	.734	.703		.584	.517	-.038	.295	.309	.333	.314	.175
.225	.682	.661	.609	.546	.487	-.040	.287	.298	.319	.309	.225
.275	.659	.621	.568	.511	.467	-.058	.277	.289	.307	.301	.275
.325	.626	.585	.536	.483	.448	-.068	.268	.280	.296	.292	.325
.375	.594	.564	.512	.462		-.055	.261	.272	.286		.375
.425	.561	.543	.493	.443	.414		.259	.259	.274		.425
.475	.534	.489	.470	.420	.413	-.014	.260	.249	.263	.260	.475
.550	.490	.457	.419	.381	.381	.021	.243	.232	.247	.251	.550
.650	.428	.419	.385	.333		.069	.255	.219	.232	.248	.650
.750	.386	.395	.340		.323	.100	.235	.242	.233	.242	.750
.800	.374									.237	.800
.850		.385	.330	.306	.321	.103	.234	.239	.226		.850
.900			.327	.310			.234	.236			.900
.950					.318	.100					.950

TABLE 3

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

HIGH-WING CONFIGURATION

$\alpha = 0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.029	-.014	-.001	.027	-.014	-.019	-.007	.0
22.5				-.038	-.014	.005	.014	-.021	-.017	-.003	22.5
45.0				-.022	-.013	.005		-.034	-.010	-.007	45.0
67.5				-.038	-.012	.005	-.019		-.003	-.001	67.5
90.0				-.038				-.010	.002	-.003	90.0
112.5				-.038	.003	-.028	-.064	-.001	.000	.009	112.5
135.0				-.038	.052	-.045	-.064	-.001	.002	.003	135.0
157.5				-.038				-.001		-.003	157.5
180.0				-.038				-.002	.006		180.0
202.5				-.038				-.002		-.003	202.5
225.0				-.038	.071	-.042	-.065	.001	.006	-.003	225.0
247.5				-.038	.070	-.015	-.065	.008	.005	-.009	247.5
270.0				-.034				.001	.005		270.0
292.5				-.036	-.010	.016	-.020	-.015	.000	-.017	292.5
315.0				-.033	-.010	.020	-.003	-.021	-.008	-.015	315.0
337.5				-.033	-.015	.014	.010	-.013		-.013	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.145	.134	.143	.057	.073	.134	.147	.160		.025
.075		.118	.101	.082	.025	.034	.085	.114	.129	.111	.075
.125	.105	.085	.063	.049	.012	.007	.053	.078	.101	.092	.125
.175	.085	.064	.040	.027	.000	-.006	.024	.051	.078	.078	.175
.225	.056	.044	.025	.005	-.005	-.013	.002	.031	.060	.070	.225
.275	.051	.031	.019	-.006		-.017	-.007	.018	.046	.052	.275
.325	.034	.015	-.006		-.008	-.018	-.014	.004	.035	.043	.325
.375	.025	.007	-.013	-.021	-.017	-.018	-.025	-.002	.024	.032	.375
.425	.007	-.005	-.032	-.026	-.017	-.024	-.031	-.013	.012		.425
.475	.004	-.017	-.032	-.033	-.017	-.024	-.036	-.028	.004	.007	.475
.550	-.014	-.033	-.045	-.050	-.026	-.030	-.054	-.044	-.010	-.004	.550
.650	-.033	-.054	-.069	-.059		-.039	-.054	-.063	-.032	-.016	.650
.750	-.041	-.065	-.073	-.060	-.031	-.041	-.068		-.043	-.030	.750
.800	-.051									-.037	.800
.850		-.066	-.075	-.060	-.036	-.038	-.068	-.058	-.051		.850
.900			-.075	-.060			-.064	-.062			.900
.950					-.040	-.038					.950
LOWER SURFACE											
.025		.150	.109	.102	.069	.073	.126	.120	.121		.025
.075	.112	.118	.078	.068	.038	.051	.073	.089	.101		.075
.125	.093	.100	.052	.034	.019	.028	.041	.057	.077	.075	.125
.175	.082	.079	.027	.011	.009	.025	.019	.034	.053	.064	.175
.225	.075	.060	.015	-.007	.005	.009	.004	.018	.036	.038	.225
.275	.063	.048	-.002	-.017	-.012	-.006	-.007	.007	.017	.032	.275
.325	.062	.037	-.014	-.024	-.012	-.012	-.018	-.008	.007	.017	.325
.375		.037	-.021	-.028	-.027	-.025	-.024	-.015	-.004	.004	.375
.425		.032	-.030	-.037	-.038	-.027	-.030	-.024	-.014	-.005	.425
.475	.029	.020	-.039	-.041	-.045	-.031	-.041	-.033	-.024	-.019	.475
.550	.013	.007	-.053	-.059	-.058	-.055	-.052	-.049	-.041	-.034	.550
.650	-.008	-.028	-.076	-.065	-.069		-.066		-.060	-.053	.650
.750	-.020	-.022	-.048	-.085	-.064	-.046	-.076	-.078	-.072	-.060	.750
.800	-.027									-.063	.800
.850		-.043		-.085	-.052	-.082	-.076	-.078	-.075		.850
.900			-.040	-.085			-.076	-.078			.900
.950					-.058	-.081					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

HIGH-WING CONFIGURATION

$\alpha = 2.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.030	-.014	-.005	.057	.026	-.013	-.014	.0
22.5				-.026	-.013	.006	.042	.017	-.006	-.009	22.5
45.0				-.014	-.013	.037		.005	-.009	-.016	45.0
67.5				-.037	-.019	.047	.016		-.010	-.010	67.5
90.0				-.043				-.001	-.010	-.014	90.0
112.5				-.043	.043	.036	-.030	-.001	-.019	-.003	112.5
135.0				-.043	.139	.012	-.036	-.010	-.019	-.003	135.0
157.5				-.038				-.010		-.012	157.5
180.0				-.043				-.014	-.009		180.0
202.5				-.038				-.014		-.008	202.5
225.0				-.040	.146	.019	-.031	-.024	-.016	-.014	225.0
247.5				-.047	.089	.044	-.029	-.010	-.016	-.020	247.5
270.0				-.036				-.010	-.016		270.0
292.5				-.033	-.019	.050	.016	-.010	-.008	-.020	292.5
315.0				-.030	-.014	.050	.033	-.001	-.008	-.014	315.0
337.5				-.030	-.014	.012	.038	.012		-.014	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.001	.009	.047	-.013	-.008	.028	.036	.057		.025
.075		.025	.009	.001	-.037	-.031	.005	.014	.036	.024	.075
.125	.025	.002	-.011	-.017	-.046	-.045	-.018	.000	.019	.013	.125
.175	.008	-.015	-.026	-.030	-.045	-.045	-.043	-.024	.001	.003	.175
.225	-.006	-.031	-.041	-.051	-.053	-.052	-.056	-.036	-.010	-.005	.225
.275	-.015	-.026	-.046	-.070		-.052	-.066	-.043	-.019	-.018	.275
.325	-.039	-.047	-.064		-.051	-.052	-.064	-.056	-.028	-.018	.325
.375	-.039	-.056	-.070	-.081	-.058	-.054	-.078	-.059	-.031	-.026	.375
.425	-.056	-.064	-.089	-.073	-.051	-.063	-.082	-.066	-.041		.425
.475	-.051	-.075	-.082	-.085	-.057	-.057	-.082	-.084	-.048	-.044	.475
.550	-.063	-.084	-.091	-.097	-.063	-.063	-.103	-.092	-.056	-.054	.550
.650	-.082	-.103	-.114	-.104		-.073	-.103	-.121	-.082	-.072	.650
.750	-.086	-.111	-.122	-.098	-.070	-.077	-.103		-.083	-.079	.750
.800	-.089									-.073	.800
.850		-.113	-.122	-.094	-.079	-.077	-.103	-.098	-.090		.850
.900			-.120	-.098			-.103	-.095			.900
.950					-.079	-.071					.950
LOWER SURFACE											
.025		.238	.222	.216	.165	.191	.238	.248	.252		.025
.075	.189	.192	.172	.155	.117	.127	.172	.186	.200		.075
.125	.167	.161	.127	.118	.108	.098	.130	.146	.166	.169	.125
.175	.145	.130	.104	.086	.095	.088	.094	.124	.134	.147	.175
.225	.137	.111	.084	.065	.078	.076	.077	.097	.105	.117	.225
.275	.115	.094	.064	.056	.054	.063	.064	.077	.092	.108	.275
.325	.099	.080	.052	.040	.062	.060	.056	.058	.079	.088	.325
.375		.069	.039	.034	.028	.037	.041	.051	.069	.075	.375
.425		.056	.032	.023	.017	.031	.032	.039	.054	.059	.425
.475	.062	.049	.019	.013	.013	.039	.023	.032	.045	.045	.475
.550	.045	.035	.005	-.013	-.006	.001	.007	.013	.015	.031	.550
.650	.028	.010	-.023	-.020	-.027		-.015		.000	.009	.650
.750	.013	-.003	-.016	-.045	-.017	.002	-.026	-.028	-.012	-.002	.750
.800	.007									-.007	.800
.850		-.010		-.039	-.013	-.027	-.026	-.028	-.018		.850
.900			-.021	-.037			-.026	-.028			.900
.950					-.028	-.036					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

HIGH-WING CONFIGURATION

$\alpha = 5.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.014	.000	.006	.090	.064	.019	.005	.0
22.5				-.014	.000	.006	.073	.057	.015	-.002	22.5
45.0				-.006	-.015	.052		.041	-.001	-.009	45.0
67.5				-.038	-.024	.080	.052		-.015	-.014	67.5
90.0				-.050				.006	-.021	-.026	90.0
112.5				-.050	.093	.107	.023	-.002	-.026	-.009	112.5
135.0				-.050	.197	.071	.002	-.016	-.021	-.008	135.0
157.5				-.037				-.010		-.012	157.5
180.0				-.038				-.010	-.003		180.0
202.5				-.045				-.016		-.009	202.5
225.0				-.045	.221	.087	.006	-.040	-.019	-.024	225.0
247.5				-.052	.098	.112	.019	-.019	-.019	-.031	247.5
270.0				-.041				-.008	-.019		270.0
292.5				-.033	-.024	.093	.055	.007	-.005	-.029	292.5
315.0				-.024	-.013	.073	.063	.030	-.001	-.007	315.0
337.5				-.010	-.002	.014	.069	.041		-.006	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.114	-.138	-.021	-.060	-.049	-.059	-.053	-.044		.025
.075		-.108	-.143	-.053	-.072	-.072	-.066	-.059	-.047	-.054	.075
.125	-.046	-.102	-.135	-.065	-.075	-.082	-.075	-.059	-.054	-.059	.125
.175	-.052	-.082	-.080	-.072	-.070	-.072	-.095	-.082	-.066	-.067	.175
.225	-.058	-.054	-.063	-.091	-.075	-.083	-.107	-.092	-.076	-.070	.225
.275	-.065	-.062	-.070	-.107		-.078	-.114	-.092	-.081	-.079	.275
.325	-.086	-.072	-.078		-.067	-.080	-.109	-.099	-.089	-.077	.325
.375	-.085	-.083	-.089	-.115	-.067	-.076	-.128	-.102	-.089	-.084	.375
.425	-.102	-.073	-.111	-.104	-.062	-.082	-.118	-.107	-.098		.425
.475	-.090	-.095	-.105	-.117	-.066	-.078	-.122	-.128	-.103	-.101	.475
.550	-.101	-.102	-.112	-.122	-.077	-.080	-.144	-.131	-.107	-.108	.550
.650	-.114	-.108	-.143	-.124		-.098	-.130	-.154	-.130	-.117	.650
.750	-.092	-.115	-.140	-.114	-.079	-.095	-.128		-.131	-.108	.750
.800	-.103									-.098	.800
.850		-.118	-.136	-.114	-.089	-.097	-.129	-.125	-.131		.850
.900			-.129	-.120			-.121	-.120			.900
.950					-.096	-.097					.950
LOWER SURFACE											
.025		.336	.316	.321	.283	.271	.331	.344	.352		.025
.075	.278	.279	.249	.235	.186	.182	.238	.263	.280		.075
.125	.252	.239	.193	.184	.135	.146	.187	.210	.235	.232	.125
.175	.226	.210	.169	.152	.117	.133	.147	.172	.201	.207	.175
.225	.215	.184	.148	.126	.116	.122	.133	.154	.179	.177	.225
.275	.190	.166	.118	.105	.105	.109	.120	.130	.148	.168	.275
.325	.173	.150	.108	.091	.105	.109	.103	.120	.135	.155	.325
.375		.132	.091	.082	.084	.092	.085	.103	.117	.129	.375
.425		.119	.076	.064	.075	.075	.075	.089	.110	.118	.425
.475	.128	.112	.068	.056	.075	.084	.068	.076	.098	.096	.475
.550	.109	.090	.044	.031	.051	.047	.049	.049	.060	.073	.550
.650	.093	.066	.030	.031	.023		.019		.044	.051	.650
.750	.076	.048	.038	-.006	.023	.045	.009	.008	.033	.041	.750
.800	.068									.041	.800
.850		.042		.000	.023	-.007	.001	.008	.023		.850
.900			.026	-.007			.001	.006			.900
.950					-.006	-.007					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.013	.010	.008	.112	.096	.043	.022	.0
22.5				-.006	.002	.003	.098	.087	.040	.009	22.5
45.0				-.013	-.028	.044		.075	.012	-.007	45.0
67.5				-.052	-.049	.096	.079		-.015	-.031	67.5
90.0				-.077				-.008	-.030	-.044	90.0
112.5				-.077	.079	.173	.065	-.012	-.035	-.021	112.5
135.0				-.070	.253	.135	.042	-.035	-.030	-.003	135.0
157.5				-.055				-.016		-.009	157.5
180.0				-.055				-.016	-.009		180.0
202.5				-.055				-.023		-.009	202.5
225.0				-.066	.282	.150	.047	-.055	-.028	-.019	225.0
247.5				-.082	.089	.182	.062	-.027	-.033	-.028	247.5
270.0				-.069				-.006	-.027		270.0
292.5				-.054	-.054	.114	.079	.037	-.012	-.036	292.5
315.0				-.029	-.023	.084	.089	.068	.009	-.009	315.0
337.5				-.003	-.002	.014	.096	.079		.005	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.148	-.143	-.088	-.110	-.100	-.124	-.126	-.123		.025
.075		-.115	-.127	-.113	-.121	-.120	-.124	-.117	-.117	-.126	.075
.125	-.103	-.118	-.124	-.120	-.115	-.121	-.128	-.114	-.118	-.128	.125
.175	-.101	-.126	-.130	-.120	-.103	-.107	-.142	-.129	-.124	-.128	.175
.225	-.105	-.133	-.142	-.141	-.120	-.120	-.150	-.137	-.131	-.129	.225
.275	-.113	-.120	-.133	-.155		-.122	-.158	-.137	-.134	-.136	.275
.325	-.136	-.139	-.147		-.084	-.121	-.145	-.156	-.138	-.125	.325
.375	-.127	-.146	-.155	-.150	-.081	-.094	-.161	-.146	-.134	-.131	.375
.425	-.141	-.149	-.155	-.142	-.077	-.082	-.156	-.155	-.143		.425
.475	-.129	-.155	-.161	-.152	-.084	-.089	-.152	-.168	-.148	-.145	.475
.550	-.139	-.155	-.161	-.160	-.096	-.100	-.168	-.163	-.149	-.152	.550
.650	-.146	-.169	-.172	-.160		-.109	-.159	-.182	-.169	-.151	.650
.750	-.129	-.161	-.166	-.150	-.091	-.116	-.160		-.151	-.136	.750
.800	-.135									-.131	.800
.850		-.153	-.166	-.147	-.111	-.116	-.166	-.148	-.148		.850
.900			-.155	-.159			-.153	-.148			.900
.950					-.115	-.116					.950
LOWER SURFACE											
.025		.419	.398	.413	.385	.376	.413	.430	.428		.025
.075	.356	.353	.319	.312	.250	.244	.317	.339	.354		.075
.125	.327	.311	.271	.263	.188	.203	.265	.285	.302	.310	.125
.175	.297	.280	.235	.217	.171	.193	.222	.242	.274	.285	.175
.225	.283	.251	.205	.191	.172	.181	.203	.226	.248	.250	.225
.275	.258	.229	.184	.167	.177	.181	.184	.200	.217	.239	.275
.325	.242	.210	.168	.147	.182	.191	.171	.179	.205	.219	.325
.375		.193	.149	.139	.152	.166	.154	.163	.190	.199	.375
.425		.178	.136	.120	.143	.152	.139	.149	.174	.181	.425
.475	.191	.170	.122	.110	.134	.158	.127	.139	.156	.159	.475
.550	.171	.145	.101	.089	.107	.109	.107	.108	.115	.137	.550
.650	.152	.118	.071	.085	.076		.078		.100	.115	.650
.750	.134	.103	.086	.045	.071	.096	.066	.065	.084	.092	.750
.800	.121									.092	.800
.850		.093		.050	.075	.051	.051	.066	.073		.850
.900			.078	.041			.058	.066			.900
.950					.037	.044					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 10.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.010	.038	.033	.128	.131	.080	.051	.0
22.5				.027	.024	.022	.115	.126	.068	.038	22.5
45.0				.002	-.022	.029		.105	.033	.000	45.0
67.5				-.062	-.065	.125	.112		-.019	-.055	67.5
90.0				-.097				-.007	-.049	-.066	90.0
112.5				-.098	.066	.217	.110	-.016	-.047	-.035	112.5
135.0				-.085	.314	.196	.086	-.050	-.037	-.023	135.0
157.5				-.055				-.027		-.014	157.5
180.0				-.052				-.020	-.007		180.0
202.5				-.062				-.029		-.009	202.5
225.0				-.080	.351	.232	.096	-.066	-.041	-.023	225.0
247.5				-.107	.080	.246	.115	-.037	-.049	-.037	247.5
270.0				-.090				-.009	-.049		270.0
292.5				-.054	-.072	.134	.118	.054	-.015	-.061	292.5
315.0				-.009	-.022	.079	.118	.103	.035	.000	315.0
337.5				.027			.119	.119		.028	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.207	-.200	-.130	-.162	-.150	-.186	-.190	-.183		.025
.075		-.166	-.180	-.163	-.158	-.159	-.185	-.181	-.171	-.184	.075
.125	-.152	-.173	-.180	-.166	-.152	-.154	-.181	-.168	-.170	-.182	.125
.175	-.155	-.173	-.180	-.159	-.145	-.130	-.195	-.181	-.173	-.179	.175
.225	-.155	-.178	-.180	-.175	-.155	-.166	-.200	-.192	-.173	-.179	.225
.275	-.159	-.161	-.180	-.186		-.166	-.200	-.186	-.174	-.181	.275
.325	-.179	-.182	-.191		-.110	-.152	-.187	-.203	-.178	-.168	.325
.375	-.166	-.182	-.191	-.182	-.092	-.108	-.194	-.195	-.170	-.173	.375
.425	-.184	-.187	-.186	-.173	-.092	-.108	-.184	-.206	-.179		.425
.475	-.163	-.187	-.198	-.185	-.100	-.113	-.184	-.205	-.182	-.179	.475
.550	-.166	-.182	-.198	-.190	-.113	-.120	-.200	-.201	-.180	-.170	.550
.650	-.165	-.186	-.193	-.192		-.131	-.197	-.203	-.176	-.167	.650
.750	-.160	-.181	-.193	-.184	-.104	-.140	-.194		-.168	-.164	.750
.800	-.163									-.162	.800
.850		-.181	-.193	-.180	-.131	-.140	-.200	-.173	-.170		.850
.900			-.185	-.190			-.190	-.179			.900
.950					-.135	-.140					.950
LOWER SURFACE											
.025		.492	.476	.491	.482	.469	.508	.520	.519		.025
.075	.427	.422	.396	.382	.308	.303	.401	.422	.437		.075
.125	.396	.381	.339	.332	.224	.245	.341	.366	.390	.402	.125
.175	.369	.347	.303	.281	.205	.236	.297	.319	.356	.366	.175
.225	.350	.315	.270	.255	.232	.245	.270	.299	.322	.335	.225
.275	.326	.293	.248	.235	.248	.268	.244	.273	.299	.320	.275
.325	.305	.269	.235	.212	.239	.258	.232	.252	.276	.299	.325
.375		.257	.207	.203	.224	.243	.214	.238	.264	.270	.375
.425		.243	.199	.182	.205	.224	.207	.223	.249	.260	.425
.475	.255	.229	.182	.179	.191	.212	.193	.210	.224	.238	.475
.550	.231	.209	.159	.153	.166	.171	.165	.181	.187	.214	.550
.650	.212	.170	.129	.143	.126		.136		.171	.192	.650
.750	.190	.158	.142	.091	.110	.145	.116	.129	.158	.172	.750
.800	.183									.172	.800
.850		.149		.092	.121	.101	.105	.128	.146		.850
.900			.127	.094			.105	.117			.900
.950					.079	.085					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 12.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.038	.072	.061	.161	.186	.116	.094	.0
22.5				.050	.051	.045	.150	.172	.101	.071	22.5
45.0				.015	-.009	.003		.143	.051	.019	45.0
67.5				-.064	-.080	.137	.151		-.015	-.070	67.5
90.0				-.126				-.003	-.070	-.114	90.0
112.5				-.136	.029	.270	.175	-.010	-.057	-.034	112.5
135.0				-.093	.324	.249	.149	-.040	-.038	-.027	135.0
157.5				-.064				-.020		-.014	157.5
180.0				-.050				-.008	.012		180.0
202.5				-.064				-.017		-.014	202.5
225.0				-.091	.395	.309	.164	-.069	-.036	-.027	225.0
247.5				-.140	.031	.296	.188	-.023	-.055	-.045	247.5
270.0				-.118				-.006	-.057		270.0
292.5				-.068	-.083	.151	.157	.070	-.008	-.075	292.5
315.0				-.006	-.006	.101	.156	.139	.057	.022	315.0
337.5				.048	.047	.049	.156	.160		.059	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.252	-.244	-.160	-.202	-.193	-.234	-.239	-.234		.025
.075		-.213	-.217	-.205	-.185	-.185	-.234	-.227	-.222	-.228	.075
.125	-.204	-.221	-.213	-.201	-.180	-.174	-.225	-.213	-.219	-.228	.125
.175	-.202	-.217	-.215	-.189	-.172	-.157	-.231	-.225	-.217	-.220	.175
.225	-.202	-.217	-.215	-.205	-.182	-.197	-.236	-.231	-.218	-.218	.225
.275	-.202	-.197	-.211	-.211		-.197	-.225	-.227	-.219	-.214	.275
.325	-.212	-.217	-.226		-.123	-.180	-.218	-.240	-.217	-.193	.325
.375	-.194	-.217	-.224	-.207	-.103	-.144	-.223	-.236	-.208	-.189	.375
.425	-.204	-.217	-.215	-.195	-.096	-.123	-.213	-.243	-.211		.425
.475	-.193	-.214	-.227	-.208	-.110	-.123	-.213	-.241	-.205	-.189	.475
.550	-.189	-.207	-.219	-.211	-.120	-.133	-.227	-.226	-.199	-.185	.550
.650	-.191	-.205	-.206	-.223		-.141	-.227	-.223	-.199	-.189	.650
.750	-.185	-.205	-.206	-.200	-.116	-.152	-.221		-.196	-.189	.750
.800	-.192									-.187	.800
.850		-.205	-.212	-.208	-.141	-.157	-.231	-.206	-.197		.850
.900			-.205	-.217			-.220	-.209			.900
.950					-.147	-.157					.950
LOWER SURFACE											
.025		.558	.543	.559	.567	.548	.582	.580	.582		.025
.075	.493	.493	.466	.455	.367	.367	.473	.490	.506		.075
.125	.462	.451	.408	.403	.242	.269	.411	.430	.458	.460	.125
.175	.433	.415	.368	.351	.220	.261	.367	.387	.422	.431	.175
.225	.419	.385	.342	.321	.280	.296	.344	.366	.392	.400	.225
.275	.390	.361	.317	.302	.297	.329	.325	.340	.358	.337	.275
.325	.366	.339	.302	.284	.281	.304	.302	.316	.341	.364	.325
.375		.320	.280	.265	.276	.295	.280	.299	.321	.340	.375
.425		.305	.261	.251	.259	.276	.267	.283	.302	.317	.425
.475	.317	.291	.239	.242	.242	.267	.252	.271	.283	.294	.475
.550	.295	.264	.214	.197	.225	.235	.225	.235	.246	.271	.550
.650	.270	.228	.186	.182	.181		.179		.219	.245	.650
.750	.251	.214		.140	.168	.197	.163	.174	.210	.229	.750
.800	.239									.229	.800
.850		.200		.140	.168	.159	.158	.174	.199		.850
.900			.172	.130			.158	.168			.900
.950					.126	.145					.950

TABLE 3. Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 15.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.068	.104	.089	.183	.221	.146	.122	.0
22.5				.083	.071	.071	.163	.208	.126	.094	22.5
45.0				.031	-.008	.001		.158	.069	.038	45.0
67.5				-.078	-.101	.169	.176		-.020	-.077	67.5
90.0				-.150				.008	-.093	-.177	90.0
112.5				-.184	-.031	.283	.250	-.051	-.082	-.066	112.5
135.0				-.115	.146	.319	.221	-.051	-.045	-.038	135.0
157.5				-.076				-.030		-.022	157.5
180.0				-.051				-.006	.010		180.0
202.5				-.080				-.027		-.036	202.5
225.0				-.102	.257	.385	.226	-.064	-.047	-.043	225.0
247.5				-.189	-.023	.353	.252	-.063	-.076	-.066	247.5
270.0				-.140				-.008	-.092		270.0
292.5				-.073	-.097	.183	.191	.089	-.005	-.085	292.5
315.0				.015	.002	.051	.183	.161	.076	.035	315.0
337.5				.083	.066	.068	.172	.197		.089	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.269	-.267	-.181	-.240	-.243	-.270	-.279	-.263		.025
.075		-.240	-.244	-.245	-.224	-.232	-.270	-.264	-.255	-.244	.075
.125	-.221	-.250	-.244	-.232	-.224	-.218	-.257	-.251	-.250	-.241	.125
.175	-.221	-.244	-.244	-.226	-.212	-.201	-.257	-.263	-.247	-.239	.175
.225	-.220	-.243	-.244	-.237	-.219	-.232	-.257	-.269	-.245	-.237	.225
.275	-.217	-.228	-.231	-.246		-.220	-.257	-.263	-.244	-.236	.275
.325	-.231	-.240	-.253		-.151	-.192	-.247	-.269	-.238	-.219	.325
.375	-.218	-.238	-.252	-.244	-.136	-.159	-.256	-.265	-.225	-.216	.375
.425	-.228	-.238	-.220	-.233	-.131	-.153	-.244	-.265	-.222		.425
.475	-.209	-.228	-.250	-.245	-.142	-.149	-.244	-.259	-.222	-.224	.475
.550	-.209	-.220	-.237	-.250	-.150	-.157	-.257	-.249	-.216	-.217	.550
.650	-.209	-.226	-.235	-.252		-.168	-.257	-.252	-.222	-.228	.650
.750	-.209	-.222	-.235	-.233	-.163	-.180	-.257		-.217	-.226	.750
.800	-.222									-.225	.800
.850		-.226	-.243	-.233	-.191	-.186	-.257	-.227	-.230		.850
.900			-.232	-.249			-.246	-.223			.900
.950					-.191	-.192					.950
LOWER SURFACE											
.025		.612	.609	.630	.631	.621	.643	.635	.629		.025
.075	.550	.552	.538	.521	.415	.400	.543	.548	.563		.075
.125	.522	.511	.480	.467	.268	.245	.480	.496	.516	.519	.125
.175	.493	.474	.439	.417	.180	.193	.436	.450	.479	.492	.175
.225	.476	.444	.411	.385	.206	.269	.406	.425	.445	.462	.225
.275	.449	.421	.393	.362	.297	.327	.381	.396	.421	.445	.275
.325	.430	.399	.374	.346	.301	.325	.358	.376	.403	.419	.325
.375		.381	.344	.332	.310	.339	.340	.358	.385	.397	.375
.425		.364	.328	.306	.310	.332	.327	.346	.366	.370	.425
.475	.373	.348	.308	.294	.300	.320	.309	.327	.345	.355	.475
.550	.346	.322	.278	.256	.291	.296	.277	.291	.303	.325	.550
.650	.319	.279	.238	.246	.245		.239		.275	.300	.650
.750	.305	.272		.203	.238	.249	.223	.230	.264	.287	.750
.800	.295									.287	.800
.850		.259		.194	.238	.223	.214	.230	.255		.850
.900			.222	.194			.214	.223			.900
.950					.197	.206					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.068	-.046	-.026	-.013	-.055	-.062	-.058	.0
22.5				-.049	-.039	-.012	-.027	-.046	-.049	-.054	22.5
45.0				-.018	-.028	.016	-.021	-.033	-.030	-.047	45.0
67.5				-.020	-.002	.021	-.014		-.015	-.018	67.5
90.0				-.009				.002	-.009	-.012	90.0
112.5				-.013	.056	.026	-.050	.011	.000	.018	112.5
135.0				-.022	.207	-.002	-.049	.011	.005	.004	135.0
157.5				-.033				.001	.001	-.008	157.5
180.0				-.041				.002	-.005		180.0
202.5				-.060				.006	-.005	-.007	202.5
225.0				-.060	-.048	-.056	-.049	-.004		.000	225.0
247.5				-.055	.006	-.048	-.067	-.008	-.008	-.008	247.5
270.0				-.044				-.014	-.016		270.0
292.5				-.051	-.015	-.018	-.037	-.041	-.028	-.021	292.5
315.0				-.058	-.026	-.001	-.027	-.063	-.035	-.025	315.0
337.5				-.062	-.033	-.002	-.018	-.056		-.041	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		.187	.180	.139	.015	.116	.131	.120	.123		.025
.075		.154	.136	.103	-.017	.081	.089	.087	.097	.080	.075
.125	.140	.123	.102	.063	-.045	.049	.056	.063	.077	.066	.125
.175	.110	.096	.075	.036	-.050	.037	.030	.037	.052	.054	.175
.225	.094	.072	.057	.017	-.064	.033	.023	.019	.036	.046	.225
.275	.076	.056	.036	-.009	-.064	.033	.011	.012	.027	.029	.275
.325	.058	.043	.019	-.021	-.066	.021	.008	.002	.016	.025	.325
.375	.050	.034	.007	-.037	-.064	.017	.002	-.006	.008	.019	.375
.425	.036	.024	.007	-.037	-.055	.008	-.011	-.021	-.001		.425
.475	.028	.012	-.009	-.051	-.055	-.006	-.011	-.021	-.008	-.007	.475
.550	.009	-.008	-.028	-.066	-.055	-.015	-.032	-.037	-.020	-.016	.550
.650	-.008	-.036	-.059	-.088		-.031	-.032	-.052	-.039	-.032	.650
.750	-.023	-.046	-.064	-.088	-.056	-.039	-.049	-.042	-.046	-.044	.750
.800	-.033									-.046	.800
.850		-.046	-.074	-.088	-.037	-.033	-.050	-.043	-.052		.850
.900			-.074	-.089			-.049	-.045			.900
.950					-.030	-.033					.950
LOWER SURFACE											
.025		.173	.149	.178	.214	-.106	.004	.009	.020		.025
.075	.122	.130	.102	.132	.152	-.091	-.047	.000	-.008		.075
.125	.103	.103	.064	.091	.106	-.087	-.070	-.017	-.002	-.021	.125
.175	.084	.076	.051	.058	.063	-.108	-.104		-.020	-.034	.175
.225	.074	.056	.038	.031	.063	-.086	-.095	-.051	-.033	-.051	.225
.275	.057	.043	.013	.014	.050	-.079	-.083	-.065	-.074	-.053	.275
.325	.037	.032	.012	-.009	.028	-.079	-.083	-.076	-.060	-.043	.325
.375		.014	-.019	-.009	.005	-.080	-.090	-.082	-.067	-.084	.375
.425		.006	-.028	-.028	.005	-.093	-.104		-.076	-.061	.425
.475	.001	-.002	-.028	-.039	-.009	-.082	-.089	-.093	-.086	-.103	.475
.550	-.016	-.018	-.054	-.050	-.026	-.082	-.098	-.124	-.117	-.110	.550
.650	-.019	-.032	-.072	-.060	-.056		-.106		-.113	-.118	.650
.750	-.037	-.050	-.051	-.095	-.083	-.100	-.119	-.139	-.128	-.121	.750
.800	-.038									-.105	.800
.850		-.048	-.065	-.085	-.076	-.092	-.111	-.122	-.128		.850
.900			-.065	-.095			-.097	-.130			.900
.950					-.084	-.082					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

HIGH – WING CONFIGURATION

$\alpha = 2.5°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.043	-.029	-.020	.022	-.005	-.014	-.022	.0
22.5				-.021	-.014	.003	.022	.002	.000	-.012	22.5
45.0				.006	.000	.070	.035	.014	.019	.007	45.0
67.5				-.001	.015	.083	.035		.028	.024	67.5
90.0				-.008				.029	.028	.017	90.0
112.5				-.021	.106	.096	.000	.022	.016	.031	112.5
135.0				-.035	.312	.073	-.008	.013	.016	.019	135.0
157.5				-.040				.007		.015	157.5
180.0				-.049				.007	.026		180.0
202.5				-.053				.013	.023	.017	202.5
225.0				-.048	.020	.003	-.023	.009	.020	.017	225.0
247.5				-.041	.035	.010	-.020	.017	.015	.017	247.5
270.0				-.040				.009	.005		270.0
292.5				-.049	-.022	.028	.009	-.002	-.006	.000	292.5
315.0				-.059	-.037	.020	.010	-.012	-.010	-.015	315.0
337.5				-.049	-.037	-.001	.031	-.006	-.022	-.027	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.056	.061	.038	-.052	.056	.017	-.009	-.004		.025
.075		.059	.046	.023	-.075	.040	.005	-.019	-.011	-.029	.075
.125	.055	.037	.019	-.007	-.097	.014	-.013	-.025	-.018	-.035	.125
.175	.039	.017	-.001	-.023	-.097	.008	-.031	-.043	-.029	-.040	.175
.225	.025	-.001	-.018	-.050	-.106	-.005	-.044	-.052	-.036	-.041	.225
.275	.009	.000	-.031	-.064	-.093	-.005	-.044	-.052	-.044	-.050	.275
.325	-.009	-.025	-.042	-.071	-.093	-.012	-.044	-.064	-.049	-.050	.325
.375	-.017	-.034	-.053	-.082	-.093	-.019	-.050	-.064	-.052	-.048	.375
.425	-.030	-.045	-.053	-.082	-.088	-.031	-.050	-.074	-.060		.425
.475	-.031	-.051	-.071	-.096	-.087	-.037	-.053	-.083	-.066	-.073	.475
.550	-.049	-.065	-.081	-.109	-.085	-.050	-.070	-.083	-.074	-.082	.550
.650	-.063	-.087	-.098	-.133		-.064	-.074	-.097	-.092	-.097	.650
.750	-.072	-.097	-.107	-.128	-.100	-.070	-.087	-.082	-.095	-.099	.750
.800	-.075									-.096	.800
.850		-.096	-.117	-.123	-.072	-.071	-.093	-.077	-.099		.850
.900			-.107	-.117			-.087	-.078			.900
.950					-.059	-.066					.950
LOWER SURFACE											
.025		.287	.280	.314	.015	.143	.176	.187			.025
.075	.219	.232	.230	.235	.246	.017	.076	.125	.135		.075
.125	.195	.199	.175	.185	.200	.006	.045	.086	.108	.108	.125
.175	.173	.167	.150	.151	.159	-.015	.014		.083	.089	.175
.225	.163	.148	.129	.125	.139	-.009	.013	.045	.063	.056	.225
.275	.139	.130	.105	.103	.131	.002	.013	.030	.026	.052	.275
.325	.122	.114	.096	.077	.102	-.006	.006	.019	.035	.052	.325
.375		.095	.074	.072	.082	-.006	-.004	.012	.026	.024	.375
.425		.084	.065	.053	.077	-.020	-.012	-.013	.018	.024	.425
.475	.081	.076	.056	.046	.059	-.017	-.008	-.008	.002	-.011	.475
.550	.063	.058	.032	.020	.041	-.020	-.024	-.032	-.031	-.020	.550
.650	.050	.032	.008	.011	.009		-.037		-.034	-.035	.650
.750	.029	.019	.007	-.024	-.032	-.046	-.051	-.065	-.053	-.047	.750
.800	.018									-.033	.800
.850		.013	-.002	-.022	-.024	-.038	-.043	-.053	-.053		.850
.900			-.007	-.030			-.040	-.063			.900
.950					-.035	-.039					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 5.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.034	-.025	-.018	.034	.008	-.018	-.030	.0
22.5				-.004	.004	.015	.036	.027	-.001	-.011	22.5
45.0				.025	.019	.068	.058	.040	.004	-.006	45.0
67.5				.004	.011	.130	.076		.002	.006	67.5
90.0				-.013				.016	-.008	-.004	90.0
112.5				-.042	.159	.154	.054	-.004	-.016	.000	112.5
135.0				-.071	.392	.124	.042	-.015	-.018	-.005	135.0
157.5				-.067				-.019	-.011	-.007	157.5
180.0				-.060				-.009	-.002		180.0
202.5				-.051				-.022	-.011	.001	202.5
225.0				-.046	.083	.041	-.001	-.016		-.013	225.0
247.5				-.050	.050	.053	.020	-.013	-.013	-.020	247.5
270.0				-.060				-.018	-.020		270.0
292.5				-.070	-.050	.035	.023	-.013	-.027	-.029	292.5
315.0				-.069	-.064	.018	.039	-.008	-.040	-.054	315.0
337.5				-.053	-.053	-.032	.041	-.001		-.049	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.040	-.039	-.047	-.116	.002	-.085	-.109	-.103		.025
.075		-.019	-.040	-.059	-.128	-.008	-.085	-.102	-.095	-.112	.075
.125	-.014	-.031	-.052	-.071	-.141	-.020	-.085	-.097	-.095	-.113	.125
.175	-.027	-.045	-.062	-.077	-.141	-.020	-.093	-.112	-.103	-.111	.175
.225	-.038	-.058	-.076	-.099	-.141	-.039	-.093	-.116	-.106	-.114	.225
.275	-.050	-.058	-.088	-.115	-.125	-.039	-.093	-.119	-.112	-.121	.275
.325	-.063	-.076	-.097	-.121	-.131	-.039	-.085	-.131	-.113	-.114	.325
.375	-.064	-.084	-.106	-.127	-.129	-.049	-.093	-.122	-.111	-.107	.375
.425	-.082	-.093	-.102	-.127	-.127	-.055	-.094	-.123	-.118		.425
.475	-.082	-.100	-.116	-.144	-.132	-.065	-.094	-.133	-.120	-.130	.475
.550	-.094	-.109	-.112	-.154	-.132	-.076	-.102	-.131	-.126	-.136	.550
.650	-.107	-.132	-.141	-.171		-.094	-.102	-.144	-.143	-.144	.650
.750	-.102	-.140	-.151	-.153	-.106	-.103	-.114	-.116	-.137	-.129	.750
.800	-.096									-.122	.800
.850		-.127	-.144	-.146	-.089	-.099	-.115	-.107	-.129		.850
.900			-.131	-.146			-.109	-.103			.900
.950					-.090	-.099					.950
LOWER SURFACE											
.025		.398	.399	.423	.442	.119	.259	.268	.271		.025
.075	.318	.331	.325	.324	.324	.082	.174	.194	.215		.075
.125	.290	.286	.265	.265	.266	.063	.128	.152	.173	.180	.125
.175	.259	.249	.230	.222	.234	.044	.096		.149	.156	.175
.225	.244	.224	.207	.196	.215	.040	.086	.110	.128	.126	.225
.275	.222	.205	.180	.165	.204	.060	.082	.093	.093	.119	.275
.325	.196	.186	.162	.143	.170	.046	.061	.079	.093	.111	.325
.375		.166	.137	.139	.160	.052	.061	.070	.079	.085	.375
.425		.155	.131	.116	.149	.041	.043	.052	.072	.078	.425
.475	.154	.144	.117	.106	.115	.041	.043	.048	.060	.051	.475
.550	.134	.118	.089	.080	.103	.039	.021	.022	.026	.034	.550
.650	.117	.093	.087	.065	.060		.011		.014	.020	.650
.750	.092	.074	.058	.021	.017	-.007	-.006	-.013	.007	.009	.750
.800	.078									.020	.800
.850		.067	.047	.021	.018	.004	.001	.007	.000		.850
.900			.042	.021			.006	-.005			.900
.950					.015	.005					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.014	-.006	.007	.044	.038	.009	-.012	.0
22.5				.030	.024	.049	.068	.059	.033	.008	22.5
45.0				.049	.033	.101	.100	.069	.021	.007	45.0
67.5				.010	.007	.173	.108		.001	-.005	67.5
90.0				-.028				.013	-.012	-.020	90.0
112.5				-.062	.159	.223	.098	-.015	-.034	-.021	112.5
135.0				-.099	.494	.187	.086	-.034	-.028	-.014	135.0
157.5				-.080				-.038	-.015	-.006	157.5
180.0				-.062				-.017	.005		180.0
202.5				-.048				-.026	-.016	-.010	202.5
225.0				-.043	.118	.100	.047	-.023		-.022	225.0
247.5				-.058	.035	.111	.061	-.015	-.023	-.029	247.5
270.0				-.078				-.016	-.026		270.0
292.5				-.090	-.092	.059	.022	.008	-.033	-.054	292.5
315.0				-.077	-.086	.019	.068	.028	-.035	-.061	315.0
337.5				-.043	-.049	-.048	.064	.024		-.051	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.123	-.119	-.119	-.167	-.047	-.166	-.186	-.179		.025
.075		-.088	-.102	-.119	-.167	-.047	-.156	-.171	-.168	-.182	.075
.125	-.077	-.091	-.102	-.124	-.182	-.047	-.156	-.166	-.163	-.181	.125
.175	-.084	-.100	-.114	-.124	-.171	-.047	-.156	-.181	-.167	-.174	.175
.225	-.092	-.109	-.123	-.145	-.171	-.068	-.153	-.178	-.166	-.176	.225
.275	-.092	-.104	-.129	-.159	-.154	-.068	-.147	-.178	-.165	-.177	.275
.325	-.107	-.119	-.136	-.164	-.161	-.068	-.140	-.188	-.168	-.169	.325
.375	-.107	-.126	-.145	-.167	-.161	-.068	-.140	-.182	-.167	-.161	.375
.425	-.126	-.135	-.141	-.168	-.161	-.077	-.140	-.192	-.170		.425
.475	-.126	-.140	-.156	-.178	-.161	-.085	-.140	-.185	-.173	-.179	.475
.550	-.133	-.148	-.143	-.192	-.167	-.100	-.153	-.177	-.175	-.168	.550
.650	-.133	-.168	-.172	-.187		-.113	-.129	-.190	-.174	-.162	.650
.750	-.120	-.168	-.158	-.166	-.128	-.121	-.137	-.161	-.163	-.158	.750
.800	-.122									-.155	.800
.850		-.150	-.154	-.166	-.103	-.121	-.145	-.152	-.162		.850
.900			-.149	-.167			-.139	-.148			.900
.950					-.102	-.121					.950
LOWER SURFACE											
.025		.498	.493	.521	.528	.202	.339	.350	.343		.025
.075	.413	.421	.410	.399	.371	.132	.245	.271	.286		.075
.125	.380	.371	.346	.345	.319	.096	.200	.226	.246	.246	.125
.175	.347	.335	.305	.290	.290	.083	.157		.212	.225	.175
.225	.331	.301	.275	.262	.286	.073	.143	.165	.189	.190	.225
.275	.305	.279	.251	.236	.272	.104	.132	.156	.152	.183	.275
.325	.281	.259	.233	.213	.236	.096	.123	.137	.155	.174	.325
.375		.232	.203	.197	.227	.098	.113	.125	.139	.149	.375
.425		.220	.189	.174	.208	.090	.093	.105	.130	.141	.425
.475	.225	.207	.177	.161	.183	.092	.090	.105	.116	.113	.475
.550	.202	.183	.142	.131	.168	.079	.077	.069	.079	.091	.550
.650	.180	.150	.112	.124	.113		.056		.067	.073	.650
.750	.156	.132	.113	.067	.064	.039	.034	.037	.056	.059	.750
.800	.139									.072	.800
.850		.122	.103	.071	.067	.045	.033	.040	.053		.850
.900			.096	.066			.040	.031			.900
.950					.056	.044					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 10.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.013	.035	.020	.064	.069	.040	.014	.0
22.5				.059	.055	.071	.121	.098	.064	.036	22.5
45.0				.063	.048	.105	.141	.113	.050	.027	45.0
67.5				.008	.001	.197	.147		.006	-.003	67.5
90.0				-.043				.034	-.043	-.044	90.0
112.5				-.098	.149	.295	.159	-.023	-.052	-.031	112.5
135.0				-.135	.556	.260	.146	-.066	-.035	-.019	135.0
157.5				-.094				-.043	-.001	-.006	157.5
180.0				-.068				-.017	.001		180.0
202.5				-.048				-.021	-.022	-.010	202.5
225.0				-.057	.149	.183	.094	-.045		-.024	225.0
247.5				-.078	.014	.181	.106	-.019	-.031	-.034	247.5
270.0				-.117				-.006	-.035		270.0
292.5				-.119	-.153	.065	.056	.042	-.041	-.107	292.5
315.0				-.083	-.103	-.015	.092	.062	-.035	-.064	315.0
337.5				-.026	-.047	-.043	.092	.052		-.037	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.190	-.188	-.175	-.207	-.119	-.219	-.243	-.231		.025
.075		-.152	-.165	-.169	-.210	-.101	-.212	-.230	-.216	-.228	.075
.125	-.142	-.159	-.167	-.169	-.210	-.094	-.200	-.219	-.209	-.221	.125
.175	-.142	-.155	-.171	-.169	-.199	-.084	-.196	-.219	-.207	-.216	.175
.225	-.142	-.167	-.174	-.184	-.199	-.084	-.190	-.226	-.207	-.219	.225
.275	-.149	-.155	-.178	-.198	-.193	-.079	-.192	-.219	-.208	-.216	.275
.325	-.171	-.168	-.186	-.203	-.201	-.078	-.188	-.224	-.207	-.200	.325
.375	-.160	-.175	-.191	-.203	-.201	-.084	-.188	-.224	-.199	-.184	.375
.425	-.168	-.175	-.185	-.205	-.198	-.094	-.187	-.224	-.204		.425
.475	-.168	-.184	-.199	-.207	-.196	-.101	-.186	-.217	-.205	-.192	.475
.550	-.177	-.185	-.193	-.207	-.191	-.113	-.203	-.212	-.196	-.182	.550
.650	-.169	-.190	-.191	-.205		-.130	-.146	-.219	-.191	-.188	.650
.750	-.158	-.182	-.180	-.193	-.139	-.139	-.145	-.196	-.187	-.185	.750
.800	-.158									-.187	.800
.850		-.177	-.180	-.193	-.120	-.139	-.156	-.184	-.194		.850
.900			-.177	-.193			-.156	-.179			.900
.950					-.121	-.139					.950
LOWER SURFACE											
.025		.586	.582	.602	.597	.305	.424	.422	.409		.025
.075	.496	.502	.487	.480	.439	.199	.327	.340	.350		.075
.125	.460	.452	.418	.415	.369	.139	.266	.297	.310	.310	.125
.175	.425	.413	.381	.366	.339	.106	.229		.279	.281	.175
.225	.410	.374	.352	.331	.352	.104	.216	.240	.260	.253	.225
.275	.379	.353	.322	.306	.342	.149	.203	.219	.221	.240	.275
.325	.351	.329	.300	.274	.298	.153	.188	.201	.216	.231	.325
.375		.299	.270	.265	.298	.170	.174	.189	.206	.206	.375
.425		.286	.252	.236	.273	.150	.158	.164	.193	.200	.425
.475	.292	.274	.229	.222	.248	.150	.151	.164	.173	.162	.475
.550	.267	.244	.199	.196	.225	.149	.138	.128	.132	.148	.550
.650	.246	.206	.174	.179	.171		.109		.125	.126	.650
.750	.220	.188	.168	.124	.118	.083	.085	.090	.116	.118	.750
.800	.205									.130	.800
.850		.179		.123	.118	.087	.085	.091	.106		.850
.900			.152	.108			.085	.084			.900
.950					.106	.087					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

HIGH-WING CONFIGURATION

$\alpha = 12.5°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.036	.056	.047	.100	.104	.078	.043	.0
22.5				.089	.081	.091	.149	.133	.090	.067	22.5
45.0				.081	.051	.114	.175	.139	.076	.044	45.0
67.5				.008	-.007	.201	.196		.000	-.026	67.5
90.0				-.075				.044	-.065	-.088	90.0
112.5				-.139	.109	.361	.214	-.082	-.099	-.050	112.5
135.0				-.181	.572	.329	.200	-.088	-.054	-.029	135.0
157.5				-.106				-.036	-.008	-.027	157.5
180.0				-.078				-.020	-.015		180.0
202.5				-.056				-.027	-.025	-.029	202.5
225.0				-.076	.142	.237	.117	-.060		-.044	225.0
247.5				-.100	-.035	.252	.130	-.012	-.042	-.048	247.5
270.0				-.173				-.001	-.050		270.0
292.5				-.151	-.177	.077	.069	.057	-.058	-.137	292.5
315.0				-.086	-.104	-.021	.081	.089	-.032	-.070	315.0
337.5				-.008	-.022	-.026	.097	.081		-.022	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.237	-.235	-.228	-.259	-.187	-.260	-.287	-.261		.025
.075		-.202	-.212	-.216	-.248	-.158	-.250	-.271	-.253	-.244	.075
.125	-.193	-.198	-.204	-.216	-.236	-.133	-.230	-.259	-.246	-.245	.125
.175	-.193	-.198	-.204	-.216	-.231	-.090	-.230	-.259	-.242	-.240	.175
.225	-.193	-.198	-.214	-.228	-.235	-.096	-.230	-.259	-.240	-.240	.225
.275	-.193	-.190	-.211	-.235	-.224	-.095	-.230	-.253	-.239	-.240	.275
.325	-.205	-.205	-.221	-.228	-.237	-.094	-.230	-.261	-.237	-.223	.325
.375	-.193	-.205	-.220	-.224	-.236	-.102	-.230	-.252	-.228	-.215	.375
.425	-.203	-.214	-.214	-.224	-.224	-.109	-.224	-.252	-.227		.425
.475	-.193	-.214	-.225	-.224	-.217	-.122	-.228	-.252	-.222	-.229	.475
.550	-.184	-.199	-.222	-.229	-.199	-.136	-.239	-.244	-.215	-.230	.550
.650	-.184	-.199	-.212	-.229		-.155	-.208	-.254	-.218	-.241	.650
.750	-.182	-.197	-.205	-.218	-.169	-.166	-.193	-.224	-.218	-.238	.750
.800	-.182									-.239	.800
.850		-.191	-.211	-.220	-.166	-.166	-.171	-.211	-.231		.850
.900			-.211	-.220			-.165	-.202			.900
.950					-.165	-.161					.950
LOWER SURFACE											
.025		.660	.647	.661	.638	.349	.489	.479	.463		.025
.075	.571	.578	.557	.547	.465	.264	.395	.407	.414		.075
.125	.533	.525	.489	.480	.376	.170	.344	.358	.379	.369	.125
.175	.499	.482	.445	.434	.358	.092	.297		.350	.355	.175
.225	.480	.445	.406	.392	.387	.075	.276	.300	.323	.325	.225
.275	.449	.418	.377	.363	.377	.155	.260	.278	.287	.313	.275
.325	.423	.394	.357	.330	.356	.177	.238	.261	.279	.302	.325
.375		.365	.330	.312	.350	.205	.221	.247	.260	.274	.375
.425		.354	.318	.293	.324	.200	.209	.227	.250	.261	.425
.475	.362	.333	.296	.290	.308	.193	.202	.221	.235	.234	.475
.550	.335	.308	.265	.240	.280	.179	.183	.187	.187	.213	.550
.650	.312	.268	.220	.230	.221		.150		.182	.196	.650
.750	.285	.252	.223	.181	.168	.108	.129	.136	.167	.183	.750
.800	.266									.187	.800
.850		.238		.170	.169	.108	.124	.141	.160		.850
.900			.204	.161			.124	.132			.900
.950					.164	.108					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 15.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.074	.091	.067	.138	.140	.104	.079	.0
22.5				.125	.117	.118	.192	.176	.126	.095	22.5
45.0				.104	.071	.144	.221	.177	.097	.069	45.0
67.5				.008	-.007	.237	.232		.020	-.025	67.5
90.0				-.081				.065	-.097	-.146	90.0
112.5				-.162	.072	.443	.288	-.095	-.130	-.051	112.5
135.0				-.209	.493	.403	.264	-.120	-.079	-.060	135.0
157.5				-.112				-.033	-.021	-.048	157.5
180.0				-.091				-.004	-.025		180.0
202.5				-.063				-.033	-.046	-.046	202.5
225.0				-.089	-.011	.269	.169	-.065		-.050	225.0
247.5				-.130	-.096	.258	.177	-.002	-.057	-.057	247.5
270.0				-.211				.000	-.079		270.0
292.5				-.152	-.182	.067	.076	.057	-.085	-.145	292.5
315.0				-.074	-.096	-.048	.121	.091	-.027	-.056	315.0
337.5				.023	.018	-.001	.118	.097		.004	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.268	-.272	-.248	-.274	-.215	-.290	-.297	-.281		.025
.075		-.233	-.247	-.247	-.256	-.215	-.273	-.288	-.275	-.259	.075
.125	-.225	-.233	-.233	-.239	-.258	-.142	-.259	-.282	-.268	-.260	.125
.175	-.215	-.233	-.233	-.230	-.258	-.106	-.267	-.280	-.270	-.258	.175
.225	-.211	-.233	-.233	-.236	-.255	-.107	-.268	-.277	-.265	-.257	.225
.275	-.208	-.217	-.234	-.235	-.237	-.107	-.267	-.278	-.266	-.257	.275
.325	-.208	-.230	-.244	-.236	-.246	-.114	-.261	-.278	-.262	-.245	.325
.375	-.203	-.224	-.244	-.236	-.242	-.121	-.261	-.275	-.253	-.236	.375
.425	-.203	-.220	-.230	-.236	-.236	-.135	-.249	-.281	-.252		.428
.475	-.202	-.211	-.240	-.236	-.236	-.142	-.248	-.280	-.250	-.247	.475
.550	-.197	-.210	-.240	-.243	-.236	-.159	-.249	-.268	-.243	-.247	.550
.650	-.197	-.210	-.224	-.243		-.174	-.244	-.285	-.247	-.258	.650
.750	-.192	-.210	-.224	-.243	-.220	-.189	-.242	-.249	-.242	-.256	.750
.800	-.192									-.253	.800
.850		-.210	-.228	-.243	-.215	-.187	-.235	-.233	-.252		.850
.900			-.227	-.243			-.220	-.208			.900
.950					-.215	-.174					.950
LOWER SURFACE											
.025		.721	.718	.734	.751	.314	.549	.527	.501		.025
.075	.637	.642	.634	.618	.558	.139	.456	.461	.455		.075
.125	.604	.593	.559	.550	.417	.090	.403	.407	.418	.423	.125
.175	.570	.549	.525	.494	.383	.066	.359		.384	.395	.175
.225	.550	.510	.482	.460	.427	.030	.333	.359	.362	.361	.225
.275	.514	.483	.449	.432	.427	.080	.326	.333	.331	.351	.275
.325	.487	.455	.432	.401	.415	.154	.304	.314	.327	.339	.325
.375		.430	.400	.388	.415	.225	.284	.300	.314	.317	.375
.425		.413	.381	.361	.392	.225	.265	.277	.294	.306	.425
.475	.428	.393	.364	.346	.366	.225	.246	.267	.271	.275	.475
.550	.390	.362	.320	.303	.345	.216	.219	.227	.233	.253	.550
.650	.373	.322	.286	.292	.286		.199		.222	.240	.650
.750	.342	.306	.277	.235	.221	.150	.177	.177	.212	.232	.750
.800	.323									.241	.800
.850		.291		.225	.226	.150	.168	.187	.200		.850
.900			.251	.214			.167	.179			.900
.950					.216	.154					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.104	-.113	-.092	-.094	-.124	-.147	-.079	.0
22.5				-.066	-.088	-.066	-.110	-.101	-.109	-.111	22.5
45.0				-.001	-.027	.005	-.049	-.066	-.057	-.061	45.0
67.5				.026	.025	.046	-.006		-.016	-.007	67.5
90.0				.045				.012	.008	.002	90.0
112.5				.038	.137	.088	.000	.028	.009	.028	112.5
135.0				-.002	.362	.067	-.009	.033	.016	.013	135.0
157.5				-.052				.014		.007	157.5
180.0				-.092				-.002			180.0
202.5				-.110				.013	-.002	-.014	202.5
225.0				-.086	-.076	-.063	-.049	.005	.007	-.004	225.0
247.5				-.063	-.034	-.083	-.066	-.002	-.005	-.002	247.5
270.0				-.045				-.035	-.033		270.0
292.5				-.052	-.028	-.061	-.077	-.059	-.066	-.087	292.5
315.0				-.076	-.038	-.039	-.052	-.090	-.066	-.057	315.0
337.5				-.103	-.074	-.031	-.065	-.115		-.053	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.230	.221	.167	-.007	.125	.137	.105	.098		.025
.075	.192	.190	.168	.120	-.039	.089	.091	.075	.072	.055	.075
.125	.169	.155	.126	.080	-.064	.068	.068	.051	.058	.040	.125
.175	.143	.123	.097	.046	-.077	.052	.050	.033	.037	.033	.175
.225	.116	.099	.073	.023	-.093	.049	.037	.018	.027	.025	.225
.275	.107	.081	.062	.007	-.087	.036	.024	.011	.019	.015	.275
.325	.083	.069	.044	-.010	-.099	.026	.015	.005	.012	.010	.325
.375	.075	.056	.032	-.020	-.098	.018	.007	-.006	.006	-.001	.375
.425	.061	.045		-.030	-.103	.010	-.002	-.010	-.007		.425
.475	.051	.031	.014	-.045	-.100	-.002	-.008	-.020	-.009	-.020	.475
.550	.026	.010	-.011	-.064	-.110	-.004	-.026	-.036	-.019	-.030	.550
.650	.006		-.049	-.085		-.025	-.021	-.051	-.036	-.038	.650
.750	-.010	-.027	-.058	-.101	-.119	-.030	-.046	-.038	-.043	-.047	.750
.800	-.010									-.050	.800
.850		-.030	-.070	-.108	-.094	-.025	-.044	-.040			.850
.900			-.063	-.108			-.039	-.041			.900
.950					-.057	-.029					.950
LOWER SURFACE											
.025		.234	.223	.277	.395	-.170	-.038	-.033	-.031		.025
.075	.179	.186	.179	.219	.293	-.143	-.089	-.024	-.037	-.041	.075
.125	.155	.156	.139	.177	.240	-.099	-.102	-.044	-.028	-.047	.125
.175	.135	.126	.118	.137	.199	-.103	-.119	-.054	-.043	-.052	.175
.225	.126	.106	.102	.113	.170	-.089	-.099	-.066	-.050	-.070	.225
.275	.103	.089	.082	.095	.149	-.087	-.079	-.071	-.087	-.071	.275
.325	.085	.077	.071	.073	.115	-.092		-.078	-.074	-.063	.325
.375		.058	.052	.064	.096	-.097	-.077	-.078	-.074	-.091	.375
.425		.050	.040	.044	.080	-.104	-.092	-.098	-.080	-.082	.425
.475	.044	.040	.033	.032	.057	-.098	-.074	-.087	-.089	-.110	.475
.550	.027	.021	.006	.009	.035	-.105	-.080	-.103	-.113	-.116	.550
.650	.015	.005	-.014	-.005	.002		-.086		-.109	-.121	.650
.750	.003	-.008	-.011	-.033	-.035	-.106	-.096	-.116	-.110	-.123	.750
.800	.000									-.111	.800
.850		-.008	-.021	-.037	-.030	-.091	-.089	-.097	-.105		.850
.900			-.026	-.045			-.077	-.106			.900
.950					-.043	-.078					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 2.5°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.103	-.099	-.095	-.115	-.114	-.145	-.114	.0
22.5				-.043	-.052	-.025	-.070	-.088	-.092	-.087	22.5
45.0				.016	.004	.056	-.012	-.031	-.049	-.042	45.0
67.5				.038	.036	.092	.043		-.015	-.005	67.5
90.0				.042				.005	.005	.004	90.0
112.5				.016	.162	.145	.038	.014	-.009	.012	112.5
135.0				-.045	.443	.115	.023	.004	-.009	-.011	135.0
157.5				-.070				-.005		-.009	157.5
180.0				-.114				-.006			180.0
202.5				-.109				.002	.000	-.015	202.5
225.0				-.080	-.036	-.028	-.038	-.022	-.002	-.012	225.0
247.5				-.057	-.002	-.038	-.043	-.012	-.012	-.023	247.5
270.0				-.052				-.046	-.060		270.0
292.5				-.065	-.039	-.034	-.045	-.059	-.062	-.061	292.5
315.0				-.103	-.067	-.028	-.031	-.083	-.066	-.061	315.0
337.5				-.115	-.113	-.053	-.062	-.113		-.063	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.119	.117	.092	-.054	.106	.057	.010	-.033		.025
.075	.126	.116	.088	.062	-.077	.069	.039	.008	-.029	-.060	.075
.125	.111	.086	.062	.032	-.100	.052	.023	.007	-.028	-.066	.125
.175	.092	.062	.038	.013	-.105	.042	.008	-.015	-.039	-.064	.175
.225	.067	.041	.023	-.013	-.123	.027	-.005	-.026	-.043	-.064	.225
.275	.058	.043	.007	-.036	-.117	.017	-.012	-.021	-.047	-.069	.275
.325	.029	.014	-.011	-.046	-.125	.004	-.010	-.033	-.048	-.058	.325
.375	.030	.005	-.023	-.058	-.128	-.002	-.024	-.030	-.047	-.063	.375
.425	.006	-.008	-.033	-.056	-.122	-.012	-.027	-.037	-.055		.425
.475	.013	-.018	-.036	-.075	-.126	-.024	-.029	-.050	-.061	-.076	.475
.550	-.010	-.029	-.046	-.094	-.128	-.027	-.052	-.056	-.063	-.083	.550
.650	-.030		-.072	-.116		-.046	-.052	-.076	-.081	-.094	.650
.750	-.039	-.064	-.081	-.119	-.131	-.056	-.067	-.072	-.080	-.094	.750
.800	-.045									-.094	.800
.850		-.066	-.094	-.123	-.116	-.051	-.070	-.071	-.087		.850
.900			-.082	-.130			-.061	-.071			.900
.950					-.080	-.052					.950
LOWER SURFACE											
.025		.341	.351	.399	.497	-.072	.085	.106	.119		.025
.075	.267	.287	.288	.314	.367	-.039	.026	.068	.084	.083	.075
.125	.239	.248	.241	.259	.307	-.034	.012	.045	.064	.059	.125
.175	.219	.219	.208	.217	.267	-.045	-.012	.025	.043	.048	.175
.225	.208	.191	.188	.182	.235	-.035	-.011	.013	.030	.022	.225
.275	.184	.175	.164	.165	.214	-.035	-.005	.004	.000	.017	.275
.325	.164	.160	.145	.138	.178	-.035		-.004	.006	.014	.325
.375		.141	.128	.128	.161	-.040	-.012	-.009	-.001	-.007	.375
.425		.130	.116	.107	.142	-.047	-.021	-.018	-.007	-.009	.425
.475	.117	.119	.102	.097	.113	-.041	-.021	-.019	-.017	-.031	.475
.550	.100	.100	.068	.063	.092	-.046	-.030	-.038	-.044	-.041	.550
.650	.081	.075	.040	.045	.051		-.040		-.044	-.054	.650
.750	.061	.057	.045	.012	.004	-.070	-.050	-.060	-.056	-.059	.750
.800	.053									-.054	.800
.850		.053	.034	.012	.011	-.056	-.045	-.057	-.056		.850
.900			.028	.004			-.034	-.054			.900
.950					.001	-.056					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 5.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.090	-.067	-.074	-.093	-.100	-.111	-.102	.0
22.5				-.007	-.015	.016	-.022	-.046	-.075	-.048	22.5
45.0				.058	.043	.110	.041	.015	-.022	-.009	45.0
67.5				.055	.054	.163	.095		.015	.022	67.5
90.0				.039				-.002	.002	.004	90.0
112.5				-.012	.167	.210	.086	-.011	-.019	-.005	112.5
135.0				-.072	.499	.181	.075	-.026	-.023	-.018	135.0
157.5				-.102				-.023		-.013	157.5
180.0				-.119				-.011			180.0
202.5				-.094				-.004	.008	-.007	202.5
225.0				-.065	.038	.022	-.020	-.027	-.005	-.011	225.0
247.5				-.050	.041	.023	-.004	-.016	-.016	-.035	247.5
270.0				-.063				-.052	-.049		270.0
292.5				-.086	-.054	-.020	-.034	-.040	-.049	-.036	292.5
315.0				-.130	-.115	-.027	-.002	-.046	-.054	-.007	315.0
337.5				-.113	-.122	-.130	-.026	-.068		-.076	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.024	.013	.001	-.117	.076	-.060	-.120	-.144		.025
.075	.047	.043	.014	-.013	-.134	.045	-.039	-.104	-.131	-.149	.075
.125	.043	.018	-.007	-.033	-.150	.025	-.042	-.082	-.129	-.150	.125
.175	.027	.001	-.025	-.047	-.144	.023	-.047	-.092	-.131	-.149	.175
.225	.012	-.014	-.038	-.072	-.166	.006	-.050	-.093	-.130	-.147	.225
.275	.004	-.011	-.051	-.092	-.156	-.006	-.054	-.080	-.129	-.150	.275
.325	-.020	-.037	-.061	-.101	-.172	-.018	-.050	-.087	-.124	-.135	.325
.375	-.018	-.047	-.072	-.110	-.166	-.026	-.061	-.078	-.114	-.138	.375
.425	-.039	-.056	-.073	-.105	-.155	-.037	-.064	-.082	-.119		.425
.475	-.037	-.064	-.085	-.119	-.163	-.055	-.063	-.093	-.120	-.148	.475
.550	-.054	-.073	-.091	-.136	-.161	-.056	-.087	-.095	-.116	-.144	.550
.650	-.073		-.111	-.156		-.076	-.087	-.114	-.127	-.146	.650
.750	-.072	-.106	-.120	-.153	-.159	-.085	-.098	-.104	-.122	-.140	.750
.800	-.067									-.137	.800
.850		-.100	-.132	-.142	-.149	-.080	-.105	-.100	-.119		.850
.900			-.112	-.144			-.097	-.097			.900
.950					-.138	-.083					.950
LOWER SURFACE											
.025		.466	.466	.505	.573	.017	.204	.221	.222		.025
.075	.378	.396	.380	.396	.436	.031	.131	.163	.168	.176	.075
.125	.347	.349	.323	.333	.373	.022	.089	.127	.142	.146	.125
.175	.317	.311	.288	.288	.330	.013	.072	.101	.120	.130	.175
.225	.303	.281	.259	.252	.304	.015	.066	.088	.103	.104	.225
.275	.278	.257	.228	.220	.284	.015	.057	.073	.076	.097	.275
.325	.251	.239	.205	.205	.245	.015		.060	.077	.089	.325
.375		.217	.189	.186	.236	.015	.046	.049	.065	.071	.375
.425		.204	.173	.165	.208	.006	.037	.044	.058	.063	.425
.475	.195	.195	.157	.153	.178	.013	.037	.036	.045	.036	.475
.550	.178	.167	.130	.120	.157	.001	.021	.019	.019	.028	.550
.650	.153	.137	.093	.099	.104		.007		.008	.018	.650
.750	.124	.119	.097	.066	.053	-.030	.001	-.014	.002	.006	.750
.800	.111									.011	.800
.850		.110	.087	.053	.059	-.022	.001	.004	.002		.850
.900			.082	.053			.005	-.005			.900
.950					.046	-.022					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.070	-.042	-.052	-.047	-.081	-.082	-.095	.0
22.5				.023	.021	.054	.022	-.002	-.046	-.025	22.5
45.0				.079	.068	.158	.101	.054	-.011	.013	45.0
67.5				.062	.060	.218	.131		.012	.025	67.5
90.0				.023				.016	-.016	-.014	90.0
112.5				-.035	.190	.277	.140	-.047	-.047	-.026	112.5
135.0				-.106	.625	.246	.127	-.042	-.043	-.023	135.0
157.5				-.140				-.034		-.009	157.5
180.0				-.122				-.013			180.0
202.5				-.086				-.014	-.005	-.019	202.5
225.0				-.059	.068	.043	.020	-.041	-.022	-.029	225.0
247.5				-.065	.019	.061	.035	-.016	-.031	-.047	247.5
270.0				-.077				-.025	-.050		270.0
292.5				-.122	-.077	-.041	-.025	-.031	-.043	-.008	292.5
315.0				-.149	-.176	-.013	.016	-.036	-.052	.006	315.0
337.5				-.104	-.134	-.138	.000	-.060		-.100	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.064	-.072	-.079	-.170	.044	-.148	-.196	-.204		.025
.075	-.024	-.031	-.058	-.080	-.179	.025	-.123	-.179	-.192	-.207	.075
.125	-.021	-.045	-.066	-.091	-.191	.005	-.111	-.148	-.186	-.204	.125
.175	-.030	-.058	-.079	-.099	-.182	-.001	-.107	-.157	-.188	-.204	.175
.225	-.039	-.073	-.089	-.117	-.201	-.013	-.100	-.156	-.184	-.203	.225
.275	-.048	-.058	-.101	-.137	-.185	-.026	-.093	-.144	-.183	-.201	.275
.325	-.074	-.087	-.110	-.147	-.193	-.037	-.082	-.151	-.179	-.184	.325
.375	-.068	-.095	-.118	-.150	-.191	-.046	-.089	-.135	-.169	-.188	.375
.425	-.088	-.103	-.103	-.147	-.178	-.060	-.094	-.143	-.169		.425
.475	-.079	-.108	-.130	-.166	-.182	-.075	-.093	-.157	-.175	-.186	.475
.550	-.093	-.117	-.132	-.174	-.186	-.080	-.112	-.148	-.168	-.180	.550
.650	-.107		-.151	-.187		-.099	-.110	-.151	-.178	-.182	.650
.750	-.095	-.142	-.157	-.170	-.182	-.110	-.120	-.114	-.168	-.179	.750
.800	-.095									-.178	.800
.850		-.124	-.155	-.162	-.184	-.110	-.130	-.121	-.171		.850
.900			-.141	-.165			-.119	-.121			.900
.950					-.179	-.105					.950
LOWER SURFACE											
.025		.575	.580	.603	.644	.103	.283	.290	.286		.025
.075	.484	.490	.476	.479	.490	.075	.201	.222	.232	.235	.075
.125	.443	.437	.407	.413	.429	.050	.157	.185	.199	.195	.125
.175	.407	.399	.362	.356	.379	.034	.128	.163	.176	.181	.175
.225	.391	.364	.328	.319	.373	.033	.117	.147	.159	.157	.225
.275	.356	.338	.300	.300	.350	.045	.106	.128	.130	.151	.275
.325	.325	.312	.283	.271	.310	.045		.116	.125	.140	.325
.375		.288	.253	.252	.297	.049	.090	.101	.114	.118	.375
.425		.275	.238	.220	.272	.044	.080	.089	.104	.111	.425
.475	.265	.258	.220	.208	.239	.045	.078	.084	.090	.084	.475
.550	.247	.230	.182	.177	.215	.041	.057	.059	.059	.069	.550
.650	.220	.197	.154	.156	.157		.049		.053	.056	.650
.750	.185	.172	.148	.112	.104	.012	.030	.031	.044	.047	.750
.800	.170									.053	.800
.850		.164	.140	.109	.105	.008	.030	.031	.044		.850
.900			.131	.101			.032	.031			.900
.950					.093	.008					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 10.0°$ $\beta = -10°$

θ, deg	c_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.053	-.009	-.032	-.013	-.042	-.061	-.069	.0
22.5				.061	.047	.062	.075	.031	.008	-.006	22.5
45.0				.099	.081	.179	.148	.087	.027	.035	45.0
67.5				.069	.052	.242	.170		.006	.022	67.5
90.0				.009				.038	-.072	-.034	90.0
112.5				-.063	.190	.344	.189	-.114	-.065	-.050	112.5
135.0				-.144	.650	.310	.174	-.076	-.047	-.040	135.0
157.5				-.165				-.046		-.025	157.5
180.0				-.115				-.049			180.0
202.5				-.088				-.062	-.050	-.038	202.5
225.0				-.061	.042	.100	.038	-.046	-.038	-.029	225.0
247.5				-.086	-.035	.096	.050	-.011	-.032	-.040	247.5
270.0				-.101				-.019	-.036		270.0
292.5				-.168	-.136	-.028	-.050	-.006	-.048	.020	292.5
315.0				-.158	-.192	-.012	.023	-.001	-.057	-.004	315.0
337.5				-.088	-.124	-.118	.023	-.057		-.116	337.5

x/c	c_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.131	-.132	-.119	-.196	.025	-.197	-.246	-.243		.025
.075	-.079	-.086	-.111	-.117	-.203	.004	-.178	-.228	-.235	-.245	.075
.125	-.073	-.095	-.113	-.122	-.211	-.013	-.161	-.193	-.232	-.245	.125
.175	-.080	-.104	-.123	-.123	-.196	-.019	-.167	-.206	-.227	-.237	.175
.225	-.087	-.116	-.131	-.145	-.209	-.033	-.165	-.213	-.224	-.238	.225
.275	-.089	-.098	-.139	-.163	-.190	-.045	-.162	-.191	-.221	-.236	.275
.325	-.111	-.124	-.145	-.172	-.198	-.055	-.122	-.213	-.219	-.212	.325
.375	-.105	-.130	-.150	-.173	-.193	-.064	-.104	-.187	-.206	-.209	.375
.425	-.125	-.136	-.129	-.167	-.184	-.079	-.110	-.199	-.209		.425
.475	-.113	-.141	-.161	-.182	-.188	-.089	-.112	-.203	-.209	-.212	.475
.550	-.124	-.143	-.160	-.188	-.193	-.101	-.135	-.186	-.203	-.206	.550
.650	-.134		-.167	-.187		-.122	-.132	-.206	-.212	-.215	.650
.750	-.124	-.167	-.160	-.176	-.193	-.131	-.143	-.178	-.201	-.214	.750
.800	-.124									-.212	.800
.850		-.150	-.157	-.169	-.193	-.126	-.151	-.154	-.208		.850
.900			-.149	-.175			-.142	-.149			.900
.950					-.191	-.124					.950
LOWER SURFACE											
.025		.675	.681	.690	.710	.182	.366	.353	.349		.025
.075	.580	.584	.568	.558	.538	.076	.279	.296	.298	.296	.075
.125	.539	.527	.491	.485	.457	.031	.232	.253	.271	.263	.125
.175	.501	.484	.446	.432	.426	.028	.198	.228	.240	.246	.175
.225	.482	.445	.413	.390	.434	.041	.185	.205	.219	.222	.225
.275	.446	.415	.384	.369	.409	.063	.170	.187	.192	.211	.275
.325	.413	.390	.351	.340	.381	.078		.176	.189	.198	.325
.375		.363	.331	.316	.370	.092	.147	.155	.176	.182	.375
.425		.348	.312	.296	.338	.086	.135	.149	.164	.167	.425
.475	.352	.330	.291	.284	.313	.080	.134	.141	.148	.148	.475
.550	.327	.301	.258	.243	.287	.084	.114	.122	.118	.127	.550
.650	.294	.259	.220	.222	.226		.083		.111	.114	.650
.750	.254	.238	.213	.175	.163	.035	.066	.078	.103	.110	.750
.800	.237									.117	.800
.850		.225	.202	.167	.163	.025	.066	.082	.093		.850
.900			.193	.162			.072	.078			.900
.950					.156	.035					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 12.5°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.027	.034	.020	.045	-.007	-.040	-.036	.0
22.5				.099	.083	.108	.137	.077	.033	.023	22.5
45.0				.127	.110	.237	.197	.148	.063	.045	45.0
67.5				.075	.066	.277	.204		.032	-.002	67.5
90.0				-.002				.082	-.065	-.076	90.0
112.5				-.089	.190	.429	.252	-.123	-.148	-.062	112.5
135.0				-.171	.653	.385	.246	-.142	-.076	-.034	135.0
157.5				-.183				-.056		-.050	157.5
180.0				-.123				-.079			180.0
202.5				-.089				-.042	-.075	-.074	202.5
225.0				-.067	-.038	.113	.079	-.054	-.031	-.054	225.0
247.5				-.103	-.094	.113	.092	-.013	-.033	-.050	247.5
270.0				-.135				-.013	-.042		270.0
292.5				-.187	-.157	.002	.007	.005	-.054	.005	292.5
315.0				-.151	-.174	-.025	.016	.002	-.087	-.029	315.0
337.5				-.065	-.090	-.088	-.033	-.055		-.110	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.180	-.178	-.148	-.229	-.017	-.236	-.277	-.271		.025
.075	-.130	-.132	-.159	-.157	-.224	-.021	-.229	-.265	-.263	-.257	.075
.125	-.118	-.142	-.155	-.160	-.226	-.041	-.203	-.240	-.257	-.259	.125
.175	-.119	-.144	-.159	-.156	-.212	-.042	-.222	-.250	-.257	-.256	.175
.225	-.122	-.150	-.162	-.176	-.221	-.057	-.231	-.250	-.251	-.254	.225
.275	-.125	-.135	-.169	-.196	-.205	-.067	-.217	-.230	-.250	-.251	.275
.325	-.148	-.156	-.175	-.196	-.212	-.075	-.175	-.247	-.247	-.232	.325
.375	-.136	-.163	-.180	-.193	-.212	-.083	-.159	-.229	-.234	-.232	.375
.425	-.157	-.167	-.151	-.184	-.200	-.103	-.134	-.240	-.236		.425
.475	-.141	-.172	-.186	-.199	-.209	-.116	-.128	-.241	-.237	-.234	.475
.550	-.150	-.167	-.184	-.204	-.211	-.125	-.150	-.218	-.228	-.229	.550
.650	-.154		-.181	-.196		-.150	-.153	-.241	-.240	-.243	.650
.750	-.143	-.169	-.173	-.188	-.212	-.154	-.159	-.220	-.229	-.241	.750
.800	-.145									-.241	.800
.850		-.165	-.174	-.186	-.219	-.147	-.167	-.202	-.233		.850
.900			-.167	-.191			-.157	-.197			.900
.950					-.212	-.141					.950
LOWER SURFACE											
.025		.755	.756	.762	.782	.203	.445	.416	.394		.025
.075	.663	.668	.640	.626	.607	.040	.365	.362	.358	.356	.075
.125	.620	.607	.566	.551	.494	-.047	.319	.323	.325	.322	.125
.175	.580	.561	.517	.491	.465	-.041	.279	.297	.297	.302	.175
.225	.560	.522	.483	.453	.501	-.024	.264	.276	.281	.281	.225
.275	.523	.488	.451	.422	.465	.057	.241	.259	.260	.268	.275
.325	.498	.460	.413	.394	.434	.080		.239	.250	.255	.325
.375		.429	.401	.380	.422	.114	.213	.227	.237	.235	.375
.425		.414	.373	.362	.399	.108	.201	.211	.225	.224	.425
.475	.425	.398	.351	.343	.370	.096	.192	.201	.207	.207	.475
.550	.397	.368	.320	.296	.348	.106	.167	.177	.175	.182	.550
.650	.360	.317	.272	.278	.280		.137		.158	.170	.650
.750	.319	.299	.271	.226	.216	.059	.123	.134	.151	.166	.750
.800	.299									.170	.800
.850		.287	.259	.215	.224	.070	.123	.137	.150		.850
.900			.246	.208			.123	.137			.900
.950					.212	.070					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 15.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.012	.062	.062	.093	.040	-.025	.000	.0
22.5				.146	.123	.158	.184	.131	.069	.056	22.5
45.0				.152	.129	.251	.247	.187	.101	.070	45.0
67.5				.076	.063	.327	.272		.053	.016	67.5
90.0				-.022				.102	-.059	-.109	90.0
112.5				-.102	.197	.499	.317	-.111	-.189	-.062	112.5
135.0				-.211	.658	.451	.301	-.194	-.112	-.054	135.0
157.5				-.191				-.115		-.100	157.5
180.0				-.160				-.041			180.0
202.5				-.105				-.126	-.090	-.082	202.5
225.0				-.087	-.159	.126	.116	-.039	-.034	-.057	225.0
247.5				-.153	-.176	.087	.148	-.028	-.060	-.002	247.5
270.0				-.170				-.027	-.078		270.0
292.5				-.205	-.186	-.022	.036	-.021	-.083	.018	292.5
315.0				-.145	-.164	-.153	.020	.001	-.121	.002	315.0
337.5				-.042	-.061	-.076	-.053	-.097		-.090	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.236	-.234	-.180	-.264	-.014	-.267	-.301	-.281		.025
.075	-.189	-.188	-.211	-.209	-.253	-.043	-.280	-.293	-.276	-.266	.075
.125	-.179	-.198	-.201	-.208	-.252	-.073	-.252	-.278	-.273	-.267	.125
.175	-.179	-.196	-.200	-.200	-.240	-.081	-.268	-.292	-.271	-.266	.175
.225	-.179	-.198	-.205	-.214	-.253	-.088	-.275	-.288	-.269	-.265	.225
.275	-.176	-.174	-.208	-.221	-.233	-.098	-.263	-.276	-.269	-.264	.275
.325	-.196	-.198	-.213	-.220	-.245	-.111	-.243	-.280	-.266	-.247	.325
.375	-.175	-.205	-.220	-.218	-.239	-.115	-.233	-.277	-.256	-.247	.375
.425	-.184	-.204	-.183	-.211	-.226	-.134	-.201	-.280	-.257		.425
.475	-.170	-.204	-.221	-.224	-.236	-.144	-.207	-.286	-.256	-.255	.475
.550	-.176	-.188	-.214	-.226	-.238	-.151	-.219	-.256	-.254	-.250	.550
.650	-.173		-.202	-.224		-.174	-.187	-.276	-.264	-.263	.650
.750	-.168	-.192	-.202	-.218	-.238	-.184	-.183	-.238	-.256	-.261	.750
.800	-.176									-.262	.800
.850		-.188	-.207	-.214	-.239	-.182	-.189	-.226	-.262		.850
.900			-.198	-.224			-.187	-.219			.900
.950					-.240	-.175					.950
LOWER SURFACE											
.025		.825	.819	.833	.817	-.135	.487	.460	.430		.025
.075	.743	.741	.711	.700	.649	-.171	.411	.418	.403	.407	.075
.125	.700	.684	.638	.629	.530	-.187	.381	.382	.374	.371	.125
.175	.663	.635	.579	.559	.501	-.148	.349	.355	.352	.348	.175
.225	.642	.593	.541	.518	.527	-.098	.340	.331	.329	.320	.225
.275	.603	.560	.517	.485	.508	.002	.325	.310	.304	.311	.275
.325	.575	.530	.486	.461	.492	.031		.293	.296	.299	.325
.375		.502	.460	.452	.492	.057	.283	.277	.283	.278	.375
.425		.484	.428	.421	.452	.058	.257	.260	.272	.267	.425
.475	.498	.464	.405	.401	.430	.073	.246	.249	.253	.252	.475
.550	.471	.431	.376	.349	.397	.082	.218	.222	.216	.234	.550
.650	.429	.378	.326	.329	.333		.186		.206	.226	.650
.750	.389	.360	.329	.274	.265	.067	.167	.171	.193	.221	.750
.800	.365									.221	.800
.850		.351	.312	.265	.277	.093	.166	.180	.193		.850
.900			.303	.258			.168	.170			.900
.950					.265	.106					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.121	-.191	-.177	-.156	-.198	-.135	.035	.0
22.5				-.089	-.121	-.125	-.168	-.183	-.170	-.011	22.5
45.0				.020	-.022	-.014	-.065	-.093	-.093	-.085	45.0
67.5				.068	.069	.077	.020		.005	.000	67.5
90.0				.104				.005	.026	.033	90.0
112.5				.084	.236	.159	.051	.005	.007	.030	112.5
135.0				.011	.518	.126	.036	-.014	-.021	-.013	135.0
157.5				-.070				-.011	-.037	-.061	157.5
180.0				-.159				-.043			180.0
202.5				-.208				-.014	-.040	-.063	202.5
225.0				-.118	-.082	-.065	-.061	-.016	-.030	-.040	225.0
247.5				-.098	-.077	-.085	-.082	-.036	-.051	-.065	247.5
270.0				-.068				-.065	-.102		270.0
292.5				-.091	-.119	-.160	-.196	-.155	-.139	-.022	292.5
315.0				-.110	-.130	-.140	-.159	-.156	-.112	.028	315.0
337.5				-.205	-.145	-.082	-.151	-.161		.033	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.280	.265	.218	-.015	.096	.134	.088	.078		.025
.075	.166	.231	.204	.154	-.043	.075	.091	.058	.054	.034	.075
.125	.154	.192	.160	.109	-.064	.056	.064	.038	.041	.023	.125
.175	.140	.159		.074	-.079	.044	.042	.024	.029	.013	.175
.225	.115	.134	.107	.051	-.088	.024	.028	.011	.020	.010	.225
.275	.109	.116	.090	.032	-.088	.014	.019	.002	.011	.001	.275
.325	.098	.097	.068	.017	-.100	.008	.012	-.004	.004	.000	.325
.375	.087	.087	.055	.005	-.106	.004	.005	-.008	.001	-.008	.375
.425	.074	.076	.047	-.011	-.109	-.005	-.006	-.017	-.007		.425
.475	.061	.064	.034	-.024	-.116	-.014	-.008	-.026	-.013	-.026	.475
.550	.044	.038	.008	-.045	-.126	-.020	-.028	-.038	-.025	-.033	.550
.650	.017		-.019	-.070		-.037	-.027	-.053	-.038	-.043	.650
.750	.008	-.002	-.037	-.088	-.146	-.036	-.049	-.032	-.043	-.049	.750
.800	.006									-.050	.800
.850		-.004	-.043	-.090	-.142	-.032	-.045	-.031	-.047		.850
.900			-.043	-.091			-.044	-.032			.900
.950					-.136	-.036					.950
LOWER SURFACE											
.025		.315	.306	.399	.566	-.167	-.116	-.115	-.106		.025
.075	.147	.259	.253	.320	.438	-.158	-.123	-.089	-.082	-.080	.075
.125	.133	.221	.219	.273	.371	-.164	-.112	-.087	-.073	-.076	.125
.175	.128	.188	.189	.219	.320	-.151	-.115	-.084	-.073	-.077	.175
.225	.128	.164	.170	.191	.288	-.135	-.097	-.082	-.079	-.092	.225
.275	.120	.148	.154	.170	.260	-.108	-.084	-.082	-.093	-.090	.275
.325	.111	.132	.135	.148	.210	-.100	-.082	-.085	-.085	-.085	.325
.375	.102	.122	.122	.137	.196	-.084	-.078	-.083	-.085	-.097	.375
.425		.108	.104	.113	.170	-.074	-.074	-.082	-.091	-.102	.425
.475	.081	.095	.093	.099	.144	-.060	-.072	-.080	-.093	-.112	.475
.550	.062	.083	.072	.089	.122	-.054	-.080	-.086	-.106	-.117	.550
.650	.051	.055	.039	.054	.074		-.086	-.093	-.103	-.126	.650
.750	.041	.043	.040	.021		-.079	-.084	-.096	-.103	-.119	.750
.800	.036									-.102	.800
.850		.036	.030	.013	.038	-.063	-.082	-.087	-.098		.850
.900			.029	.012			-.073	-.086			.900
.950					.026	-.064					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 2.5°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.182	-.157	-.148	-.182	-.199	-.135	-.035	.0
22.5				-.071	-.084	-.064	-.115	-.134	-.149	-.128	22.5
45.0				.034	.008	.054	-.006	-.034	-.072	-.044	45.0
67.5				.092	.087	.133	.084		.007	.013	67.5
90.0				.105				-.007	.022	.029	90.0
112.5				.063	.273	.221	.097	-.007	-.015	.006	112.5
135.0				-.027	.605	.192	.092	-.045	-.055	-.058	135.0
157.5				-.094				-.057	-.034	-.077	157.5
180.0				-.174				-.043			180.0
202.5				-.189				-.024	-.054	-.054	202.5
225.0				-.142	-.013	-.037	-.068	-.029	-.044	-.075	225.0
247.5				-.097	.017	-.062	-.073	-.044	-.057	-.065	247.5
270.0				-.080				-.078	-.091		270.0
292.5				-.096	-.128	-.200	-.100	-.115	-.127	-.084	292.5
315.0				-.152	-.140	-.112	-.113	-.138	-.110	-.020	315.0
337.5				-.210	-.160	-.079	-.111	-.147		-.041	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.167	.161	.137	-.068	.103	.077	.008	-.037		.025
.075	.110	.165	.133	.094	-.081	.083	.047	.000	-.028	-.073	.075
.125	.104	.129	.101	.060	-.099	.054	.027	-.001	-.026	-.072	.125
.175	.091	.103	.086	.044	-.098	.045	.012	-.017	-.034	-.068	.175
.225	.079	.078	.052	.012	-.120	.019	-.002	-.028	-.037	-.065	.225
.275	.070	.084	.037	-.014	-.115	.007	-.012	-.025	-.041	-.067	.275
.325	.049	.049	.023	-.027	-.128	.000	-.007	-.039	-.042	-.063	.325
.375	.051	.039	.009	-.040	-.134	-.005	-.025	-.032	-.043	-.065	.375
.425	.026	.027	.015	-.036	-.121	-.011	-.025	-.043	-.049		.425
.475	.034	.018	-.008	-.057	-.143	-.021	-.031	-.056	-.054	-.076	.475
.550	.018	.004	-.018	-.078	-.144	-.030	-.052	-.059	-.057	-.079	.550
.650	-.006		-.041	-.099		-.045	-.047	-.082	-.075	-.087	.650
.750	-.011	-.040	-.053	-.098	-.167	-.056	-.059	-.065	-.071	-.092	.750
.800	-.011									-.087	.800
.850		-.038	-.069	-.103	-.163	-.057	-.068	-.064	-.076		.850
.900			-.054	-.114			-.059	-.062			.900
.950					-.154	-.063					.950
LOWER SURFACE											
.025		.426	.446	.524	.657	-.109	-.007	.048	.073		.025
.075	.278	.384	.376	.419	.491	-.096	-.025	.026	.035	.045	.075
.125	.258	.321	.321	.359	.420	-.104	-.028	.004	.025	.025	.125
.175	.245	.284	.281	.293	.376	-.104	-.041	.001	.017	.013	.175
.225	.237	.258	.255	.265	.357	-.086	-.027	-.005	.011	-.005	.225
.275	.218	.237	.233	.236	.325	-.072	-.017	-.009	-.015	-.005	.275
.325	.199	.224	.208	.209	.279	-.076	-.019	-.017	-.008	-.001	.325
.375	.182	.204	.188	.204	.263	-.060	-.021	-.019	-.014	-.025	.375
.425		.189	.174	.175	.229	-.060	-.034	-.033	-.018	-.021	.425
.475	.158	.178	.163	.158	.201	-.054	-.031	-.031	-.027	-.053	.475
.550	.136	.156	.131	.125	.181	-.052	-.040	-.048	-.056	-.057	.550
.650	.118	.127	.093	.106	.125		-.050	-.059	-.052	-.066	.650
.750	.103	.108	.097	.067		-.073	-.057	-.060	-.063	-.066	.750
.800	.097									-.058	.800
.850		.099	.083	.066	.078	-.059	-.052	-.048	-.058		.850
.900			.078	.060			-.039	-.052			.900
.950					.065	-.065					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

HIGH – WING CONFIGURATION

$\alpha = 5.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.135	-.113	-.101	-.147	-.179	-.178	-.077	.0
22.5				-.003	-.024	.010	-.063	-.086	-.112	-.094	22.5
45.0				.089	.058	.127	.049	.024	-.022	-.020	45.0
67.5				.120	.109	.208	.123		.031	.035	67.5
90.0				.100				.014	.010	.019	90.0
112.5				.027	.276	.286	.147	-.040	-.041	-.027	112.5
135.0				-.070	.700	.257	.130	-.058	-.085	-.078	135.0
157.5				-.143				-.079	-.071	-.095	157.5
180.0				-.207				-.064			180.0
202.5				-.160				-.036	-.057	-.080	202.5
225.0				-.111	.068	-.028	-.076	-.036	-.034	-.078	225.0
247.5				-.070	.107	-.016	-.063	-.047	-.051	-.073	247.5
270.0				-.094				-.085	-.094		270.0
292.5				-.116	-.156	-.162	-.075	-.080	-.086	-.101	292.5
315.0				-.208	-.134	-.070	-.098	-.097	-.111	-.085	315.0
337.5				-.188	-.211	-.090	-.075	-.115		-.066	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.063	.052	.034	-.147	.099	-.015	-.136	-.168		.025
.075	.041	.072	.050	-.004	-.156	.089	-.008	-.116	-.150	-.194	.075
.125	.033	.056	.031	-.027	-.168	.054	-.019	-.097	-.143	-.188	.125
.175	.025	.036	.013	-.049	-.175	.044	-.030	-.077	-.142	-.182	.175
.225	.015	.018	-.012	-.046	-.187	.027	-.040	-.072	-.138	-.176	.225
.275	.009	.006	-.027	-.084	-.175	.009	-.046	-.075	-.135	-.175	.275
.325	-.006	-.007	-.039	-.091	-.187	-.002	-.056	-.085	-.124	-.166	.325
.375	-.017	-.018	-.050	-.102	-.191	-.013	-.060	-.086	-.111	-.167	.375
.425	-.025	-.030	-.054	-.109	-.188	-.021	-.064	-.086	-.084		.425
.475	-.033	-.036	-.066	-.117	-.195	-.030	-.067	-.091	-.082	-.170	.475
.550	-.047	-.054	-.082	-.134	-.202	-.039	-.086	-.098	-.092	-.167	.550
.650	-.058		-.108	-.155		-.056	-.085	-.114	-.107	-.172	.650
.750	-.053	-.089	-.117	-.161	-.200	-.064	-.098	-.091	-.106	-.170	.750
.800	-.046									-.165	.800
.850		-.075	-.122	-.162	-.193	-.077	-.098	-.092	-.112		.850
.900			-.103	-.162			-.096	-.089			.900
.950					-.179	-.086					.950
LOWER SURFACE											
.025		.560	.578	.630	.728	-.012	.138	.174	.178		.025
.075	.419	.477	.482	.501	.560	-.019	.085	.126	.137	.141	.075
.125	.385	.425	.414	.431	.487	-.035	.073	.098	.110	.113	.125
.175	.358	.381	.367	.371	.443	-.048	.051	.084	.089	.098	.175
.225	.343	.352	.336	.337	.430	-.053	.050	.070	.077	.080	.225
.275	.317	.324	.308	.310	.392	-.034	.043	.059	.059	.072	.275
.325	.295	.307	.278	.286	.338	-.044	.030	.044	.057	.061	.325
.375	.276	.283	.261	.268	.328	-.037	.025	.037	.047	.048	.375
.425		.268	.246	.237	.292	-.043	.018		.041	.043	.425
.475	.241	.252	.226	.219	.265	-.044	.017	.024	.031	.025	.475
.550	.216	.229	.194	.180	.237	-.050	.006	.006	.006	.011	.550
.650	.185	.189	.154	.158	.172		-.011	-.005	.005	.008	.650
.750	.163	.168	.149	.122		-.083	-.009	-.013	-.004	-.002	.750
.800	.151									-.001	.800
.850		.158	.139	.115	.123	-.072	-.005	.004	-.002		.850
.900			.132	.109			-.001	.006			.900
.950					.103	-.071					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.117	-.075	-.080	-.104	-.142	-.178	-.128	.0
22.5				.028	.010	.057	-.019	-.033	-.091	-.068	22.5
45.0				.122	.091	.183	.103	.073	-.014	.009	45.0
67.5				.133	.118	.262	.190		.013	.033	67.5
90.0				.093				.044	.002	.006	90.0
112.5				.013	.249	.346	.191	-.106	-.063	-.050	112.5
135.0				-.090	.757	.310	.182	-.096	-.108	-.101	135.0
157.5				-.158				-.097	-.079	-.082	157.5
180.0				-.220				-.082			180.0
202.5				-.142				-.030	-.061	-.087	202.5
225.0				-.098	.132	-.057	-.066	-.040	-.042	-.086	225.0
247.5				-.080	.104	-.031	-.049	-.056	-.051	-.076	247.5
270.0				-.101				-.058	-.090		270.0
292.5				-.139	-.149	-.167	-.062	-.083	-.083	-.086	292.5
315.0				-.224	-.162	-.050	-.083	-.096	-.118	-.048	315.0
337.5				-.177	-.207	-.119	-.054	-.130		-.042	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.017	-.025	-.025	-.176	.098	-.115	-.188	-.224		.025
.075	-.005	.019	-.012	-.052	-.185	.063	-.088	-.175	-.210	-.236	.075
.125	-.001	.001	-.024	-.069	-.193	.054	-.058	-.154	-.209	-.235	.125
.175	-.009	-.014	-.033	-.081	-.193	.036	-.060	-.178	-.203	-.228	.175
.225	-.019	-.028	-.056	-.101	-.207	.005	-.069	-.161	-.196	-.227	.225
.275	-.025	-.030	-.065	-.121	-.195	-.013	-.073	-.134	-.197	-.225	.275
.325	-.044	-.050	-.077	-.129	-.213	-.031	-.073	-.101	-.189	-.206	.325
.375	-.044	-.058	-.086	-.136	-.214	-.036	-.085	-.099	-.173	-.207	.375
.425	-.060	-.065	-.084	-.134	-.204	-.046	-.085	-.105	-.170		.425
.475	-.059	-.073	-.099	-.149	-.207	-.057	-.090	-.116	-.171	-.212	.475
.550	-.071	-.088	-.109	-.160	-.206	-.069	-.104	-.116	-.164	-.209	.550
.650	-.081		-.133	-.179		-.084	-.107	-.133	-.151	-.217	.650
.750	-.070	-.118	-.146	-.180	-.201	-.092	-.114	-.123	-.137	-.216	.750
.800	-.064									-.212	.800
.850		-.090	-.147	-.169	-.192	-.107	-.120	-.120	-.138		.850
.900			-.123	-.168			-.114	-.119			.900
.950					-.184	-.105					.950
LOWER SURFACE											
.025		.685	.694	.727	.793	.060	.240	.249	.241		.025
.075	.553	.589	.578	.587	.611	.051	.190	.194	.198	.201	.075
.125	.509	.526	.502	.508	.536	.021	.151	.162	.169	.168	.125
.175	.473	.480	.450	.447	.505	-.004	.116	.143	.151	.154	.175
.225	.456	.444	.419	.410	.501	-.012	.102	.125	.135	.135	.225
.275	.424	.414	.390	.378	.462	-.006	.086	.113	.115	.129	.275
.325	.399	.389	.356	.346	.408	-.021	.073	.096	.108	.119	.325
.375	.374	.367	.334	.328	.398	-.028	.067	.091	.102	.102	.375
.425		.345	.308	.304	.354	-.038	.059		.091	.095	.425
.475	.334	.331	.293	.285	.326	-.048	.058	.073	.082	.076	.475
.550	.301	.301	.256	.240	.300	-.058	.043	.052	.058	.063	.550
.650	.259	.254	.209	.219	.232		.032	.046	.052	.052	.650
.750	.225	.236	.209	.172		-.076	.020	.032	.045	.047	.750
.800	.217									.052	.800
.850		.224	.196	.174	.171	-.064	.019	.046	.044		.850
.900			.191	.160			.022	.047			.900
.950					.157	-.061					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING – BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 10.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.082	-.043	-.017	-.045	-.097	-.143	-.133	.0
22.5				.073	.059	.116	.052	.019	-.026	-.028	22.5
45.0				.161	.137	.242	.156	.108	.044	.029	45.0
67.5				.148	.136	.325	.229		.034	.050	67.5
90.0				.089				.075	-.027	-.009	90.0
112.5				-.019	.324	.429	.253	-.118	-.095	-.083	112.5
135.0				-.128	.808	.395	.242	-.123	-.112	-.122	135.0
157.5				-.196				-.072	-.072	-.086	157.5
180.0				-.200				-.085			180.0
202.5				-.162				-.079	-.082	-.066	202.5
225.0				-.098	.135	-.086	-.031	-.142	-.148	-.073	225.0
247.5				-.101	.001	-.059	-.019	-.020	-.045	-.085	247.5
270.0				-.146				-.051	-.112		270.0
292.5				-.172	-.139	-.111	-.052	-.072	-.102	-.095	292.5
315.0				-.226	-.201	-.027	-.061	-.097	-.107	-.079	315.0
337.5				-.139	-.179	-.149	-.082	-.150		-.066	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.110	-.112	-.089	-.218	.125	-.191	-.240	-.264		.025
.075	-.073	-.064	-.084	-.118	-.221	.036	-.188	-.238	-.261	-.272	.075
.125	-.063	-.070	-.086	-.125	-.232	-.007	-.170	-.219	-.261	-.274	.125
.175	-.066	-.077	-.073	-.127	-.227	-.006	-.102	-.239	-.255	-.269	.175
.225	-.072	-.086	-.105	-.147	-.234	-.027	-.110	-.239	-.249	-.266	.225
.275	-.075	-.077	-.114	-.163	-.211	-.044	-.108	-.215	-.249	-.266	.275
.325	-.093	-.097	-.123	-.169	-.222	-.058	-.104	-.221	-.242	-.243	.325
.375	-.086	-.104	-.133	-.175	-.215	-.071	-.111	-.182	-.229	-.243	.375
.425	-.104	-.112	-.129	-.173	-.209	-.090	-.115	-.169	-.231		.425
.475	-.098	-.118	-.144	-.185	-.211	-.101	-.117	-.165	-.233	-.245	.475
.550	-.108	-.127	-.148	-.198	-.208	-.115	-.136	-.153	-.223	-.242	.550
.650	-.109		-.170	-.207		-.131	-.134	-.166	-.231	-.251	.650
.750	-.095	-.153	-.170	-.189	-.212	-.141	-.141	-.147	-.211	-.253	.750
.800	-.092									-.252	.800
.850		-.124	-.167	-.180	-.211	-.150	-.150	-.147	-.217		.850
.900			-.151	-.175			-.140	-.147			.900
.950					-.211	-.146					.950
LOWER SURFACE											
.025		.790	.792	.816	.868	.094	.352	.305	.288		.025
.075	.661	.685	.669	.668	.661	.077	.269	.257	.255	.249	.075
.125	.615	.618	.592	.590	.574	.017	.218	.228	.229	.223	.125
.175	.575	.568	.537	.524	.541	-.013	.174	.203	.207	.209	.175
.225	.554	.530	.492	.484	.568	-.035	.165	.179	.189	.183	.225
.275	.516	.498	.462	.442	.515	-.048	.146	.170	.171	.177	.275
.325	.491	.470	.432	.413	.479	-.071	.136	.157	.165	.166	.325
.375	.467	.442	.406	.396	.462	-.076	.126	.150	.155	.153	.375
.425		.416	.373	.374	.423	-.076	.113		.148	.138	.425
.475	.421	.401	.355	.355	.393	-.083	.112	.130	.136	.128	.475
.550	.382	.371	.327	.301	.366	-.076	.090		.105	.112	.550
.650	.328	.319	.275	.287	.293		.066	.090	.097	.099	.650
.750	.289	.303	.268	.236		-.055	.052	.081	.087	.098	.750
.800	.273									.107	.800
.850		.286	.255	.224	.229	-.033	.064	.099	.092		.850
.900			.250	.214			.068	.099			.900
.950					.216	-.031					.950

TABLE 3, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

HIGH–WING CONFIGURATION

$\alpha = 12.5°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.035	.001	.013	-.006	-.048	-.098	-.091	.0
22.5				.124	.089	.168	.129	.062	.028	.017	22.5
45.0				.185	.167	.295	.231	.161	.087	.073	45.0
67.5				.166	.149	.374	.274		.070	.051	67.5
90.0				.073				.110	-.045	-.044	90.0
112.5				-.044	.308	.506	.314	-.113	-.138	-.129	112.5
135.0				-.166	.797	.463	.296	-.159	-.103	-.098	135.0
157.5				-.228				-.085	-.089	-.062	157.5
180.0				-.194				-.062			180.0
202.5				-.170				-.091	-.076	-.050	202.5
225.0				-.093	.012	-.016	-.036	-.131	-.040	-.069	225.0
247.5				-.134	-.104	-.017	.047	-.007	-.112	-.155	247.5
270.0				-.191				-.121	-.159		270.0
292.5				-.198	-.147	-.078	-.050	-.105	-.134	-.035	292.5
315.0				-.209	-.224	-.049	-.079	-.089	-.106	-.026	315.0
337.5				-.120	-.145	-.148	-.156	-.141		-.045	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.185	-.181	-.133	-.261	.128	-.246	-.281	-.281		.025
.075	-.143	-.130	-.148	-.173	-.254	.056	-.250	-.282	-.283	-.280	.075
.125	-.128	-.129	-.143	-.171	-.259	-.008	-.246	-.268	-.284	-.281	.125
.175	-.127	-.129	-.139	-.173	-.244	-.050	-.231	-.283	-.282	-.280	.175
.225	-.127	-.135	-.155	-.185	-.244	-.078	-.219	-.284	-.279	-.279	.225
.275	-.127	-.127	-.160	-.199	-.223	-.091	-.124	-.264	-.279	-.279	.275
.325	-.146	-.146	-.167	-.207	-.230	-.107	-.136	-.278	-.276	-.259	.325
.375	-.134	-.152	-.175	-.209	-.227	-.118	-.147	-.249	-.262	-.262	.375
.425	-.154	-.156	-.168	-.205	-.225	-.139	-.149	-.240	-.263		.425
.475	-.148	-.159	-.179	-.218	-.223	-.148	-.152	-.231	-.267	-.263	.475
.550	-.150	-.165	-.184	-.220	-.223	-.159	-.167	-.209	-.258	-.262	.550
.650	-.141	-.173	-.187	-.213		-.178	-.163	-.223	-.265	-.268	.650
.750	-.129	-.172	-.175	-.201	-.230	-.185	-.172	-.195	-.254	-.270	.750
.800	-.130									-.270	.800
.850		-.154	-.173	-.194	-.224	-.187	-.177	-.186	-.243		.850
.900			-.168	-.195			-.167	-.184			.900
.950					-.226	-.186					.950
LOWER SURFACE											
.025		.877	.878	.887	.959	.002	.421	.353	.322		.025
.075	.754	.771	.748	.742	.737	-.026	.341	.306	.295	.293	.075
.125	.706	.702	.670	.657	.608	-.077	.284	.276	.273	.268	.125
.175	.662	.648	.610	.592	.601	-.119	.251	.257	.255	.252	.175
.225	.643	.605	.566	.544	.614	-.157	.228	.237	.238	.238	.225
.275	.599	.573	.534	.507	.573	-.145	.204	.223	.224	.223	.275
.325	.573	.545	.497	.486	.549	-.143	.196	.210	.214	.215	.325
.375	.546	.515	.472	.470	.529	-.129	.186	.199	.204	.202	.375
.425		.490	.441	.434	.483	-.119	.182		.198	.189	.425
.475	.494	.471	.422	.412	.458	-.094	.178	.182	.179	.176	.475
.550	.452	.436	.389	.366	.418	-.057	.162		.152	.162	.550
.650	.392	.380	.328	.345	.348		.140	.144	.147	.162	.650
.750	.353	.364	.329	.286		-.085	.136	.147	.150	.166	.750
.800	.335									.168	.800
.850		.345	.318	.277	.290	-.067	.144	.162	.155		.850
.900			.305	.270			.146	.163			.900
.950					.271	-.053					.950

TABLE 3, Concluded

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

HIGH-WING CONFIGURATION

$\alpha = 15.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.006	.054	.047	.042	-.003	-.055	-.048	.0
22.5				.178	.147	.224	.176	.114	.064	.072	22.5
45.0				.226	.200	.346	.283	.218	.135	.098	45.0
67.5				.171	.160	.410	.311		.098	.056	67.5
90.0				.066				.143	-.023	-.031	90.0
112.5				-.070	.317	.589	.380	-.094	-.177	-.151	112.5
135.0				-.181	.764	.534	.361	-.151	-.136	-.098	135.0
157.5				-.219				-.066	-.118	-.077	157.5
180.0				-.188				-.101			180.0
202.5				-.171				-.133	-.047	-.049	202.5
225.0				-.107	-.108	-.002	.034	-.086	-.092	-.114	225.0
247.5				-.200	-.179	-.056	.015	-.117	-.176	-.100	247.5
270.0				-.191				-.146	-.155		270.0
292.5				-.203	-.161	-.114	-.094	-.107	-.150	-.093	292.5
315.0				-.196	-.218	-.196	-.077	-.080	-.163	-.075	315.0
337.5				-.070	-.114	-.119	-.122	-.160		-.137	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.232	-.225	-.159	-.279	.194	-.266	-.289	-.284		.025
.075	-.189	-.176	-.191	-.202	-.270	.073	-.276	-.289	-.283	-.266	.075
.125	-.176	-.178	-.184	-.205	-.264	.001	-.270	-.281	-.283	-.269	.125
.175	-.169	-.174	-.166	-.199	-.253	-.033	-.271	-.291	-.284	-.267	.175
.225	-.167	-.178	-.191	-.213	-.253	-.050	-.265	-.294	-.284	-.263	.225
.275	-.167	-.162	-.189	-.228	-.230	-.097	-.231	-.279	-.282	-.262	.275
.325	-.187	-.179	-.197	-.228	-.240	-.129	-.207	-.290	-.280	-.253	.325
.375	-.173	-.186	-.204	-.228	-.238	-.147	-.219	-.275	-.273	-.258	.375
.425	-.189	-.188	-.193	-.221	-.231	-.168	-.197	-.276	-.275		.425
.475	-.172	-.187	-.208	-.227	-.233	-.176	-.182	-.272	-.276	-.261	.475
.550	-.166	-.182	-.208	-.221	-.227	-.188	-.173	-.249	-.271	-.256	.550
.650	-.155	-.178	-.193	-.219		-.204	-.182	-.268	-.281	-.263	.650
.750	-.149	-.170	-.185	-.211	-.236	-.211	-.189	-.245	-.267	-.265	.750
.800	-.160									-.261	.800
.850		-.166	-.193	-.204	-.234	-.219	-.198	-.233	-.256		.850
.900			-.188	-.207			-.188	-.230			.900
.950					-.236	-.214					.950
LOWER SURFACE											
.025		.958	.956	.966	1.035	-.219	.467	.399	.352		.025
.075	.835	.854	.829	.819	.794	-.159	.381	.367	.345	.332	.075
.125	.788	.784	.750	.736	.656	-.223	.334	.339	.321	.316	.125
.175	.747	.731	.684	.664	.649	-.217	.299	.314	.303	.302	.175
.225	.724	.687	.641	.621	.668	-.206	.295	.293	.294	.284	.225
.275	.680	.646	.606	.588	.637	-.142	.289	.280	.280	.275	.275
.325	.655	.616	.572	.556	.637	-.164	.263	.261	.271	.264	.325
.375	.627	.590	.542	.534	.615	-.165	.250	.252	.262	.256	.375
.425		.562	.514	.505	.553	-.160	.240		.254	.244	.425
.475	.574	.541	.488	.479	.533	-.109	.236	.240	.238	.235	.475
.550	.525	.507	.450	.427	.492	-.059	.235		.209	.224	.550
.650	.462	.449	.394	.401	.412		.214	.202	.210	.228	.650
.750	.415	.430	.392	.347		-.028	.208	.204	.209	.215	.750
.800	.399									.219	.800
.850		.416	.378	.329	.350	-.004	.206	.211	.209		.850
.900			.370	.329			.201	.206			.900
.950					.332	.017					.950

TABLE 4

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha =$ 0° $\beta =$ 0°

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.038				-.002	.006		.0
22.5				-.038				-.001		-.003	22.5
45.0				-.038	.052	-.045	-.064	-.001	.002	.003	45.0
67.5				-.038	.003	-.028	-.064	-.001	.000	.009	67.5
90.0				-.038				-.010	.002	-.003	90.0
112.5				-.038	-.012	.005	-.019		-.003	-.001	112.5
135.0				-.022	-.013	.005		-.034	-.010	-.007	135.0
157.5				-.038	-.014	.005	.014	-.021	-.017	-.003	157.5
180.0				-.029	-.014	-.001	.027	-.014	-.019	-.007	180.0
202.5				-.033	-.015	.014	.010	-.013		-.013	202.5
225.0				-.033	-.010	.020	-.003	-.021	-.008	-.015	225.0
247.5				-.036	-.010	.016	-.020	-.015	.000	-.017	247.5
270.0				-.034				.001	.005		270.0
292.5				-.038	.070	-.015	-.065	.008	.005	-.009	292.5
315.0				-.038	.071	-.042	-.065	.001	.006	-.003	315.0
337.5				-.038				-.002		-.003	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.150	.109	.102	.069	.073	.126	.120	.121		.025
.075	.112	.118	.078	.068	.038	.051	.073	.089	.101		.075
.125	.093	.100	.052	.034	.019	.028	.041	.057	.077	.075	.125
.175	.082	.079	.027	.011	.009	.025	.019	.034	.053	.064	.175
.225	.075	.060	.015	-.007	.005	.009	.004	.018	.036	.038	.225
.275	.063	.048	-.002	-.017	-.012	-.006	-.007	.007	.017	.032	.275
.325	.062	.037	-.014	-.024	-.012	-.012	-.018	-.008	.007	.017	.325
.375		.037	-.021	-.028	-.027	-.025	-.024	-.015	-.004	.004	.375
.425		.032	-.030	-.037	-.038	-.027	-.030	-.024	-.014	-.005	.425
.475	.029	.020	-.039	-.041	-.045	-.031	-.041	-.033	-.024	-.019	.475
.550	.013	.007	-.053	-.059	-.058	-.055	-.052	-.049	-.041	-.034	.550
.650	-.008	-.028	-.076	-.065	-.069		-.066		-.060	-.053	.650
.750	-.020	-.022	-.048	-.085	-.064	-.046	-.076	-.078	-.072	-.060	.750
.800	-.027									-.063	.800
.850		-.043		-.085	-.052	-.082	-.076	-.078	-.075		.850
.900			-.040	-.085			-.076	-.078			.900
.950					-.058	-.081					.950
LOWER SURFACE											
.025		.145	.134	.143	.057	.073	.134	.147	.160		.025
.075		.118	.101	.082	.025	.034	.085	.114	.129	.111	.075
.125	.105	.085	.063	.049	.012	.007	.053	.078	.101	.092	.125
.175	.085	.064	.040	.027	.000	-.006	.024	.051	.078	.078	.175
.225	.056	.044	.025	.005	-.005	-.013	.002	.031	.060	.070	.225
.275	.051	.031	.019	-.006		-.017	-.007	.018	.046	.052	.275
.325	.034	.015	-.006		-.008	-.018	-.014	.004	.035	.043	.325
.375	.025	.007	-.013	-.021	-.017	-.018	-.025	-.002	.024	.032	.375
.425	.007	-.005	-.032	-.026	-.017	-.024	-.031	-.013	.012		.425
.475	.004	-.017	-.032	-.033	-.017	-.024	-.036	-.028	.004	.007	.475
.550	-.014	-.033	-.045	-.050	-.026	-.030	-.054	-.044	-.010	-.004	.550
.650	-.033	-.054	-.069	-.059		-.039	-.054	-.063	-.032	-.016	.650
.750	-.041	-.065	-.073	-.060	-.031	-.041	-.068		-.043	-.030	.750
.800	-.051									-.037	.800
.850		-.066	-.075	-.060	-.036	-.038	-.068	-.058	-.051		.850
.900			-.075	-.060			-.064	-.062			.900
.950					-.040	-.038					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 2.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.023				.014	.005		.0
22.5				-.024				.012		-.010	22.5
45.0				-.035	-.012	-.105	-.096	.008	-.001	-.001	45.0
67.5				-.041	.009	-.079	-.100	.005	-.008	.000	67.5
90.0				-.047				-.005	-.015	-.014	90.0
112.5				-.049	-.020	-.027	-.051		-.017	-.014	112.5
135.0				-.029	-.020	.005		-.061	-.028	-.014	135.0
157.5				-.042	-.015	.008	-.019	-.050	-.036	-.014	157.5
180.0				-.029	-.019	.005	-.002	-.038	-.041	-.015	180.0
202.5				-.041	-.016	.005	-.020	-.048		-.013	202.5
225.0				-.044	-.021	-.005	-.036	-.058	-.027	-.022	225.0
247.5				-.047	-.023	-.016	-.054	-.036	-.013	-.028	247.5
270.0				-.043				-.008	-.006		270.0
292.5				-.036	.029	-.075	-.105	.000	-.003	-.014	292.5
315.0				-.030	.001	-.103	-.080	.006	.001	-.007	315.0
337.5				-.023				.015		-.008	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		.017	-.006	-.007	-.012	-.017	.018	-.012	-.024		.025
.075	-.003	.001	-.015	-.024	-.021	-.026	-.030	-.011	-.013		.075
.125	-.012	-.014	-.043	-.053	-.043	-.038	-.049	-.027	-.017	-.021	.125
.175	-.026	-.031	-.049	-.071	-.063	-.046	-.082	-.044	-.030	-.030	.175
.225	-.026	-.040	-.057	-.082	-.065	-.052	-.082	-.057	-.043	-.050	.225
.275	-.040	-.052	-.076	-.089	-.080	-.075	-.082	-.065	-.070	-.054	.275
.325	-.053	-.058	-.079	-.105	-.076	-.075	-.086	-.073	-.069	-.060	.325
.375		-.071	-.094	-.097	-.097	-.083	-.090	-.083	-.070	-.079	.375
.425		-.080	-.099	-.112	-.104	-.096	-.104	-.097	-.079	-.079	.425
.475	-.077	-.085	-.103	-.108	-.104	-.080	-.094	-.098	-.085	-.098	.475
.550	-.092	-.098	-.123	-.117	-.116	-.105	-.105	-.124	-.112	-.108	.550
.650	-.096	-.108	-.135	-.121	-.116		-.121		-.118	-.118	.650
.750	-.102	-.121	-.128	-.141	-.108	-.090	-.130	-.147	-.129	-.112	.750
.800	-.094									-.103	.800
.850		-.121		-.125	-.089	-.110	-.122	-.137	-.123		.850
.900			-.125	-.124			-.110	-.136			.900
.950					-.089	-.099					.950
LOWER SURFACE											
.025		.259	.245	.225	.114	.131	.233	.252	.266		.025
.075		.210	.178	.155	.072	.083	.165	.193	.215	.202	.075
.125	.191	.160	.136	.113	.046	.051	.124	.156	.178	.179	.125
.175	.156	.134	.107	.088	.037	.037	.086	.122	.152	.157	.175
.225	.139	.109	.084	.064	.025	.030	.065	.101	.129	.143	.225
.275	.127	.102	.069	.040		.024	.050	.084	.112	.126	.275
.325	.100	.075	.051		.017	.018	.044	.060	.095	.114	.325
.375	.096	.063	.039	.020	.008	.018	.027	.057	.083	.101	.375
.425	.075	.053	.039	.020	.008	.015	.023	.045	.071		.425
.475	.070	.040	.020	.008	.005	.015	.018	.031	.060	.072	.475
.550	.051	.020	.005	-.008	-.001	.012	-.005	.017	.045	.058	.550
.650	.030	-.006	-.021	-.023		-.001	-.011	-.009	.013	.038	.650
.750	.015	-.014	-.027	-.024	-.007	-.004	-.021		.004	.023	.750
.800	.002									.012	.800
.850		-.020	-.034	-.019	-.001	-.004	-.021	-.013	.005		.850
.900			-.028	-.025			-.013	-.017			.900
.950					-.013	.000					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 5.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.008				.045	.023		.0
22.5				-.008				.037		-.001	22.5
45.0				-.036	-.075	-.167	-.111	.023	.013	-.002	45.0
67.5				-.050	-.001	-.140	-.127	-.005	-.007	-.002	67.5
90.0				-.069				-.007	-.021	-.019	90.0
112.5				-.073	-.038	-.059	-.080		-.023	-.019	112.5
135.0				-.044	-.026	-.015		-.072	-.026	-.019	135.0
157.5				-.050	-.016	.002	-.047	-.072	-.033	-.008	157.5
180.0				-.034	-.016	.002	-.028	-.064	-.040	-.009	180.0
202.5				-.052	-.013	-.006	-.048	-.075		-.016	202.5
225.0				-.055	-.027	-.028	-.061	-.064	-.028	-.030	225.0
247.5				-.073	-.040	-.057	-.079	-.028	-.029	-.047	247.5
270.0				-.062				-.019	-.015		270.0
292.5				-.048	-.009	-.134	-.122	-.002	.005	-.016	292.5
315.0				-.023	-.062	-.166	-.110	.023	.012	-.008	315.0
337.5				-.008				.036		.005	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.082	-.119	-.109	-.101	-.111	-.079	-.124	-.135		.025
.075	-.079	-.079	-.117	-.113	-.096	-.090	-.124	-.109	-.115		.075
.125	-.085	-.082	-.130	-.128	-.108	-.108	-.141	-.119	-.113	-.121	.125
.175	-.090	-.093	-.130	-.148	-.138	-.120	-.162	-.126	-.115	-.134	.175
.225	-.089	-.105	-.138	-.156	-.130	-.120	-.159	-.135	-.127	-.139	.225
.275	-.099	-.112	-.151	-.171	-.145	-.134	-.159	-.145	-.146	-.139	.275
.325	-.115	-.118	-.149	-.184	-.145	-.133	-.165	-.151	-.139	-.139	.325
.375		-.132	-.162	-.178	-.165	-.141	-.165	-.158	-.145	-.155	.375
.425		-.134	-.168	-.190	-.171	-.155	-.179	-.170	-.149	-.149	.425
.475	-.135	-.139	-.171	-.190	-.170	-.140	-.166	-.165	-.155	-.170	.475
.550	-.145	-.150	-.186	-.190	-.172	-.159	-.172	-.187	-.175	-.177	.550
.650	-.145	-.158	-.190	-.190	-.166		-.178		-.179	-.174	.650
.750	-.130	-.165	-.167	-.190	-.156	-.135	-.181	-.190	-.185	-.151	.750
.800	-.119									-.140	.800
.850		-.157		-.174	-.130	-.148	-.175	-.170	-.174		.850
.900			-.151	-.174			-.161	-.177			.900
.950					-.116	-.128					.950
LOWER SURFACE											
.025		.355	.340	.319	.182	.201	.316	.347	.355		.025
.075		.288	.257	.229	.131	.142	.235	.270	.292	.284	.075
.125	.267	.236	.203	.181	.095	.105	.186	.227	.253	.256	.125
.175	.238	.203	.173	.154	.082	.092	.147	.188	.220	.231	.175
.225	.211	.177	.150	.124	.066	.068	.122	.163	.192	.217	.225
.275	.197	.167	.140	.103		.058	.103	.148	.176	.196	.275
.325	.167	.142	.114		.062	.058	.098	.126	.157	.179	.325
.375	.161	.124	.097	.082	.057	.058	.081	.117	.143	.166	.375
.425	.136	.111	.094	.075	.057	.052	.072	.103	.129		.425
.475	.131	.096	.077	.060	.057	.051	.069	.084	.118	.134	.475
.550	.109	.073	.058	.040	.057	.044	.041	.070	.102	.114	.550
.650	.084	.049	.034	.027		.034	.045	.038	.067	.095	.650
.750	.072	.037	.025	.027	.037	.034	.025		.056	.075	.750
.800	.059									.069	.800
.850		.031	.015	.027	.041	.036	.019	.038	.043		.850
.900			.027	.024			.027	.037			.900
.950					.032	.037					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.019				.062	.038		.0
22.5				.008				.054		.008	22.5
45.0				-.028	-.118	-.206	-.132	.037	.015	-.002	45.0
67.5				-.057	-.038	-.183	-.143	.009	-.010	-.009	67.5
90.0				-.090				-.037	-.030	-.034	90.0
112.5				-.104	-.070	-.094	-.110		-.048	-.024	112.5
135.0				-.052	-.038	-.049		-.087	-.027	-.010	135.0
157.5				-.055	-.022	-.027	-.066	-.089	-.035	-.007	157.5
180.0				-.016	-.010	-.003	-.041	-.078	-.036	-.003	180.0
202.5				-.051	-.028	-.027	-.073	-.094		-.008	202.5
225.0				-.070	-.035	-.051	-.089	-.068	-.034	-.027	225.0
247.5				-.097	-.063	-.080	-.106	-.031	-.057	-.058	247.5
270.0				-.083				-.040	-.022		270.0
292.5				-.050	-.043	-.174	-.139	-.001	-.003	-.019	292.5
315.0				-.012	-.111	-.198	-.122	.043	.021	.005	315.0
337.5				.015				.062		.020	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.159	-.174	-.169	-.165	-.162	-.130	-.192	-.199		.025
.075	-.153	-.145	-.163	-.163	-.146	-.148	-.185	-.163	-.175		.075
.125	-.149	-.139	-.172	-.178	-.154	-.148	-.185	-.165	-.159	-.173	.125
.175	-.148	-.146	-.172	-.187	-.175	-.161	-.207	-.174	-.159	-.173	.175
.225	-.148	-.153	-.173	-.197	-.171	-.155	-.207	-.180	-.165	-.173	.225
.275	-.151	-.157	-.182	-.205	-.180	-.172	-.207	-.180	-.180	-.173	.275
.325	-.162	-.159	-.184	-.218	-.180	-.179	-.207	-.187	-.176	-.173	.325
.375		-.172	-.194	-.217	-.197	-.184	-.212	-.192	-.176	-.182	.375
.425		-.176	-.194	-.230	-.206	-.193	-.230	-.205	-.180	-.182	.425
.475	-.173	-.178	-.200	-.230	-.206	-.181	-.213	-.205	-.186	-.194	.475
.550	-.179	-.189	-.221	-.230	-.206	-.201	-.213	-.220	-.204	-.194	.550
.650	-.172	-.190	-.211	-.217	-.199		-.210		-.204	-.187	.650
.750	-.153	-.187	-.184	-.217	-.180	-.159	-.210	-.208	-.200	-.173	.750
.800	-.149									-.162	.800
.850		-.171		-.206	-.139	-.166	-.208	-.194	-.184		.850
.900			-.179	-.208			-.201	-.194			.900
.950					-.118	-.139					.950
LOWER SURFACE											
.025		.440	.427	.394	.245	.262	.393	.420	.434		.025
.075		.366	.333	.297	.182	.195	.301	.337	.364	.355	.075
.125	.346	.311	.277	.245	.148	.153	.246	.291	.319	.326	.125
.175	.313	.278	.244	.215	.133	.146	.207	.247	.289	.298	.175
.225	.284	.250	.219	.183	.117	.114	.180	.226	.260	.286	.225
.275	.269	.236	.200	.162		.102	.162	.200	.236	.259	.275
.325	.243	.210	.179		.107	.102	.153	.187	.217	.241	.325
.375	.228	.191	.162	.139	.103	.104	.139	.173	.200	.224	.375
.425	.205	.179	.153	.130	.108	.098	.127	.154	.186		.425
.475	.198	.159	.130	.116	.110	.092	.116	.135	.169	.192	.475
.550	.172	.135	.115	.103	.099	.088	.097	.118	.150	.170	.550
.650	.148	.105	.086	.082		.073	.089	.088	.115	.148	.650
.750	.135	.092	.077	.073	.083	.069	.069		.104	.131	.750
.800	.123									.121	.800
.850		.089	.077	.073	.086	.077	.060	.085	.091		.850
.900			.078	.073			.070	.079			.900
.950					.079	.079					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 10.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.050				.082	.059		.0
22.5				.041				.073		.013	22.5
45.0				-.014	-.142	-.236	-.153	.048	.009	-.010	45.0
67.5				-.071	-.075	-.227	-.171	.010	-.020	-.038	67.5
90.0				-.127				-.061	-.043	-.062	90.0
112.5				-.147	-.097	-.119	-.132		-.070	-.041	112.5
135.0				-.064	-.049	-.085		-.097	-.038	-.012	135.0
157.5				-.057	-.041	-.055	-.096	-.106	-.037	-.002	157.5
180.0				-.008	-.012	-.005	-.055	-.101	-.036	-.009	180.0
202.5				-.062	-.049	-.069	-.130	-.111		-.015	202.5
225.0				-.096	-.052	-.080	-.134	-.062	-.050	-.028	225.0
247.5				-.141	-.098	-.111	-.143	-.050	-.084	-.077	247.5
270.0				-.113				-.077	-.044		270.0
292.5				-.055	-.073	-.218	-.161	.015	-.013	-.047	292.5
315.0				-.001	-.140	-.234	-.151	.048	.020	-.001	315.0
337.5				.045				.082		.020	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.225	-.244	-.235	-.202	-.201	-.159	-.254	-.263		.025
.075	-.214	-.203	-.222	-.228	-.187	-.188	-.223	-.216	-.232		.075
.125	-.207	-.196	-.230	-.234	-.196	-.191	-.223	-.216	-.220	-.234	.125
.175	-.204	-.200	-.230	-.244	-.204	-.185	-.233	-.216	-.216	-.232	.175
.225	-.200	-.203	-.228	-.246	-.215	-.185	-.238	-.223	-.216	-.227	.225
.275	-.202	-.203	-.235	-.253	-.227	-.195	-.238	-.221	-.226	-.222	.275
.325	-.205	-.206	-.235	-.264	-.227	-.204	-.242	-.229	-.221	-.222	.325
.375		-.210	-.245	-.264	-.242	-.220	-.246	-.229	-.221	-.228	.375
.425		-.214	-.245	-.271	-.254	-.228	-.253	-.235	-.226	-.228	.425
.475	-.209	-.216	-.251	-.267	-.254	-.219	-.253	-.239	-.227	-.228	.475
.550	-.210	-.217	-.259	-.267	-.254	-.238	-.252	-.249	-.232	-.228	.550
.650	-.195	-.203	-.249	-.259	-.245		-.234		-.221	-.215	.650
.750	-.188	-.200	-.211	-.259	-.211	-.190	-.239	-.233	-.215	-.206	.750
.800	-.187									-.202	.800
.850		-.196		-.253	-.169	-.183	-.239	-.222	-.214		.850
.900			-.211	-.248			-.236	-.221			.900
.950					-.152	-.152					.950
LOWER SURFACE											
.025		.504	.490	.450	.224	.304	.455	.485	.504		.025
.075		.430	.388	.352	.187	.233	.355	.400	.435	.425	.075
.125	.405	.373	.335	.297	.168	.193	.303	.348	.388	.393	.125
.175	.376	.337	.301	.268	.155	.181	.256	.307	.354	.363	.175
.225	.343	.308	.272	.232	.181	.159	.231	.278	.324	.347	.225
.275	.327	.291	.251	.206		.148	.212	.259	.297	.321	.275
.325	.297	.263	.223		.141	.144	.195	.233	.277	.305	.325
.375	.284	.245	.207	.172	.141	.144	.186	.225	.263	.286	.375
.425	.263	.230	.201	.163	.143	.140	.174	.204	.242		.425
.475	.251	.211	.178	.144	.141	.134	.168	.183	.228	.250	.475
.550	.223	.185	.163	.121	.130	.125	.136	.170	.208	.228	.550
.650	.199	.154	.136	.108		.110	.128	.135	.169	.203	.650
.750	.182	.144	.123	.114	.110	.110	.107		.158	.186	.750
.800	.175									.176	.800
.850		.140	.116	.114	.121	.115	.101	.135	.144		.850
.900			.116				.108	.126			.900
.950					.116	.116					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 12.5°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.085				.114	.079		.0
22.5				.062				.091		.033	22.5
45.0				-.002	-.168	-.245	-.157	.051	.009	-.019	45.0
67.5				-.078	-.105	-.245	-.191	-.002	-.036	-.076	67.5
90.0				-.157				-.022	-.069	-.098	90.0
112.5				-.189	-.120	-.162	-.158		-.071	-.040	112.5
135.0				-.086	-.075	-.115		-.092	-.033	-.021	135.0
157.5				-.077	-.075	-.103	-.110	-.119	-.028	-.010	157.5
180.0				.020	-.009	-.005	-.068	-.106	-.028	-.019	180.0
202.5				-.072	-.091	-.129	-.165	-.122		-.020	202.5
225.0				-.100	-.084	-.121	-.172	-.079	-.045	-.034	225.0
247.5				-.189	-.134	-.158	-.175	-.077	-.090	-.087	247.5
270.0				-.139				-.085	-.071		270.0
292.5				-.068	-.094	-.228	-.182	.001	-.026	-.071	292.5
315.0				.006	-.160	-.246	-.157	.056	.021	.000	315.0
337.5				.070				.103		.041	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.256	-.263	-.264	-.235	-.229	-.184	-.270	-.280		.025
.075	-.248	-.240	-.248	-.254	-.221	-.219	-.248	-.248	-.249		.075
.125	-.243	-.231	-.245	-.253	-.232	-.229	-.248	-.248	-.245	-.242	.125
.175	-.235	-.231	-.245	-.256	-.224	-.202	-.248	-.248	-.240	-.242	.175
.225	-.233	-.230	-.245	-.261	-.219	-.213	-.260	-.248	-.241	-.242	.225
.275	-.231	-.230	-.245	-.264	-.232	-.222	-.260	-.247	-.237	-.242	.275
.325	-.219	-.232	-.245	-.264	-.237	-.231	-.267	-.248	-.244	-.242	.325
.375		-.231	-.245	-.270	-.249	-.238	-.267	-.248	-.244	-.230	.375
.425		-.231	-.251	-.269	-.264	-.242	-.261	-.247	-.244	-.232	.425
.475	-.210	-.231	-.260	-.264	-.256	-.245	-.270	-.255	-.244	-.225	.475
.550	-.207	-.218	-.251	-.270	-.263	-.260	-.263	-.254	-.229	-.223	.550
.650	-.203	-.210	-.245	-.266	-.245		-.260		-.230	-.223	.650
.750	-.203	-.209	-.227	-.260	-.211	-.218	-.251	-.238	-.230	-.223	.750
.800	-.204									-.223	.800
.850		-.207		-.261	-.173	-.215	-.251	-.240	-.223		.850
.900			-.228	-.253			-.259	-.240			.900
.950					-.161	-.171					.950
LOWER SURFACE											
.025		.552	.547	.512	.337	.361	.521	.552	.563		.025
.075		.491	.451	.416	.276	.290	.422	.470	.497	.485	.075
.125	.463	.437	.398	.360	.236	.246	.371	.414	.452	.456	.125
.175	.433	.400	.360	.326	.216	.232	.324	.372	.417	.428	.175
.225	.403	.368	.328	.297	.204	.207	.295	.345	.384	.412	.225
.275	.385	.349	.309	.266		.195	.272	.322	.357	.386	.275
.325	.361	.321	.285		.197	.198	.261	.302	.336	.365	.325
.375	.339	.304	.271	.240	.195	.199	.246	.285	.320	.347	.375
.425	.315	.288	.261		.201	.194	.236	.266	.303		.425
.475	.304	.270	.240	.221	.197	.186	.226	.249	.286	.312	.475
.550	.278	.236	.220	.198	.181	.178	.191	.225	.269	.286	.550
.650	.252	.210	.185	.168		.162	.182	.186	.221	.263	.650
.750	.237	.194	.172	.162	.166	.162	.157		.215	.242	.750
.800	.231									.236	.800
.850		.186	.168	.167	.169	.162	.154	.187	.198		.850
.900			.168	.167			.160	.178			.900
.950					.169	.162					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 15.0°$ $\beta = 0°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.121				.156	.106		.0
22.5				.091				.131		.040	22.5
45.0				.009	-.169	-.238	-.154	.082	-.002	-.021	45.0
67.5				-.089	-.114	-.240	-.180	.002	-.058	-.113	67.5
90.0				-.169				-.093	-.106	-.135	90.0
112.5				-.206	-.129	-.177	-.204		-.072	-.057	112.5
135.0				-.097	-.141	-.184		-.057	-.040	-.040	135.0
157.5				-.089	-.100	-.121	-.133	-.126	-.047	-.023	157.5
180.0				.049	-.002	-.021	-.085	-.105	-.040	-.034	180.0
202.5				-.096	-.117	-.196	-.174	-.156		-.037	202.5
225.0				-.126	-.132	-.167	-.213	-.105	-.054	-.051	225.0
247.5				-.204	-.143	-.187	-.183	-.110	-.124	-.078	247.5
270.0				-.147				-.103	-.112		270.0
292.5				-.068	-.106	-.223	-.180	-.007	-.051	-.101	292.5
315.0				.028	-.157	-.240	-.164	.073	.017	-.002	315.0
337.5				.098				.135		.057	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.267	-.288	-.287	-.247	-.244	-.193	-.284	-.286		.025
.075	-.254	-.262	-.273	-.275	-.247	-.238	-.248	-.270	-.252		.075
.125	-.251	-.252	-.258	-.268	-.247	-.251	-.247	-.266	-.268	-.251	.125
.175	-.244	-.254	-.273	-.280	-.209	-.202	-.236	-.266	-.257	-.235	.175
.225	-.244	-.251	-.274	-.279	-.230	-.225	-.254	-.266	-.260	-.235	.225
.275	-.241	-.251	-.266	-.280	-.245	-.225	-.273	-.266	-.237	-.235	.275
.325	-.225	-.245	-.279	-.267	-.245	-.243	-.273	-.266	-.260	-.244	.325
.375		-.237	-.266	-.282	-.254	-.249	-.273	-.266	-.258	-.232	.375
.425		-.232	-.264	-.273	-.275	-.238	-.260	-.250	-.258	-.242	.425
.475	-.231	-.230	-.277	-.273	-.268	-.254	-.276	-.266	-.250	-.230	.475
.550	-.229	-.224	-.262	-.287	-.274	-.267	-.275	-.256	-.237	-.229	.550
.650	-.237	-.231	-.260	-.279	-.243		-.269		-.244	-.229	.650
.750	-.237	-.224	-.238	-.270	-.226	-.235	-.256	-.243	-.244	-.229	.750
.800	-.236									-.236	.800
.850		-.230		-.273	-.194	-.235	-.256	-.258	-.244		.850
.900			-.234	-.262			-.267	-.251			.900
.950					-.187	-.187					.950
LOWER SURFACE											
.025		.604	.596	.561	.386	.411	.575	.600	.627		.025
.075		.545	.507	.468	.320	.337	.480	.527	.567	.553	.075
.125	.513	.497	.457	.414	.281	.290	.429	.475	.520	.523	.125
.175	.491	.462	.417	.371	.264	.271	.380	.434	.487	.498	.175
.225	.462	.430	.395	.347	.251	.258	.353	.402	.455	.481	.225
.275	.444	.403	.370	.326		.260	.332	.380	.428	.451	.275
.325	.416	.383	.350		.242	.254	.320	.358	.404	.457	.325
.375	.395	.365	.331	.297	.250	.250	.306	.342	.408	.437	.375
.425	.366	.352	.315	.284	.251	.243	.286	.324	.389		.425
.475	.357	.328	.307	.273	.243	.236	.276	.302	.370	.398	.475
.550	.328	.290	.268	.271	.222	.229	.242	.278	.352	.373	.550
.650	.302	.269	.229	.218		.210	.241	.241	.290	.329	.650
.750	.286	.252	.216	.205	.211	.210	.205		.299	.314	.750
.800	.292									.304	.800
.850		.241	.216	.205	.210	.211	.205	.270	.262		.850
.900			.216	.211			.205	.261			.900
.950					.206	.212					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.041				.002	-.005		.0
22.5				-.033				.001	.001	-.008	22.5
45.0				-.022	.207	-.002	-.049	.011	.005	.004	45.0
67.5				-.013	.056	.026	-.050	.011	.000	.018	67.5
90.0				-.009				.002	-.009	-.012	90.0
112.5				-.020	-.002	.021	-.014		-.015	-.018	112.5
135.0				-.018	-.028	.016	-.021	-.033	-.030	-.047	135.0
157.5				-.049	-.039	-.012	-.027	-.046	-.049	-.054	157.5
180.0				-.068	-.046	-.026	-.013	-.055	-.062	-.058	180.0
202.5				-.062	-.033	-.002	-.018	-.056		-.041	202.5
225.0				-.058	-.026	-.001	-.027	-.063	-.035	-.025	225.0
247.5				-.051	-.015	-.018	-.037	-.041	-.028	-.021	247.5
270.0				-.044				-.014	-.016		270.0
292.5				-.055	.006	-.048	-.067	-.008	-.008	-.008	292.5
315.0				-.060	-.048	-.056	-.049	-.004		.000	315.0
337.5				-.060				.006	-.005	-.007	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.173	.149	.178	.214	-.106	.004	.009	.020		.025
.075	.122	.130	.102	.132	.152	-.091	-.047	.000	-.008		.075
.125	.103	.103	.064	.091	.106	-.087	-.070	-.017	-.002	-.021	.125
.175	.084	.076	.051	.058	.063	-.108	-.104		-.020	-.034	.175
.225	.074	.056	.038	.031	.063	-.086	-.095	-.051	-.033	-.051	.225
.275	.057	.043	.013	.014	.050	-.079	-.083	-.065	-.074	-.053	.275
.325	.037	.032	.012	-.009	.028	-.079	-.083	-.076	-.060	-.043	.325
.375		.014	-.019	-.009	.005	-.080	-.090	-.082	-.067	-.084	.375
.425		.006	-.028	-.028	.005	-.093	-.104		-.076	-.061	.425
.475	.001	-.002	-.028	-.039	-.009	-.082	-.089	-.093	-.086	-.103	.475
.550	-.016	-.018	-.054	-.050	-.026	-.082	-.098	-.124	-.117	-.110	.550
.650	-.019	-.032	-.072	-.060	-.056		-.106		-.113	-.118	.650
.750	-.037	-.050	-.051	-.095	-.083	-.100	-.119	-.139	-.128	-.121	.750
.800	-.038									-.105	.800
.850		-.048	-.065	-.085	-.076	-.092	-.111	-.122	-.128		.850
.900			-.065	-.095			-.097	-.130			.900
.950					-.084	-.082					.950
LOWER SURFACE											
.025		.187	.180	.139	.015	.116	.131	.120	.123		.025
.075		.154	.136	.103	-.017	.081	.089	.087	.097	.080	.075
.125	.140	.123	.102	.063	-.045	.049	.056	.063	.077	.066	.125
.175	.110	.096	.075	.036	-.050	.037	.030	.037	.052	.054	.175
.225	.094	.072	.057	.017	-.064	.033	.023	.019	.036	.046	.225
.275	.076	.056	.036	-.009	-.064	.033	.011	.012	.027	.029	.275
.325	.058	.043	.019	-.021	-.066	.021	.008	.002	.016	.025	.325
.375	.050	.034	.007	-.037	-.064	.017	.002	-.006	.008	.019	.375
.425	.036	.024	.007	-.037	-.055	.008	-.011	-.021	-.001		.425
.475	.028	.012	-.009	-.051	-.055	-.006	-.011	-.021	-.008	-.007	.475
.550	.009	-.008	-.028	-.066	-.055	-.015	-.032	-.037	-.020	-.016	.550
.650	-.008	-.036	-.059	-.088		-.031	-.032	-.052	-.039	-.032	.650
.750	-.023	-.046	-.064	-.088	-.056	-.039	-.049	-.042	-.046	-.044	.750
.800	-.033									-.046	.800
.850		-.046	-.074	-.088	-.037	-.033	-.050	-.043	-.052		.850
.900			-.074	-.089			-.049	-.045			.900
.950					-.030	-.033					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 2.5°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.036				.013	.004		.0
22.5				-.011				.026	.015	.005	22.5
45.0				.000	.139	-.055	-.077	.030	.022	.021	45.0
67.5				-.006	.051	-.026	-.089	.016	.006	.014	67.5
90.0				-.015				.005	-.012	-.016	90.0
112.5				-.035	-.021	-.026	-.049		-.029	-.035	112.5
135.0				-.032	-.049	-.029	-.049	-.055	-.055	-.061	135.0
157.5				-.065	-.050	-.021	-.042	-.074	-.070	-.058	157.5
180.0				-.069	-.035	-.016	-.030	-.082	-.071	-.042	180.0
202.5				-.055	-.021	.006	-.036	-.078		-.033	202.5
225.0				-.046	-.015	-.011	-.046	-.084	-.043	-.035	225.0
247.5				-.043	-.015	-.036	-.058	-.049	-.028	-.026	247.5
270.0				-.056				-.005	-.007		270.0
292.5				-.064	-.036	-.089	-.095	.001	-.002	-.002	292.5
315.0				-.074	-.116	-.097	-.068	.007		-.005	315.0
337.5				-.051				.007	-.002	-.014	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.090	.074	.079	.133	-.169	-.070	-.107	-.120		.025
.075	.056	.061	.044	.058	.098	-.154	-.124	-.093	-.106		.075
.125	.037	.038	.007	.028	.056	-.142	-.133	-.093	-.091	-.106	.125
.175	.023	.015	-.001	.005	.007	-.155	-.154		-.089	-.106	.175
.225	.017	-.004	-.011	-.007	.018	-.135	-.154	-.107	-.098	-.119	.225
.275	.002	-.014	-.030	-.028	.005	-.123	-.141	-.122	-.130	-.122	.275
.325	-.015	-.025	-.035	-.052	-.013	-.123	-.141	-.122	-.120	-.115	.325
.375		-.040	-.051	-.043	-.030	-.118	-.141	-.131	-.120	-.132	.375
.425		-.047	-.059	-.061	-.037	-.124	-.149		-.120	-.124	.425
.475	-.046	-.053	-.059	-.068	-.046	-.113	-.132	-.142	-.129	-.145	.475
.550	-.062	-.071	-.079	-.084	-.058	-.109	-.132	-.169	-.156	-.154	.550
.650	-.065	-.082	-.091	-.096	-.084		-.133		-.156	-.157	.650
.750	-.070	-.096	-.097	-.122	-.103	-.115	-.149	-.167	-.156	-.150	.750
.800	-.067									-.133	.800
.850		-.086	-.106	-.107	-.103	-.109	-.144	-.150	-.156		.850
.900			-.105	-.112			-.130	-.156			.900
.950					-.100	-.094					.950
LOWER SURFACE											
.025		.313	.300	.249	.084	.165	.216	.215	.219		.025
.075		.253	.234	.189	.046	.119	.154	.161	.176	.166	.075
.125	.230	.214	.185	.140	.023	.084	.116	.122	.149	.145	.125
.175	.204	.179	.152	.106	.011	.076	.085	.095	.121	.126	.175
.225	.177	.157	.129	.075	-.001	.068	.071	.077	.102	.118	.225
.275	.161	.140	.106	.058	-.001	.077	.062	.071	.088	.099	.275
.325	.135	.114	.088	.039	-.008	.057	.055	.051	.077	.090	.325
.375	.125	.102	.076	.027	-.008	.047	.045	.043	.066	.081	.375
.425	.104	.088	.064	.021	-.015	.042	.036	.036	.053		.425
.475	.093	.077	.047	.008	-.017	.033	.031	.028	.046	.050	.475
.550	.076	.051	.025	-.013	-.025	.026	.014	.015	.035	.037	.550
.650	.055	.026	-.013	-.044		.002	.014	-.013	.007	.018	.650
.750	.036	.013	-.019	-.044	-.014	.002	-.012	-.004	.002	.011	.750
.800	.023									.004	.800
.850		.005	-.030	-.044	.000	.002	-.012	.001	-.009		.850
.900			-.030	-.044			-.012	-.004			.900
.950					.002	.002					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 5.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.019				.018	.007		.0
22.5				.006				.035	.025	.012	22.5
45.0				.014	.071	-.105	-.111	.042	.032	.025	45.0
67.5				-.013	.042	-.082	-.127	.000	-.005	.020	67.5
90.0				-.033				-.020	-.021	-.032	90.0
112.5				-.063	-.055	-.080	-.092		-.050	-.062	112.5
135.0				-.062	-.078	-.074	-.077	-.085	-.080	-.083	135.0
157.5				-.089	-.062	-.034	-.068	-.101	-.092	-.067	157.5
180.0				-.074	-.039	-.018	-.055	-.110	-.069	-.039	180.0
202.5				-.058	-.022	-.019	-.077	-.105		-.036	202.5
225.0				-.049	-.013	-.029	-.073	-.110	-.057	-.041	225.0
247.5				-.062	-.027	-.084	-.098	-.044	-.023	-.015	247.5
270.0				-.080				-.002	-.006		270.0
292.5				-.091	-.085	-.136	-.111	-.014	-.011	-.015	292.5
315.0				-.081	-.188	-.141	-.084	-.001		-.025	315.0
337.5				-.048				.000	-.011	-.019	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.001	-.028	-.027	.034	-.251	-.149	-.217	-.227		.025
.075	-.020	-.013	-.041	-.039	.034	-.232	-.207	-.195	-.204		.075
.125	-.028	-.026	-.083	-.069	-.008	-.227	-.215	-.189	-.191	-.200	.125
.175	-.038	-.043	-.083	-.084	-.053	-.234	-.224		-.191	-.200	.175
.225	-.040	-.059	-.095	-.092	-.054	-.222	-.225	-.196	-.189	-.213	.225
.275	-.056	-.069	-.110	-.111	-.070	-.205	-.225	-.192	-.206	-.207	.275
.325	-.073	-.076	-.111	-.137	-.079	-.192	-.224	-.202	-.204	-.194	.325
.375		-.095	-.128	-.122	-.098	-.180	-.224	-.204	-.204	-.208	.375
.425		-.101	-.128	-.141	-.107	-.188	-.224		-.204	-.200	.425
.475	-.100	-.106	-.133	-.140	-.115	-.163	-.207	-.215	-.204	-.213	.475
.550	-.113	-.117	-.154	-.150	-.123	-.150	-.207	-.228	-.219	-.211	.550
.650	-.109	-.129	-.155	-.159	-.141		-.208		-.208	-.208	.650
.750	-.101	-.145	-.150	-.175	-.154	-.151	-.220	-.218	-.204	-.198	.750
.800	-.093									-.185	.800
.850		-.122	-.148	-.155	-.154	-.138	-.220	-.204	-.199		.850
.900			-.143	-.155			-.204	-.209			.900
.950					-.137	-.121					.950
LOWER SURFACE											
.025		.419	.403	.348	.152	.206	.282	.278	.291		.025
.075		.345	.313	.261	.103	.152	.208	.220	.236	.229	.075
.125	.309	.289	.254	.201	.072	.119	.166	.177	.209	.204	.125
.175	.274	.251	.214	.165	.057	.105	.134	.149	.178	.184	.175
.225	.242	.217	.183	.134	.046	.096	.113	.130	.154	.176	.225
.275	.228	.197	.163	.103	.046	.088	.103	.112	.142	.153	.275
.325	.204	.175	.143	.093	.031	.083	.088	.101	.127	.136	.325
.375	.184	.158	.126	.080	.025	.076	.088	.090	.114	.119	.375
.425	.168	.145	.114	.068	.025	.068	.071	.078	.100		.425
.475	.152	.128	.093	.051	.012	.061	.070	.065	.091	.097	.475
.550	.131	.096	.069	.037	.012	.055	.039	.048	.071	.078	.550
.650	.105	.067	.036	.007		.034	.042	.025	.049	.066	.650
.750	.081	.057	.024	-.006	.021	.024	.014	.033	.041	.053	.750
.800	.069									.047	.800
.850		.045	.018	-.001	.026	.029	.014	.028	.030		.850
.900			.018	-.001			.017	.023			.900
.950					.033	.029					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 7.5°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.018				.046	.033		.0
22.5				.043				.069	.051	.032	22.5
45.0				.033	.034	-.140	-.125	.076	.049	.036	45.0
67.5				.006	.060	-.113	-.152	.027	.004	.019	67.5
90.0				-.039				-.022	-.021	-.040	90.0
112.5				-.089	-.092	-.132	-.101		-.058	-.077	112.5
135.0				-.089	-.104	-.082	-.084	-.074	-.102	-.101	135.0
157.5				-.098	-.048	-.032	-.074	-.104	-.101	-.055	157.5
180.0				-.064	-.030	-.026	-.069	-.116	-.067	-.029	180.0
202.5				-.046	-.004	-.007	-.067	-.124		-.030	202.5
225.0				-.050	-.014	-.043	-.077	-.111	-.036	-.018	225.0
247.5				-.074	-.036	-.113	-.127	-.033	-.007	-.008	247.5
270.0				-.103				.006	-.001		270.0
292.5				-.109	-.134	-.165	-.109	-.006	-.012	-.025	292.5
315.0				-.076	-.220	-.167	-.083	.014		-.030	315.0
337.5				-.025				.029	.005	-.009	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.088	-.121	-.107	-.033	-.280	-.182	-.272	-.272		.025
.075	-.091	-.081	-.114	-.110	-.032	-.267	-.251	-.250	-.247		.075
.125	-.092	-.083	-.143	-.125	-.062	-.269	-.249	-.246	-.243	-.234	.125
.175	-.097	-.095	-.135	-.140	-.098	-.262	-.249		-.237	-.234	.175
.225	-.095	-.106	-.142	-.154	-.097	-.269	-.258	-.244	-.236	-.234	.225
.275	-.104	-.115	-.161	-.163	-.112	-.245	-.258	-.238	-.233	-.234	.275
.325	-.123	-.121	-.166	-.182	-.123	-.249	-.258	-.246	-.241	-.234	.325
.375		-.135	-.176	-.180	-.137	-.236	-.258	-.244	-.237	-.231	.375
.425		-.140	-.186	-.194	-.142	-.227	-.253		-.238	-.236	.425
.475	-.139	-.143	-.187	-.194	-.149	-.199	-.253	-.249	-.238	-.226	.475
.550	-.150	-.156	-.199	-.204	-.161	-.192	-.253	-.249	-.234	-.224	.550
.650	-.139	-.166	-.208	-.204	-.180		-.251		-.225	-.221	.650
.750	-.120	-.173	-.179	-.212	-.179	-.167	-.251	-.237	-.222	-.218	.750
.800	-.115									-.215	.800
.850		-.148	-.166	-.195	-.175	-.149	-.244	-.236	-.222		.850
.900			-.166	-.195			-.246	-.236			.900
.950					-.163	-.134					.950
LOWER SURFACE											
.025		.527	.507	.437	.232	.248	.346	.359	.359		.025
.075		.435	.402	.341	.182	.196	.276	.290	.302	.299	.075
.125	.403	.378	.336	.282	.141	.163	.233	.253	.271	.271	.125
.175	.365	.330	.295	.241	.124	.151	.197	.222	.242	.249	.175
.225	.333	.297	.263	.208	.107	.133	.179	.197	.226	.236	.225
.275	.311	.274	.234	.176	.107	.133	.166	.183	.203	.216	.275
.325	.287	.254	.210	.163	.094	.128	.157	.170	.189	.202	.325
.375	.270	.230	.192	.148	.087	.121	.148	.160	.171	.186	.375
.425	.248	.217	.183	.137	.087	.114	.133	.140	.158		.425
.475	.233	.201	.162	.120	.075	.105	.119	.128	.147	.156	.475
.550	.202	.166	.137	.100	.074	.095	.090	.107	.127	.139	.550
.650	.173	.137	.097	.065		.078	.088	.081	.099	.126	.650
.750	.151	.121	.086	.058	.068	.078	.068	.082	.088	.106	.750
.800	.139									.105	.800
.850		.112	.078	.058	.080	.078	.068	.078	.083		.850
.900			.078	.058			.074	.074			.900
.950					.087	.078					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 10.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.058				.091	.080		.0
22.5				.086				.105		.065	22.5
45.0				.055	-.006	-.163	-.165	.097	.076	.056	45.0
67.5				.007	.062	-.149	-.158	.035	.033	.023	67.5
90.0				-.058				-.042	-.014	-.041	90.0
112.5				-.125	-.143	-.150	-.104		-.054	-.083	112.5
135.0				-.118	-.116	-.088	-.097	-.057	-.113	-.105	135.0
157.5				-.088	-.058	-.054	-.099	-.121	-.075	-.022	157.5
180.0				-.017	-.045	-.077	-.123	-.154	-.043	-.007	180.0
202.5				-.045	.000	-.002	-.061	-.090	-.031	-.023	202.5
225.0				-.058	-.040	-.102	-.092	-.090	-.014	.000	225.0
247.5				-.095	-.049	-.143	-.211	.003	.013	.000	247.5
270.0				-.147				-.001	.013		270.0
292.5				-.126	-.169	-.214	-.122	.040	.001	.007	292.5
315.0				-.059	-.233	-.201	-.102	.054	.008	-.021	315.0
337.5				.012				.063	.050	.007	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.157	-.170	-.154	-.070	-.266	-.179	-.279	-.284		.025
.075	-.157	-.142	-.155	-.154	-.050	-.261	-.246	-.270	-.247		.075
.125	-.152	-.134	-.166	-.166	-.077	-.266	-.240	-.264	-.265	-.238	.125
.175	-.150	-.142	-.167	-.177	-.097	-.250	-.230		-.256	-.240	.175
.225	-.147	-.151	-.174	-.183	-.123	-.265	-.247	-.263	-.254	-.240	.225
.275	-.154	-.156	-.182	-.194	-.138	-.243	-.265	-.254	-.237	-.238	.275
.325	-.161	-.160	-.188	-.202	-.149	-.261	-.265	-.263	-.257	-.243	.325
.375		-.168	-.198	-.208	-.160	-.246	-.265	-.263	-.248	-.226	.375
.425		-.174	-.198	-.213	-.167	-.226	-.245		-.247	-.237	.425
.475	-.176	-.175	-.202	-.220	-.179	-.219	-.257	-.259	-.244	-.224	.475
.550	-.178	-.181	-.208	-.227	-.188	-.205	-.252	-.248	-.234	-.224	.550
.650	-.164	-.194	-.218	-.222	-.201		-.252		-.238	-.230	.650
.750	-.150	-.192	-.190	-.216	-.195	-.174	-.241	-.230	-.238	-.228	.750
.800	-.147									-.233	.800
.850		-.174	-.181	-.212	-.190	-.153	-.238	-.243	-.237		.850
.900			-.182	-.208			-.250	-.233			.900
.950					-.177	-.121					.950
LOWER SURFACE											
.025		.610	.602	.514	.307	.268	.402	.411	.419		.025
.075		.518	.480	.418	.243	.227	.329	.344	.364	.362	.075
.125	.484	.455	.413	.345	.202	.195	.281	.312	.334	.334	.125
.175	.443	.411	.361	.303	.178	.185	.249	.275	.301	.309	.175
.225	.404	.381	.330	.271	.169	.176	.230	.250	.287	.298	.225
.275	.389	.349	.309	.245	.157	.172	.215	.236	.259	.277	.275
.325	.360	.325	.278	.223	.147	.167	.205	.220	.243	.264	.325
.375	.338	.304	.259	.209	.139	.160	.193	.212	.230	.250	.375
.425	.319	.285	.242	.193	.135	.156	.176	.193	.213		.425
.475	.297	.267	.226	.178	.131	.146	.169	.180	.200	.214	.475
.550	.268	.229	.193	.148	.131	.137	.142	.154	.181	.196	.550
.650	.237	.197	.153	.119		.119	.142	.120	.146	.178	.650
.750	.218	.180	.134	.108	.114	.119	.110	.129	.140	.161	.750
.800	.207									.156	.800
.850		.175	.135	.112	.123	.119	.110	.122	.131		.850
.900			.131	.112			.110	.121			.900
.950					.133	.119					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 12.5°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.098				.126	.094		.0
22.5				.117				.149		.072	22.5
45.0				.079	-.024	-.174	-.177	.129	.066	.048	45.0
67.5				.013	.031	-.163	-.156	.035	.013	-.007	67.5
90.0				-.069				-.050	-.041	-.085	90.0
112.5				-.139	-.174	-.133	-.117		-.101	-.139	112.5
135.0				-.130	-.121	-.133	-.127	-.119	-.141	-.066	135.0
157.5				-.091	-.105	-.110	-.161	-.147	-.091	-.041	157.5
180.0				.002	-.080	-.111	-.133	-.156	-.055	-.048	180.0
202.5				-.043	-.001	.000	-.080	-.115	-.027	-.037	202.5
225.0				-.078	-.097	-.154	-.111	-.080	-.033	-.008	225.0
247.5				-.112	-.096	-.177	-.247	-.036	-.008	-.015	247.5
270.0				-.171				-.058	-.016		270.0
292.5				-.129	-.167	-.209	-.126	.001	-.035	-.035	292.5
315.0				-.051	-.224	-.197	-.112	.059	-.022	-.044	315.0
337.5				.024				.077	.045	.007	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.203	-.213	-.199	-.128	-.271	-.187	-.277	-.269		.025
.075	-.200	-.184	-.202	-.194	-.104	-.270	-.249	-.272	-.247		.075
.125	-.197	-.179	-.184	-.197	-.111	-.277	-.246	-.272	-.268	-.221	.125
.175	-.191	-.181	-.200	-.212	-.117	-.256	-.228		-.260	-.223	.175
.225	-.187	-.184	-.210	-.220	-.150	-.268	-.255	-.268	-.263	-.223	.225
.275	-.190	-.185	-.210	-.223	-.163	-.255	-.277	-.268	-.227	-.230	.275
.325	-.178	-.191	-.220	-.217	-.176	-.272	-.277	-.273	-.263	-.258	.325
.375		-.187	-.220	-.231	-.176	-.264	-.277	-.271	-.259	-.227	.375
.425		-.192	-.230	-.219	-.195	-.238	-.250		-.253	-.257	.425
.475	-.192	-.194	-.230	-.236	-.200	-.249	-.271	-.272	-.258	-.226	.475
.550	-.189	-.193	-.221	-.253	-.210	-.239	-.270	-.259	-.245	-.226	.550
.650	-.184	-.210	-.233	-.238	-.220		-.270		-.260	-.242	.650
.750	-.176	-.198	-.200	-.231	-.213	-.213	-.247	-.240	-.259	-.234	.750
.800	-.172									-.253	.800
.850		-.192	-.191	-.233	-.208	-.194	-.247	-.262	-.262		.850
.900			-.195	-.219			-.275	-.243			.900
.950					-.199	-.135					.950
LOWER SURFACE											
.025		.675	.657	.580	.369	.304	.460	.466	.472		.025
.075		.584	.548	.476	.295	.260	.386	.407	.425	.415	.075
.125	.551	.521	.481	.407	.265	.229	.338	.371	.393	.387	.125
.175	.511	.475	.432	.362	.236	.222	.306	.333	.357	.365	.175
.225	.471	.439	.394	.329	.214	.222	.288	.312	.340	.354	.225
.275	.458	.409	.363	.299	.205	.222	.272	.298	.318	.333	.275
.325	.428	.387	.337	.279	.191	.217	.259	.278	.300	.318	.325
.375	.400	.363	.318	.265	.191	.208	.243	.265	.286	.303	.375
.425	.377	.339	.303	.249	.189	.201	.227	.252	.273		.425
.475	.358	.323	.282	.230	.179	.192	.218	.237	.257	.272	.475
.550	.328	.279	.249	.209	.179	.186	.191	.209	.236	.248	.550
.650	.297	.253	.210	.170		.170	.189	.180	.199	.228	.650
.750	.277	.235	.191	.163	.163	.170	.159	.183	.193	.214	.750
.800	.263									.211	.800
.850		.224	.182	.163	.167	.170	.159	.177	.181		.850
.900			.182	.163			.159	.175			.900
.950					.177	.170					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 15.0°$ $\beta = -5°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.116				.153	.103		.0
22.5				.140				.169	.110	.074	22.5
45.0				.099	-.059	-.197	-.174	.131	.062	.040	45.0
67.5				.006	.005	-.191	-.192	.029	-.018	-.038	67.5
90.0				-.096				-.082	-.089	-.138	90.0
112.5				-.169	-.199	-.149	-.149		-.130	-.176	112.5
135.0				-.156	-.133	-.163	-.182	-.135	-.187	-.087	135.0
157.5				-.114	-.165	-.184	-.198	-.164	-.120	-.089	157.5
180.0				-.062	-.117	-.116	-.154	-.174	-.056	-.062	180.0
202.5				-.063	-.027	-.049	-.113	-.141		-.050	202.5
225.0				-.109	-.175	-.205	-.188	-.094	-.050	-.035	225.0
247.5				-.156	-.131	-.223	-.269	-.087	-.035	-.054	247.5
270.0				-.191				-.134	-.045		270.0
292.5				-.141	-.197	-.234	-.164	-.031	-.122	-.128	292.5
315.0				-.050	-.238	-.224	-.148	.031		-.080	315.0
337.5				.045				.088	.042	.008	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.205	-.202	-.199	-.122	-.248	-.160	-.245	-.232		.025
.075	-.205	-.195	-.202	-.199	-.123	-.244	-.209	-.239	-.203		.075
.125	-.204	-.194	-.150	-.193	-.116	-.244	-.209	-.237	-.231	-.184	.125
.175	-.198	-.192	-.193	-.212	-.093	-.219	-.195		-.228	-.190	.175
.225	-.198	-.194	-.215	-.228	-.153	-.238	-.218	-.238	-.234	-.190	.225
.275	-.200	-.195	-.184	-.219	-.166	-.226	-.252	-.240	-.189	-.189	.275
.325	-.172	-.194	-.219	-.176	-.175	-.244	-.247	-.242	-.231	-.226	.325
.375		-.173	-.186	-.221	-.164	-.238	-.244	-.242	-.234	-.195	.375
.425		-.186	-.201	-.196	-.194	-.214	-.216		-.235	-.224	.425
.475	-.186	-.188	-.215	-.212	-.194	-.231	-.240	-.241	-.235	-.192	.475
.550	-.173	-.182	-.202	-.245	-.205	-.222	-.245	-.219	-.213	-.197	.550
.650	-.186	-.203	-.221	-.222	-.214		-.245		-.242	-.205	.650
.750	-.182	-.181	-.200	-.212	-.214	-.211	-.222	-.214	-.239	-.205	.750
.800	-.182									-.226	.800
.850		-.195	-.183	-.226	-.209	-.196	-.220	-.239	-.238		.850
.900			-.187	-.203			-.242	-.216			.900
.950					-.199	-.169					.950
LOWER SURFACE											
.025		.737	.721	.649	.439	.341	.491	.516	.513		.025
.075		.649	.612	.544	.360	.309	.436	.465	.472	.461	.075
.125	.613	.587	.543	.478	.315	.273	.394	.429	.444	.434	.125
.175	.582	.545	.501	.428	.298	.273	.360	.398	.411	.417	.175
.225	.539	.506	.467	.394	.278	.270	.335	.372	.391	.406	.225
.275	.525	.473	.438	.368	.268	.270	.323	.358	.369	.386	.275
.325	.490	.447	.404	.346	.254	.270	.312	.335	.352	.368	.325
.375	.466	.426	.384	.329	.254	.253	.296	.320	.337	.354	.375
.425	.442	.405	.367	.312	.248	.248	.277	.308	.320		.425
.475	.416	.386	.346	.301	.248	.239	.277	.290	.307	.320	.475
.550	.386	.337	.311	.272	.240	.230	.241	.256	.280	.297	.550
.650	.353	.312	.272	.236		.217	.241	.235	.246	.280	.650
.750	.335	.302	.252	.227	.217	.217	.218	.232	.243	.272	.750
.800	.327									.270	.800
.850		.289	.252	.227	.224	.217	.218	.226	.235		.850
.900			.246	.227			.219	.223			.900
.950					.231	.217					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.092				-.002			.0
22.5				-.052				.014		.007	22.5
45.0				-.002	.362	.067	-.009	.033	.016	.013	45.0
67.5				.038	.137	.088	.000	.028	.009	.028	67.5
90.0				.045				.012	.008	.002	90.0
112.5				.026	.025	.046	-.006		-.016	-.007	112.5
135.0				-.001	-.027	.005	-.049	-.066	-.057	-.061	135.0
157.5				-.066	-.088	-.066	-.110	-.101	-.109	-.111	157.5
180.0				-.104	-.113	-.092	-.094	-.124	-.147	-.079	180.0
202.5				-.103	-.074	-.031	-.065	-.115		-.053	202.5
225.0				-.076	-.038	-.039	-.052	-.090	-.066	-.057	225.0
247.5				-.052	-.028	-.061	-.077	-.059	-.066	-.087	247.5
270.0				-.045				-.035	-.033		270.0
292.5				-.063	-.034	-.083	-.066	-.002	-.005	-.002	292.5
315.0				-.086	-.076	-.063	-.049	.005	.007	-.004	315.0
337.5				-.110				.013	-.002	-.014	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.234	.223	.277	.395	-.170	-.038	-.033	-.031		.025
.075	.179	.186	.179	.219	.293	-.143	-.089	-.024	-.037	-.041	.075
.125	.155	.156	.139	.177	.240	-.099	-.102	-.044	-.028	-.047	.125
.175	.135	.126	.118	.137	.199	-.103	-.119	-.054	-.043	-.052	.175
.225	.126	.106	.102	.113	.170	-.089	-.099	-.066	-.050	-.070	.225
.275	.103	.089	.082	.095	.149	-.087	-.079	-.071	-.087	-.071	.275
.325	.085	.077	.071	.073	.115	-.092		-.078	-.074	-.063	.325
.375		.058	.052	.064	.096	-.097	-.077	-.078	-.074	-.091	.375
.425		.050	.040	.044	.080	-.104	-.092	-.098	-.080	-.082	.425
.475	.044	.040	.033	.032	.057	-.098	-.074	-.087	-.089	-.110	.475
.550	.027	.021	.006	.009	.035	-.105	-.080	-.103	-.113	-.116	.550
.650	.015	.005	-.014	-.005	.002		-.086		-.109	-.121	.650
.750	.003	-.008	-.011	-.033	-.035	-.106	-.096	-.116	-.110	-.123	.750
.800	.000									-.111	.800
.850		-.008	-.021	-.037	-.030	-.091	-.089	-.097	-.105		.850
.900			-.026	-.045			-.077	-.106			.900
.950					-.043	-.078					.950
LOWER SURFACE											
.025		.230	.221	.167	-.007	.125	.137	.105	.098		.025
.075	.192	.190	.168	.120	-.039	.089	.091	.075	.072	.055	.075
.125	.169	.155	.126	.080	-.064	.068	.068	.051	.058	.040	.125
.175	.143	.123	.097	.046	-.077	.052	.050	.033	.037	.033	.175
.225	.116	.099	.073	.023	-.093	.049	.037	.018	.027	.025	.225
.275	.107	.081	.062	.007	-.087	.036	.024	.011	.019	.015	.275
.325	.083	.069	.044	-.010	-.099	.026	.015	.005	.012	.010	.325
.375	.075	.056	.032	-.020	-.098	.018	.007	-.006	.006	-.001	.375
.425	.061	.045		-.030	-.103	.010	-.002	-.010	-.007		.425
.475	.051	.031	.014	-.045	-.100	-.002	-.008	-.020	-.009	-.020	.475
.550	.026	.010	-.011	-.064	-.110	-.004	-.026	-.036	-.019	-.030	.550
.650	.006		-.049	-.085		-.025	-.021	-.051	-.036	-.038	.650
.750	-.010	-.027	-.058	-.101	-.119	-.030	-.046	-.038	-.043	-.047	.750
.800	-.010									-.050	.800
.850		-.030	-.070	-.108	-.094	-.025	-.044	-.040			.850
.900			-.063	-.108			-.039	-.041			.900
.950					-.057	-.029					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 2.5°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.073				.005			.0
22.5				-.011				.043		.014	22.5
45.0				.028	.288	.008	-.042	.049	.043	.055	45.0
67.5				.047	.144	.038	-.042	.029	.006	.030	67.5
90.0				.030				.002	.007	-.004	90.0
112.5				-.009	-.002	-.016	-.049		-.034	-.042	112.5
135.0				-.045	-.067	-.076	-.103	-.086	-.083	-.088	135.0
157.5				-.110	-.123	-.128	-.118	-.122	-.134	-.128	157.5
180.0				-.109	-.110	-.059	-.088	-.125	-.138	-.057	180.0
202.5				-.091	-.052	-.032	-.073	-.124		-.056	202.5
225.0				-.068	-.042	-.049	-.077	-.127	-.072	-.068	225.0
247.5				-.053	-.014	-.035	-.080	-.086	-.083	-.101	247.5
270.0				-.066				-.021	-.022		270.0
292.5				-.091	-.080	-.129	-.104	.014	.000	-.006	292.5
315.0				-.122	-.167	-.099	-.050	.015	.001	-.021	315.0
337.5				-.110				.014	-.014	-.039	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.150	.131	.160	.308	-.249	-.129	-.178	-.189		.025
.075	.106	.112	.089	.126	.238	-.226	-.183	-.149	-.170	-.176	.075
.125	.083	.087	.048	.098	.176	-.208	-.191	-.147	-.149	-.175	.125
.175	.067	.061	.028	.067	.128	-.175	-.194	-.147	-.147	-.173	.175
.225	.061	.039	.020	.044	.095	-.125	-.175	-.154	-.143	-.170	.225
.275	.042	.024	.001	.025	.082	-.122	-.156	-.149	-.170	-.173	.275
.325	.024	.013	-.004	.005	.058	-.123		-.158	-.156	-.162	.325
.375		-.004	-.020	.006	.028	-.125	-.150	-.156	-.154	-.175	.375
.425		-.014	-.032	-.020	.019	-.135	-.161	-.175	-.157	-.167	.425
.475	-.016	-.021	-.032	-.028	.005	-.129	-.144	-.165	-.163	-.184	.475
.550	-.029	-.037	-.051	-.045	-.012	-.134	-.142	-.181	-.181	-.181	.550
.650	-.034	-.056	-.063	-.058	-.046		-.132		-.169	-.180	.650
.750	-.040	-.070	-.067	-.084	-.074	-.132	-.138	-.171	-.165	-.170	.750
.800	-.039									-.148	.800
.850		-.059	-.076	-.084	-.071	-.113	-.128	-.148	-.156		.850
.900			-.070	-.090			-.117	-.157			.900
.950					-.078	-.099					.950
LOWER SURFACE											
.025		.349	.332	.282	.075	.175	.211	.194	.183		.025
.075	.304	.297	.263	.216	.035	.123	.154	.149	.143	.129	.075
.125	.272	.249	.212	.166	.004	.095	.125	.120	.120	.110	.125
.175	.241	.209	.180	.131	-.006	.086	.097	.098	.100	.095	.175
.225	.210	.178	.150	.099	-.025	.077	.085	.081	.084	.087	.225
.275	.195	.172	.131	.075	-.020	.073	.069	.070	.073	.072	.275
.325	.166	.141	.111	.056	-.030	.058	.067	.058	.065	.066	.325
.375	.158	.130	.097	.043	-.033	.051	.051	.054	.057	.058	.375
.425	.137	.118		.035	-.029	.046	.042	.044	.047		.425
.475	.135	.099	.072	.014	-.039	.035	.042	.029	.038	.035	.475
.550	.105	.080	.057	-.005	-.049	.030	.015	.018	.027	.024	.550
.650	.081		.018	-.033		.012	.018	-.004	.002	.011	.650
.750	.056	.036	.002	-.043	-.070	.006	.000	.002	-.004	-.002	.750
.800	.044									-.004	.800
.850		.030	-.011	-.042	-.050	.008	-.006	-.002	-.011		.850
.900			-.007	-.054			.001	-.005			.900
.950					-.014	.006					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 5.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.035				.020			.0
22.5				.033				.070		.026	22.5
45.0				.062	.229	-.032	-.066	.075	.060	.070	45.0
67.5				.072	.159	-.012	-.080	.035	.007	.059	67.5
90.0				.029				-.008	-.007	-.018	90.0
112.5				-.027	-.028	-.061	-.094		-.046	-.061	112.5
135.0				-.067	-.100	-.121	-.143	-.081	-.107	-.106	135.0
157.5				-.141	-.145	-.133	-.097	-.128	-.152	-.025	157.5
180.0				-.095	-.075	-.038	-.088	-.123	-.108	-.015	180.0
202.5				-.066	-.050	-.036	-.088	-.140		-.025	202.5
225.0				-.049	-.020	-.045	-.117	-.148	-.101	-.053	225.0
247.5				-.048	-.005	-.064	-.091	-.069	-.075	-.080	247.5
270.0				-.074				-.014	-.015		270.0
292.5				-.118	-.110	-.147	-.115	.026	-.002	.007	292.5
315.0				-.127	-.250	-.124	-.056	.009	-.005	-.035	315.0
337.5				-.088				.004	-.027	-.052	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.072	.054	.065	.221	-.294	-.165	-.258	-.260		.025
.075	.045	.050	.021	.039	.178	-.273	-.243	-.226	-.237	-.239	.075
.125	.032	.030	-.012	.018	.115	-.266	-.241	-.221	-.221	-.235	.125
.175	.015	.009	-.028	-.007	.072	-.258	-.237	-.211	-.219	-.226	.175
.225	.011	-.007	-.050	-.019	.047	-.171	-.230	-.215	-.211	-.227	.225
.275	-.005	-.021	-.059	-.032	.027	-.152	-.215	-.216	-.223	-.226	.275
.325	-.024	-.033	-.066	-.056	.007	-.156		-.217	-.216	-.214	.325
.375		-.048	-.082	-.057	-.019	-.156	-.207	-.216	-.214	-.220	.375
.425		-.055	-.089	-.073	-.025	-.162	-.217	-.227	-.214	-.221	.425
.475	-.057	-.064	-.089	-.085	-.039	-.155	-.204	-.220	-.214	-.223	.475
.550	-.070	-.079	-.108	-.095	-.056	-.156	-.204	-.227	-.222	-.219	.550
.650	-.067	-.092	-.118	-.105	-.084		-.203		-.204	-.210	.650
.750	-.066	-.107	-.111	-.126	-.106	-.145	-.197	-.220	-.202	-.206	.750
.800	-.057									-.195	.800
.850		-.086	-.109	-.125	-.106	-.129	-.175	-.200	-.197		.850
.900			-.098	-.123			-.152	-.197			.900
.950					-.106	-.108					.950
LOWER SURFACE											
.025		.473	.452	.381	.147	.182	.263	.256	.254		.025
.075	.409	.397	.354	.299	.100	.145	.204	.207	.209	.200	.075
.125	.367	.335	.294	.238	.066	.122	.170	.176	.184	.180	.125
.175	.329	.290	.253	.194	.054	.111	.142	.150	.157	.159	.175
.225	.298	.259	.225	.163	.033	.106	.124	.135	.143	.152	.225
.275	.279	.244	.201	.135	.031	.099	.114	.124	.126	.135	.275
.325	.249	.216	.173	.117	.024	.088	.110	.106	.116	.126	.325
.375	.236	.199	.158	.099	.023	.083	.097	.102	.108	.116	.375
.425	.211	.181	.153	.092	.021	.075	.083	.091	.096		.425
.475	.205	.164	.130	.070	.008	.066	.082	.077	.087	.092	.475
.550	.174	.138	.107	.044	.000	.060	.054	.062	.077	.078	.550
.650	.144		.086	.017		.044	.054	.037	.054	.061	.650
.750	.118	.089	.051	.008	-.026	.043	.036	.042	.047	.053	.750
.800	.099									.050	.800
.850		.083	.036	.005	.004	.045	.032	.041	.036		.850
.900			.043	-.001			.043	.038			.900
.950					.030	.043					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.011				.040			.0
22.5				.067				.091		.042	22.5
45.0				.091	.176	-.069	-.082	.123	.075	.075	45.0
67.5				.068	.161	-.045	-.107	.040	.030	.075	67.5
90.0				.018				-.032	-.014	-.022	90.0
112.5				-.057	-.063	-.115	-.131		-.073	-.084	112.5
135.0				-.106	-.145	-.179	-.115	-.095	-.131	-.015	135.0
157.5				-.169	-.144	-.094	-.103	-.129	-.164	.009	157.5
180.0				-.088	-.064	-.060	-.118	-.142	-.095	.020	180.0
202.5				-.067	-.067	-.100	-.152	-.170		.036	202.5
225.0				-.041	-.005	-.011	-.103	-.163	-.141	.000	225.0
247.5				-.066	-.050	-.110	-.117	-.038	-.068	.007	247.5
270.0				-.102				-.014	-.009		270.0
292.5				-.165	-.157	-.163	-.138	.004	-.018	.027	292.5
315.0				-.130	-.272	-.136	-.086	-.013	-.032	-.048	315.0
337.5				-.069				-.008	-.039	-.053	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.013	-.030	-.015	.143	-.308	-.193	-.289	-.292		.025
.075	-.025	-.021	-.038	-.028	.124	-.301	-.281	-.275	-.271	-.253	.075
.125	-.032	-.027	-.065	-.052	.061	-.301	-.274	-.272	-.271	-.252	.125
.175	-.038	-.045	-.074	-.072	.022	-.288	-.263	-.265	-.265	-.252	.175
.225	-.043	-.056	-.090	-.085	.004	-.258	-.268	-.262	-.261	-.252	.225
.275	-.056	-.067	-.104	-.092	-.012	-.200	-.260	-.258	-.252	-.252	.275
.325	-.069	-.076	-.111	-.110	-.030	-.200		-.263	-.261	-.252	.325
.375		-.091	-.126	-.105	-.054	-.200	-.259	-.259	-.256	-.241	.375
.425		-.098	-.135	-.122	-.061	-.200	-.255	-.254	-.249	-.246	.425
.475	-.100	-.104	-.135	-.128	-.076	-.183	-.253	-.261	-.243	-.239	.475
.550	-.108	-.117	-.161	-.136	-.091	-.178	-.250	-.256	-.237	-.239	.550
.650	-.102	-.130	-.165	-.144	-.116		-.247		-.235	-.239	.650
.750	-.089	-.139	-.151	-.164	-.133	-.155	-.247	-.252	-.240	-.240	.750
.800	-.084									-.236	.800
.850		-.109	-.134	-.156	-.131	-.126	-.240	-.239	-.240		.850
.900			-.128	-.145			-.214	-.234			.900
.950					-.129	-.109					.950
LOWER SURFACE											
.025		.594	.569	.479	.224	.182	.315	.313	.310		.025
.075	.518	.492	.453	.380	.168	.152	.251	.262	.266	.258	.075
.125	.463	.428	.381	.305	.130	.135	.219	.226	.242	.234	.125
.175	.425	.378	.332	.267	.112	.136	.192	.200	.215	.219	.175
.225	.387	.345	.301	.229	.095	.130	.173	.179	.200	.208	.225
.275	.369	.323	.274	.198	.094	.124	.157	.169	.181	.190	.275
.325	.337	.297	.247	.178	.081	.116	.151	.157	.168	.180	.325
.375	.320	.278	.225	.157	.079	.112	.133	.142	.159	.169	.375
.425	.293	.257	.217	.150	.075	.105	.126	.135	.146		.425
.475	.280	.237	.193	.129	.062	.098	.120	.120	.137	.143	.475
.550	.249	.198	.167	.101	.048	.095	.092	.104	.123	.127	.550
.650	.214		.119	.067		.080	.098	.075	.094	.105	.650
.750	.181	.155	.102	.058	.023	.077	.076	.084	.087	.098	.750
.800	.163									.092	.800
.850		.143	.087	.060	.058	.080	.073	.083	.080		.850
.900			.092	.055			.080	.081			.900
.950					.074	.079					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 10.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.026				.055			.0
22.5				.099				.123		.050	22.5
45.0				.108	.136	-.100	-.101	.138	.089	.075	45.0
67.5				.073	.157	-.077	-.136	.038	.049	.056	67.5
90.0				.000				-.047	-.032	-.036	90.0
112.5				-.091	-.114	-.150	-.149		-.087	-.114	112.5
135.0				-.142	-.193	-.155	-.102	-.094	-.155	-.007	135.0
157.5				-.185	-.122	-.120	-.135	-.137	-.188	-.006	157.5
180.0				-.082	-.113	-.114	-.158	-.178	-.097	-.009	180.0
202.5				-.077	-.069	-.114	-.145	-.217		.023	202.5
225.0				-.049	-.005	-.033	-.117	-.122	-.138	-.022	225.0
247.5				-.103	-.099	-.185	-.159	-.045	-.048	.025	247.5
270.0				-.124				-.009	-.016		270.0
292.5				-.197	-.206	-.206	-.197	-.012	-.033	.026	292.5
315.0				-.142	-.278	-.167	-.096	-.050	-.055	-.076	315.0
337.5				-.046				-.001	-.056	-.055	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.091	-.097	-.074	.073	-.306	-.198	-.295	-.287		.025
.075	-.097	-.084	-.091	-.084	.079	-.299	-.280	-.287	-.271	-.247	.075
.125	-.092	-.085	-.111	-.100	.022	-.304	-.275	-.286	-.281	-.248	.125
.175	-.097	-.092	-.117	-.115	-.009	-.287	-.261	-.280	-.279	-.247	.175
.225	-.094	-.106	-.124	-.132	-.033	-.287	-.276	-.280	-.278	-.247	.225
.275	-.103	-.110	-.139	-.141	-.050	-.253	-.285	-.275	-.253	-.248	.275
.325	-.115	-.119	-.145	-.151	-.064	-.252		-.279	-.274	-.261	.325
.375		-.129	-.155	-.148	-.087	-.240	-.280	-.276	-.273	-.247	.375
.425		-.135	-.159	-.158	-.092	-.226	-.269	-.263	-.271	-.263	.425
.475	-.136	-.139	-.164	-.159	-.108	-.214	-.280	-.275	-.266	-.249	.475
.550	-.141	-.152	-.177	-.169	-.117	-.189	-.280	-.269	-.254	-.247	.550
.650	-.133	-.161	-.190	-.172	-.137		-.280		-.266	-.255	.650
.750	-.117	-.166	-.169	-.184	-.154	-.141	-.266	-.269	-.266	-.252	.750
.800	-.111									-.263	.800
.850		-.139	-.154	-.175	-.154	-.128	-.259	-.262	-.263		.850
.900			-.152	-.170			-.265	-.242			.900
.950					-.141	-.118					.950
LOWER SURFACE											
.025		.690	.661	.557	.293	.191	.348	.354	.354		.025
.075	.605	.578	.531	.446	.229	.169	.287	.304	.314	.300	.075
.125	.549	.508	.453	.372	.189	.155	.260	.269	.292	.282	.125
.175	.504	.455	.409	.327	.172	.154	.231	.245	.263	.265	.175
.225	.462	.419	.363	.287	.150	.153	.210	.222	.245	.254	.225
.275	.445	.394	.338	.257	.148	.150	.199	.209	.227	.238	.275
.325	.412	.362	.311	.234	.135	.142	.187	.195	.214	.224	.325
.375	.390	.336	.291	.217	.120	.138	.172	.184	.205	.208	.375
.425	.360	.317	.273	.203	.118	.132	.158	.175	.193		.425
.475	.343	.297	.255	.182	.113	.126	.154	.156	.183	.189	.475
.550	.311	.262	.222	.157	.095	.125	.126	.138	.166	.169	.550
.650	.272		.169	.116		.112	.132	.116	.133	.153	.650
.750	.236	.205	.149	.105	.085	.111	.113	.123	.130	.143	.750
.800	.222									.139	.800
.850		.195	.138	.102	.102	.114	.111	.125	.124		.850
.900			.142	.098			.117	.123			.900
.950					.116	.111					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 12.5°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.063				.104			.0
22.5				.132				.158		.071	22.5
45.0				.128	.095	-.125	-.111	.163	.089	.081	45.0
67.5				.076	.146	-.108	-.158	.043	.040	.040	67.5
90.0				-.015				-.047	-.053	-.066	90.0
112.5				-.116	-.146	-.170	-.123		-.124	-.109	112.5
135.0				-.156	-.212	-.132	-.108	-.097	-.176	.005	135.0
157.5				-.173	-.134	-.115	-.152	-.141	-.157	-.032	157.5
180.0				-.076	-.177	-.168	-.169	-.171	-.082	-.005	180.0
202.5				-.076	-.057	-.073	-.141	-.200		.045	202.5
225.0				-.071	-.039	-.066	-.132	-.093	-.123	.030	225.0
247.5				-.119	-.186	-.248	-.201	-.089	-.053	.063	247.5
270.0				-.168				-.035	-.056		270.0
292.5				-.200	-.213	-.219	-.225	-.127	-.087	.053	292.5
315.0				-.125	-.276	-.184	-.114	-.068	-.130	-.071	315.0
337.5				-.032				.037	-.047	-.078	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.155	-.161	-.139	.014	-.293	-.192	-.283	-.269		.025
.075	-.153	-.140	-.140	-.135	.042	-.282	-.263	-.281	-.250	-.231	.075
.125	-.151	-.133	-.140	-.143	-.002	-.292	-.262	-.282	-.269	-.229	.125
.175	-.147	-.138	-.150	-.157	-.040	-.268	-.248	-.275	-.270	-.229	.175
.225	-.146	-.146	-.161	-.170	-.066	-.284	-.267	-.279	-.273	-.227	.225
.275	-.151	-.149	-.165	-.178	-.080	-.255	-.288	-.275	-.241	-.231	.275
.325	-.149	-.152	-.171	-.177	-.097	-.266		-.282	-.271	-.248	.325
.375		-.155	-.175	-.185	-.116	-.257	-.280	-.282	-.270	-.230	.375
.425		-.162	-.182	-.185	-.122	-.237	-.260	-.257	-.270	-.248	.425
.475	-.160	-.165	-.189	-.185	-.133	-.234	-.284	-.280	-.270	-.231	.475
.550	-.158	-.165	-.194	-.203	-.149	-.207	-.286	-.264	-.257	-.230	.550
.650	-.152	-.182	-.203	-.203	-.164		-.283		-.274	-.241	.650
.750	-.144	-.171	-.175	-.196	-.173	-.163	-.262	-.258	-.267	-.238	.750
.800	-.141									-.251	.800
.850		-.160	-.166	-.195	-.164	-.176	-.253	-.270	-.263		.850
.900			-.168	-.191			-.264	-.243			.900
.950					-.149	-.181					.950
LOWER SURFACE											
.025		.762	.742	.633	.369	.230	.373	.388	.392		.025
.075	.685	.652	.608	.517	.300	.207	.330	.350	.363	.349	.075
.125	.623	.582	.527	.440	.250	.190	.296	.318	.343	.331	.125
.175	.579	.532	.477	.389	.230	.194	.270	.293	.316	.313	.175
.225	.529	.492	.436	.354	.210	.196	.251	.273	.295	.304	.225
.275	.514	.461	.405	.319	.205	.194	.242	.258	.281	.287	.275
.325	.483	.429	.373	.295	.185	.181	.225	.249	.267	.276	.325
.375	.455	.404	.358	.283	.180	.180	.215	.235	.256	.264	.375
.425	.424	.385	.337	.265	.174	.175	.206	.220	.240		.425
.475	.406	.361	.315	.248	.165	.169	.195	.206	.229	.233	.475
.550	.371	.315	.280	.214	.146	.170	.176	.183	.208	.216	.550
.650	.332		.223	.174		.155	.180	.162	.184	.202	.650
.750	.295	.270	.207	.155	.144	.152	.156	.173	.180	.198	.750
.800	.280									.194	.800
.850		.255	.199	.152	.161	.157	.152	.173	.176		.850
.900			.195	.149			.157	.172			.900
.950					.168	.154					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 15.0°$ $\beta = -10°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.096				.125			.0
22.5				.165				.184		.093	22.5
45.0				.151	.060	-.129	-.130	.179	.110	.084	45.0
67.5				.079	.120	-.124	-.165	.056	.035	.025	67.5
90.0				-.027				-.030	-.050	-.093	90.0
112.5				-.111	-.149	-.156	-.138		-.136	-.176	112.5
135.0				-.147	-.189	-.145	-.131	-.096	-.165	-.109	135.0
157.5				-.151	-.142	-.131	-.168	-.144	-.176	-.131	157.5
180.0				-.066	-.206	-.151	-.183	-.163	-.107	-.136	180.0
202.5				-.093	-.054	-.111	-.140	-.195		-.072	202.5
225.0				-.115	-.111	-.133	-.162	-.124	-.082	-.084	225.0
247.5				-.141	-.192	-.250	-.254	-.084	-.056	-.108	247.5
270.0				-.157				-.062	-.068		270.0
292.5				-.193	-.199	-.211	-.217	-.130	-.084	.021	292.5
315.0				-.114	-.258	-.191	-.147	-.019	-.110	-.120	315.0
337.5				-.011				.042	-.014	-.043	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.182	-.191	-.170	-.031	-.275	-.181	-.267	-.258		.025
.075	-.182	-.167	-.171	-.168	.007	-.266	-.246	-.263	-.237	-.215	.075
.125	-.179	-.162	-.158	-.168	-.025	-.272	-.245	-.263	-.254	-.211	.125
.175	-.171	-.167	-.175	-.180	-.043	-.250	-.226	-.261	-.254	-.213	.175
.225	-.171	-.167	-.184	-.189	-.082	-.268	-.249	-.261	-.259	-.213	.225
.275	-.173	-.171	-.176	-.195	-.098	-.245	-.271	-.261	-.217	-.213	.275
.325	-.154	-.172	-.200	-.178	-.112	-.258		-.266	-.256	-.240	.325
.375		-.160	-.181	-.201	-.121	-.253	-.266	-.267	-.258	-.215	.375
.425		-.169	-.193	-.188	-.138	-.223	-.242	-.237	-.259	-.241	.425
.475	-.161	-.172	-.208	-.195	-.150	-.240	-.269	-.263	-.256	-.214	.475
.550	-.156	-.172	-.196	-.215	-.161	-.219	-.269	-.245	-.242	-.221	.550
.650	-.161	-.189	-.202	-.210	-.176		-.269		-.261	-.228	.650
.750	-.155	-.169	-.179	-.198	-.181	-.185	-.240	-.243	-.259	-.224	.750
.800	-.157									-.245	.800
.850		-.174	-.165	-.211	-.175	-.200	-.233	-.266	-.256		.850
.900			-.167	-.196			-.249	-.237			.900
.950					-.161	-.194					.950
LOWER SURFACE											
.025		.833	.809	.710	.445	.272	.402	.412	.423		.025
.075	.753	.728	.672	.592	.367	.254	.367	.396	.405	.385	.075
.125	.688	.657	.596	.516	.318	.243	.343	.358	.390	.370	.125
.175	.642	.604	.543	.449	.290	.230	.316	.338	.363	.355	.175
.225	.591	.561	.502	.418	.274	.242	.296	.319	.342	.350	.225
.275	.571	.521	.464	.388	.268	.235	.284	.302	.326	.330	.275
.325	.545	.497	.431	.368	.253	.231	.267	.293	.312	.318	.325
.375	.516	.461	.417	.341	.244	.224	.262	.278	.298	.305	.375
.425	.490	.448	.386	.325	.225	.225	.243	.265	.282		.425
.475	.454	.425	.377	.307	.224	.218	.238	.250	.269	.276	.475
.550	.427	.374	.332	.278	.209	.212	.215	.224	.246	.257	.550
.650	.383		.276	.228		.200	.224	.209	.231	.254	.650
.750	.346	.322	.255	.200	.206	.197	.191	.217	.224	.248	.750
.800	.338									.243	.800
.850		.311	.256	.200	.219	.199	.197	.216	.220		.850
.900			.247	.213			.194	.211			.900
.950					.226	.194					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.159				-.043			.0
22.5				-.070				-.011	-.037	-.061	22.5
45.0				.011	.518	.126	.036	-.014	-.021	-.013	45.0
67.5				.084	.236	.159	.051	.005	.007	.030	67.5
90.0				.104				.005	.026	.033	90.0
112.5				.068	.069	.077	.020		.005	.000	112.5
135.0				.020	-.022	-.014	-.065	-.093	-.093	-.085	135.0
157.5				-.089	-.121	-.125	-.168	-.183	-.170	-.011	157.5
180.0				-.121	-.191	-.177	-.156	-.198	-.135	.035	180.0
202.5				-.205	-.145	-.082	-.151	-.161		.033	202.5
225.0				-.110	-.130	-.140	-.159	-.156	-.112	.028	225.0
247.5				-.091	-.119	-.160	-.196	-.155	-.139	-.022	247.5
270.0				-.068				-.065	-.102		270.0
292.5				-.098	-.077	-.085	-.082	-.036	-.051	-.065	292.5
315.0				-.118	-.082	-.065	-.061	-.016	-.030	-.040	315.0
337.5				-.208				-.014	-.040	-.063	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.315	.306	.399	.566	-.167	-.116	-.115	-.106		.025
.075	.147	.259	.253	.320	.438	-.158	-.123	-.089	-.082	-.080	.075
.125	.133	.221	.219	.273	.371	-.164	-.112	-.087	-.073	-.076	.125
.175	.128	.188	.189	.219	.320	-.151	-.115	-.084	-.073	-.077	.175
.225	.128	.164	.170	.191	.288	-.135	-.097	-.082	-.079	-.092	.225
.275	.120	.148	.154	.170	.260	-.108	-.084	-.082	-.093	-.090	.275
.325	.111	.132	.135	.148	.210	-.100	-.082	-.085	-.085	-.085	.325
.375	.102	.122	.122	.137	.196	-.084	-.078	-.083	-.085	-.097	.375
.425		.108	.104	.113	.170	-.074	-.074	-.082	-.091	-.102	.425
.475	.081	.095	.093	.099	.144	-.060	-.072	-.080	-.093	-.112	.475
.550	.062	.083	.072	.069	.122	-.054	-.080	-.086	-.106	-.117	.550
.650	.051	.055	.039	.054	.074		-.086	-.093	-.103	-.126	.650
.750	.041	.043	.040	.021		-.079	-.084	-.096	-.103	-.119	.750
.800	.036									-.102	.800
.850		.036	.030	.013	.038	-.063	-.082	-.087	-.098		.850
.900			.029	.012			-.073	-.086			.900
.950					.026	-.064					.950
LOWER SURFACE											
.025		.280	.265	.218	-.015	.096	.134	.088	.078		.025
.075	.166	.231	.204	.154	-.043	.075	.091	.058	.054	.034	.075
.125	.154	.192	.160	.109	-.064	.056	.064	.038	.041	.023	.125
.175	.140	.159		.074	-.079	.044	.042	.024	.029	.013	.175
.225	.115	.134	.107	.051	-.088	.024	.028	.011	.020	.010	.225
.275	.109	.116	.090	.032	-.088	.014	.019	.002	.011	.001	.275
.325	.098	.097	.068	.017	-.100	.008	.012	-.004	.004	.000	.325
.375	.087	.087	.055	.005	-.106	.004	.005	-.008	.001	-.008	.375
.425	.074	.076	.047	-.011	-.109	-.005	-.006	-.017	-.007		.425
.475	.061	.064	.034	-.024	-.116	-.014	-.008	-.026	-.013	-.026	.475
.550	.044	.038	.008	-.045	-.126	-.020	-.028	-.038	-.025	-.033	.550
.650	.017		-.019	-.070		-.037	-.027	-.053	-.038	-.043	.650
.750	.008	-.002	-.037	-.088	-.146	-.036	-.049	-.032	-.043	-.049	.750
.800	.006									-.050	.800
.850		-.004	-.043	-.090	-.142	-.032	-.045	-.031	-.047		.850
.900			-.043	-.091			-.044	-.032			.900
.950					-.136	-.036					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 2.5°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.111				-.044			.0
22.5				-.020				.026	.022	-.021	22.5
45.0				.057	.452	.089	.005	.027	.016	.026	45.0
67.5				.105	.261	.111	.016	.037	.030	.055	67.5
90.0				.106				.013	.027	.030	90.0
112.5				.047	.047	.033	-.022		-.037	-.014	112.5
135.0				-.017	-.054	-.054	-.107	-.106	-.122	-.105	135.0
157.5				-.140	-.165	-.169	-.182	-.172	-.203	-.050	157.5
180.0				-.196	-.212	-.115	-.126	-.161	-.149	-.023	180.0
202.5				-.155	-.128	-.094	-.154	-.172		-.057	202.5
225.0				-.105	-.139	-.182	-.158	-.167	-.119	-.023	225.0
247.5				-.065	-.051	-.076	-.139	-.165	-.176	-.108	247.5
270.0				-.077				-.063	-.105		270.0
292.5				-.106	-.139	-.125	-.093	-.040	-.027	-.070	292.5
315.0				-.193	-.126	-.099	-.075	-.003	-.034	-.020	315.0
337.5				-.184				-.005	-.061	-.099	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.228	.201	.275	.483	-.228	-.172	-.235	-.241		.025
.075	.056	.176	.155	.228	.378	-.241	-.220	-.204	-.224	-.227	.075
.125	.037	.147	.111	.191	.311	-.210	-.218	-.197	-.199	-.219	.125
.175	.028	.119	.092	.146	.254	-.199	-.224	-.188	-.194	-.217	.175
.225	.028	.097	.078	.119	.213	-.172	-.198	-.183	-.187	-.220	.225
.275	.021	.080	.067	.102	.198	-.131	-.162	-.180	-.213	-.215	.275
.325	.013	.070	.057	.083	.155	-.112	-.147	-.173	-.188	-.195	.325
.375	.006	.052	.042	.071	.132	-.083	-.143	-.166	-.183	-.217	.375
.425		.041	.030	.049	.116	-.070	-.155	-.176	-.181	-.199	.425
.475	.003	.031	.027	.037	.090	-.057	-.133	-.162	-.184	-.217	.475
.550	-.006	.012	.004	.013	.068	-.061	-.133	-.169	-.201	-.213	.550
.650	-.006	-.010	-.015	-.001	.028		-.135	-.162	-.185	-.205	.650
.750	-.009	-.025	-.015	-.030		-.084	-.146	-.153	-.183	-.190	.750
.800	-.010									-.173	.800
.850		-.020	-.022	-.032	-.006	-.069	-.136	-.133	-.170		.850
.900			-.021	-.037			-.122	-.135			.900
.950					-.015	-.069					.950
LOWER SURFACE											
.025		.407	.392	.334	.065	.102	.199	.168	.158		.025
.075	.272	.343	.315	.251	.028	.085	.152	.127	.125	.108	.075
.125	.257	.294	.262	.193	-.005	.068	.121	.107	.107	.093	.125
.175	.234	.257	.227	.159	-.017	.059	.096	.084	.090	.079	.175
.225	.214	.229	.197	.127	-.032	.050	.082	.070	.076	.073	.225
.275	.204	.211	.173	.099	-.032	.043	.071	.064	.067	.062	.275
.325	.180	.185	.149	.078	-.045	.039	.064	.054	.060	.053	.325
.375	.171	.172	.134	.064	-.052	.032	.051	.047	.052	.047	.375
.425	.154	.154	.126	.054	-.047	.027	.046	.040	.042		.425
.475	.147	.142	.109	.037	-.057	.024	.041	.030	.037	.029	.475
.550	.121	.113	.081	.013	-.065	.021	.020	.018	.025	.017	.550
.650	.092		.040	-.021		.008	.024	-.002	.005	.007	.650
.750	.077	.069	.024	-.033	-.091	.006	.002	.011	.003	-.001	.750
.800	.071									-.007	.800
.850		.062	.014	-.038	-.088	.009	.004	.008	-.001		.850
.900			.014	-.046			.006	.008			.900
.950					-.076	.007					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 5.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				-.076				-.019			.0
22.5				.023				.063	.041	.016	22.5
45.0				.092	.390	.040	-.023	.058	.040	.057	45.0
67.5				.131	.266	.068	-.019	.044	.047	.064	67.5
90.0				.092				.007	.014	.019	90.0
112.5				.019	.016	-.006	-.068		-.050	-.028	112.5
135.0				-.055	-.096	-.111	-.146	-.141	-.148	-.019	135.0
157.5				-.182	-.193	-.165	-.126	-.188	-.206	.035	157.5
180.0				-.188	-.167	-.112	-.132	-.162	-.150	.042	180.0
202.5				-.124	-.160	-.132	-.176	-.191		.084	202.5
225.0				-.099	-.111	-.170	-.199	-.164	-.142	.070	225.0
247.5				-.071	-.010	-.078	-.140	-.128	-.200	.062	247.5
270.0				-.107				-.063	-.085		270.0
292.5				-.142	-.168	-.209	-.119	-.015	-.040	.080	292.5
315.0				-.217	-.179	-.128	-.084	.000	-.048	.075	315.0
337.5				-.154				-.036	-.092	.061	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		.149	.133	.163	.407	-.273	-.195	-.252	-.283		.025
.075	.005	.114	.093	.140	.316	-.284	-.255	-.253	-.269	-.266	.075
.125	-.015	.087	.052	.119	.248	-.226	-.257	-.257	-.260	-.259	.125
.175	-.028	.064	.032	.087	.201	-.209	-.252	-.250	-.252	-.258	.175
.225	-.033	.044	.015	.061	.155	-.190	-.245	-.249	-.246	-.257	.225
.275	-.040	.032	.001	.045	.142	-.147	-.228	-.241	-.256	-.253	.275
.325	-.052	.018	-.011	.028	.107	-.122	-.218	-.234	-.248	-.250	.325
.375	-.062	.002	-.025	.020	.082	-.104	-.215	-.226	-.243	-.249	.375
.425		-.008	-.036	.004	.066	-.093	-.223	-.222	-.243	-.248	.425
.475	-.063	-.015	-.038	-.007	.045	-.081	-.205	-.208	-.242	-.244	.475
.550	-.063	-.032	-.052	-.032	.025	-.113	-.210	-.211	-.248	-.239	.550
.650	-.059	-.052	-.068	-.046	-.008		-.205	-.202	-.241	-.236	.650
.750	-.049	-.066	-.058	-.069		-.155	-.202	-.201	-.226	-.235	.750
.800	-.041									-.228	.800
.850		-.044	-.062	-.072	-.038	-.111	-.174	-.180	-.197		.850
.900			-.047	-.076			-.149	-.168			.900
.950					-.045	-.072					.950
LOWER SURFACE											
.025		.535	.513	.435	.135	.107	.234	.214	.211		.025
.075	.399	.452	.413	.335	.090	.086	.185	.174	.171	.160	.075
.125	.368	.390	.347	.268	.059	.078	.156	.150	.158	.140	.125
.175	.332	.343	.314	.225	.039	.063	.130	.128	.136	.129	.175
.225	.303	.306	.269	.192	.025	.064	.116	.113	.120	.120	.225
.275	.288	.288	.239	.166	.023	.057	.105	.104	.109	.106	.275
.325	.261	.263	.220	.136	.015	.054	.095	.089	.098	.101	.325
.375	.246	.242	.197	.121	.011	.051	.083	.083	.091	.090	.375
.425	.226	.227	.185	.108	.009	.047	.075	.077	.081		.425
.475	.213	.211	.163	.088	-.002	.039	.073	.065	.074	.068	.475
.550	.186	.175	.135	.062	-.014	.040	.051	.049	.060	.057	.550
.650	.148		.090	.027		.032	.054	.028	.039	.044	.650
.750	.124	.124	.073	.009	-.043	.030	.036	.043	.039	.037	.750
.800	.123									.035	.800
.850		.115	.062	.009	-.036	.030	.033	.040	.032		.850
.900			.064	.005			.036	.040			.900
.950					-.024	.030					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 7.5°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				-.042				-.013			.0
22.5				.061				.079	.066	.029	22.5
45.0				.125	.326	.008	-.044	.106	.136	.113	45.0
67.5				.134	.268	.031	-.055	.056	.036	.077	67.5
90.0				.082				.001	-.003	.010	90.0
112.5				-.019	-.026	-.064	-.096		-.073	-.050	112.5
135.0				-.085	-.133	-.166	-.181	-.155	-.166	.054	135.0
157.5				-.211	-.211	-.133	-.126	-.167	-.225	.043	157.5
180.0				-.185	-.143	-.098	-.154	-.169	-.143	.044	180.0
202.5				-.135	-.192	-.160	-.164	-.199		.085	202.5
225.0				-.101	-.121	-.111	-.192	-.175	-.168	.073	225.0
247.5				-.084	-.035	-.104	-.162	-.100	-.164	.070	247.5
270.0				-.126				-.083	-.047		270.0
292.5				-.190	-.192	-.239	-.159	-.020	-.055	.066	292.5
315.0				-.212	-.259	-.149	-.124	-.041	-.082	.063	315.0
337.5				-.139				-.080	-.114	.042	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		.070	.055	.076	.338	-.306	-.209	-.278	-.282		.025
.075	-.036	.051	.030	.057	.270	-.306	-.280	-.274	-.277	-.267	.075
.125	-.053	.035	.001	.037	.196	-.230	-.275	-.275	-.284	-.264	.125
.175	-.064	.017	-.020	.020	.158	-.211	-.265	-.275	-.279	-.263	.175
.225	-.065	.002	-.034	.004	.112	-.196	-.273	-.275	-.282	-.265	.225
.275	-.078	-.010	-.048	-.006	.097	-.153	-.265	-.270	-.270	-.261	.275
.325	-.088	-.019	-.062	-.025	.064	-.126	-.264	-.278	-.281	-.268	.325
.375	-.095	-.034	-.074	-.033	.042	-.112	-.259	-.267	-.279	-.254	.375
.425		-.045	-.084	-.043	.034	-.110	-.261	-.259	-.281	-.263	.425
.475	-.105	-.052	-.085	-.053	.011	-.117	-.261	-.259	-.277	-.253	.475
.550	-.098	-.067	-.102	-.069	-.008	-.159	-.258	-.257	-.267	-.254	.550
.650	-.077	-.085	-.121	-.083	-.042		-.249	-.246	-.264	-.259	.650
.750	-.067	-.100	-.105	-.103		-.191	-.226	-.247	-.239	-.256	.750
.800	-.063									-.257	.800
.850		-.072	-.090	-.105	-.068	-.191	-.196	-.229	-.168		.850
.900			-.078	-.102			-.173	-.215			.900
.950					-.072	-.163					.950
LOWER SURFACE											
.025		.663	.634	.544	.220	.069	.252	.254	.254		.025
.075	.527	.553	.508	.428	.167	.081	.212	.218	.226	.211	.075
.125	.477	.477	.431	.353	.132	.081	.190	.194	.207	.193	.125
.175	.436	.426	.374	.301	.113	.078	.165	.173	.187	.181	.175
.225	.399	.392	.340	.267	.098	.063	.152	.159	.169	.173	.225
.275	.379	.364	.313	.233	.098	.062	.142	.148	.157	.157	.275
.325	.348	.341	.282	.212	.085	.064	.132	.137	.147	.148	.325
.375	.333	.319	.263	.192	.079	.066	.121	.126	.141	.138	.375
.425	.310	.297	.245	.173	.071	.062	.113	.116	.128		.425
.475	.293	.274	.227	.156	.063	.066	.108	.103	.121	.119	.475
.550	.252	.240	.188	.124	.051	.066	.084	.088	.108	.101	.550
.650	.211		.152	.089		.063	.092	.073	.085	.090	.650
.750	.184	.185	.133	.071	.013	.056	.070	.085	.084	.084	.750
.800	.172									.084	.800
.850		.174	.122	.070	.023	.062	.071	.086	.078		.850
.900			.122	.070			.071	.080			.900
.950					.039	.063					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 10.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
.0				.002				.031			.0
22.5				.110				.134	.086	.056	22.5
45.0				.156	.289	-.026	-.049	.192	.152	.128	45.0
67.5				.146	.273	-.007	-.079	.065	.080	.129	67.5
90.0				.072				-.010	-.016	.012	90.0
112.5				-.035	-.056	-.097	-.132		-.094	.009	112.5
135.0				-.124	-.167	-.190	-.167	-.145	-.179	.082	135.0
157.5				-.218	-.176	-.126	-.133	-.153	-.204	.058	157.5
180.0				-.163	-.142	-.114	-.153	-.171	-.133	.047	180.0
202.5				-.131	-.219	-.164	-.179	-.183		.075	202.5
225.0				-.086	-.056	-.103	-.160	-.154	-.174	.069	225.0
247.5				-.125	-.124	-.150	-.176	-.093	-.112	.065	247.5
270.0				-.184				-.075	-.059		270.0
292.5				-.203	-.200	-.244	-.202	-.029	-.073	.063	292.5
315.0				-.203	-.283	-.161	-.162	-.089	-.117	.054	315.0
337.5				-.098				-.056	-.117	.056	337.5

x/c	C_p AT WING STATION 1	2	3	4	5	6	7	8	9	10	x/c
UPPER SURFACE											
.025		-.016	-.022	.005	.265	-.312	-.223	-.290	-.292		.025
.075	-.089	-.015	-.025	-.007	.223	-.310	-.289	-.288	-.277	-.255	.075
.125	-.100	-.022	-.047	-.023	.153	-.300	-.286	-.289	-.288	-.252	.125
.175	-.107	-.036	-.059	-.043	.109	-.263	-.275	-.289	-.284	-.249	.175
.225	-.107	-.048	-.072	-.062	.077	-.249	-.289	-.292	-.289	-.250	.225
.275	-.114	-.057	-.090	-.072	.059	-.211	-.299	-.284	-.268	-.249	.275
.325	-.125	-.064	-.103	-.081	.034	-.180	-.297	-.290	-.285	-.255	.325
.375	-.135	-.082	-.112	-.077	.006	-.141	-.293	-.288	-.284	-.246	.375
.425		-.087	-.119	-.092	-.002	-.136	-.283	-.272	-.286	-.257	.425
.475	-.126	-.092	-.119	-.093	-.021	-.140	-.297	-.285	-.285	-.245	.475
.550	-.105	-.106	-.136	-.108	-.042	-.177	-.295	-.279	-.270	-.250	.550
.650	-.093	-.123	-.153	-.115	-.072		-.282	-.275	-.273	-.257	.650
.750	-.094	-.134	-.138	-.134		-.223	-.262	-.272	-.262	-.256	.750
.800	-.090									-.262	.800
.850		-.107	-.116	-.128	-.094	-.234	-.243	-.271	-.249		.850
.900			-.115	-.122			-.242	-.255			.900
.950					-.099	-.207					.950
LOWER SURFACE											
.025		.779	.744	.638	.300	.103	.274	.282	.286		.025
.075	.650	.657	.599	.506	.243	.115	.244	.256	.267	.250	.075
.125	.590	.578	.516	.430	.201	.115	.218	.233	.255	.231	.125
.175	.544	.521		.371	.180	.114	.197	.214	.231	.220	.175
.225	.499	.478	.428	.332	.167	.102	.184	.198	.215	.217	.225
.275	.478	.452	.393	.300	.160	.101	.175	.188	.204	.201	.275
.325	.450	.420	.362	.277	.147	.098	.165	.175	.191	.193	.325
.375	.424	.398	.339	.252	.139	.099	.155	.168	.182	.184	.375
.425	.400	.375	.321	.233	.130	.095	.146	.155	.169		.425
.475	.377	.349	.298	.216	.120	.095	.139	.141	.162	.158	.475
.550	.329	.308	.262	.187	.113	.096	.117	.126	.142	.143	.550
.650	.283		.208	.146		.086	.127	.114	.130	.136	.650
.750	.245	.251	.186	.126	.066	.086	.107	.124	.126	.134	.750
.800	.234									.131	.800
.850		.240	.179	.129	.081	.092	.105	.124	.119		.850
.900			.178	.123			.109	.123			.900
.950					.103	.092					.950

TABLE 4, Continued

PRESSURE COEFFICIENT DATA FOR WING-BODY COMBINATION

LOW-WING CONFIGURATION

$\alpha = 12.5°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.043				.087			.0
22.5				.157				.182	.123	.086	22.5
45.0				.195	.252	-.054	-.057	.221	.155	.135	45.0
67.5				.162	.248	-.027	-.103	.068	.113	.125	67.5
90.0				.063				-.031	-.030	-.001	90.0
112.5				-.052	-.082	-.123	-.149		-.107	-.086	112.5
135.0				-.143	-.195	-.176	-.148	-.143	-.192	.023	135.0
157.5				-.207	-.160	-.161	-.141	-.133	-.182	.007	157.5
180.0				-.171	-.160	-.149	-.169	-.175	-.120	-.047	180.0
202.5				-.171	-.210	-.184	-.188	-.164		.027	202.5
225.0				-.084	-.055	-.118	-.158	-.133	-.170	.020	225.0
247.5				-.170	-.210	-.210	-.204	-.085	-.070	.019	247.5
270.0				-.192				-.062	-.023		270.0
292.5				-.213	-.206	-.239	-.236	-.105	-.024	.016	292.5
315.0				-.178	-.292	-.195	-.188	-.127	-.056	.012	315.0
337.5				-.073				-.038	-.100	-.040	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.088	-.091	-.061	.202	-.309	-.225	-.294	-.289		.025
.075	-.136	-.070	-.081	-.060	.183	-.308	-.283	-.291	-.274	-.254	.075
.125	-.140	-.074	-.091	-.077	.124	-.306	-.278	-.291	-.287	-.251	.125
.175	-.141	-.079	-.101	-.093	.079	-.269	-.267	-.288	-.284	-.252	.175
.225	-.141	-.089	-.111	-.109	.040	-.267	-.282	-.290	-.284	-.249	.225
.275	-.148	-.094	-.120	-.116	.022	-.236	-.299	-.284	-.258	-.249	.275
.325	-.147	-.100	-.133	-.122	.002	-.227	-.299	-.293	-.286	-.263	.325
.375	-.149	-.110	-.138	-.122	-.025	-.204	-.294	-.291	-.283	-.247	.375
.425		-.116	-.144	-.126	-.031	-.173	-.277	-.271	-.287	-.260	.425
.475	-.133	-.124	-.152	-.127	-.053	-.157	-.299	-.289	-.283	-.244	.475
.550	-.119	-.130	-.163	-.139	-.071	-.204	-.301	-.276	-.268	-.247	.550
.650	-.116	-.149	-.181	-.153	-.093		-.284	-.275	-.281	-.254	.650
.750	-.118	-.152	-.140	-.158		-.223	-.262	-.270	-.274	-.258	.750
.800	-.118									-.268	.800
.850		-.130	-.128	-.162	-.113	-.240	-.251	-.277	-.265		.850
.900			-.130	-.156			-.262	-.261			.900
.950					-.107	-.254					.950
LOWER SURFACE											
.025		.868	.835	.717	.376	.141	.273	.308	.317		.025
.075	.748	.738	.684	.578	.308	.154	.262	.294	.306	.286	.075
.125	.681	.658	.593	.491	.264	.156	.244	.275	.299	.279	.125
.175	.631	.605		.438	.239	.154	.223	.259	.277	.264	.175
.225	.584	.554	.496	.394	.223	.155	.213	.246	.263	.265	.225
.275	.565	.516	.459	.362	.214	.152	.206	.230	.248	.245	.275
.325	.527	.484	.427	.334	.206	.152	.197	.221	.238	.239	.325
.375	.496	.461	.399	.315	.194	.149	.187	.213	.227	.230	.375
.425	.465	.434	.380	.291	.182	.143	.180	.198	.217		.425
.475	.442	.413	.356	.273	.176	.141	.178	.185	.209	.205	.475
.550	.399	.363	.318	.249	.157	.140	.154	.167	.191	.192	.550
.650	.341		.262	.199		.135	.166	.155	.177	.187	.650
.750	.301	.311	.240	.182	.120	.135	.144	.172	.173	.184	.750
.800	.288									.183	.800
.850		.292	.234	.176	.144	.137	.149	.172	.165		.850
.900			.231	.174			.149	.167			.900
.950					.167	.137					.950

TABLE 4, Concluded

PRESSURE COEFFICIENT DATA FOR WING–BODY COMBINATION

LOW–WING CONFIGURATION

$\alpha = 15.0°$ $\beta = -15°$

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
.0				.091				.104			.0
22.5				.199				.217	.149	.121	22.5
45.0				.230	.216	-.072	-.064	.261	.174	.159	45.0
67.5				.168	.216	-.056	-.120	.082	.107	.118	67.5
90.0				.054				-.029	-.027	-.014	90.0
112.5				-.069	-.090	-.142	-.173		-.128	-.106	112.5
135.0				-.153	-.212	-.177	-.148	-.160	-.203	.017	135.0
157.5				-.192	-.166	-.174	-.160	-.136	-.184	.006	157.5
180.0				-.154	-.177	-.162	-.181	-.171	-.125	-.043	180.0
202.5				-.188	-.192	-.181	-.206	-.177		.019	202.5
225.0				-.114	-.111	-.132	-.170	-.112	-.162	.047	225.0
247.5				-.183	-.199	-.237	-.245	-.063	-.068	.037	247.5
270.0				-.183				-.189	-.155		270.0
292.5				-.211	-.205	-.239	-.249	-.198	-.134	.021	292.5
315.0				-.169	-.276	-.204	-.196	-.128	-.180	.012	315.0
337.5				-.062				-.013	-.077	-.078	337.5

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
UPPER SURFACE											
.025		-.132	-.132	-.109	.135	-.276	-.211	-.264	-.250		.025
.075	-.164	-.117	-.119	-.105	.149	-.267	-.243	-.260	-.224	-.210	.075
.125	-.163	-.116	-.103	-.110	.096	-.270	-.236	-.254	-.244	-.206	.125
.175	-.162	-.118	-.126	-.123	.060	-.244	-.218	-.250	-.242	-.209	.175
.225	-.163	-.121	-.138	-.136	.017	-.258	-.241	-.254	-.250	-.206	.225
.275	-.168	-.122	-.127	-.138	-.006	-.230	-.267	-.250	-.205	-.208	.275
.325	-.141	-.124	-.137	-.119	-.026	-.236	-.262	-.257	-.248	-.234	.325
.375	-.123	-.118	-.132	-.145	-.040	-.217	-.258	-.258	-.245	-.210	.375
.425		-.126	-.142	-.131	-.052	-.182	-.228	-.222	-.252	-.234	.425
.475	-.133	-.132	-.168	-.138	-.071	-.195	-.261	-.252	-.248	-.205	.475
.550	-.124	-.129	-.153	-.164	-.090	-.201	-.265	-.231	-.225	-.209	.550
.650	-.135	-.163	-.181	-.169	-.109		-.255	-.227	-.250	-.217	.650
.750	-.138	-.140	-.148	-.157		-.227	-.222	-.224	-.243	-.218	.750
.800	-.135									-.238	.800
.850		-.140	-.135	-.178	-.124	-.222	-.212	-.250	-.243		.850
.900			-.135	-.155			-.238	-.224			.900
.950					-.103	-.214					.950
LOWER SURFACE											
.025		.942	.916	.808	.455	.185	.297	.322	.334		.025
.075	.833	.814	.758	.662	.390	.204	.299	.323	.336	.313	.075
.125	.761	.718	.668	.575	.344	.205	.282	.309	.331	.309	.125
.175	.713	.669		.509	.312	.200	.262	.293	.315	.300	.175
.225	.663	.622	.567	.467	.289	.203	.256	.281	.300	.298	.225
.275	.641	.583	.532	.429	.283	.201	.245	.271	.287	.284	.275
.325	.608	.554	.496	.402	.268	.200	.237	.259	.277	.278	.325
.375	.571	.527	.469	.383	.258	.199	.230	.251	.267	.268	.375
.425	.541	.503	.450	.365	.251	.195	.222	.236	.254		.425
.475	.512	.481	.429	.346	.242	.194	.219	.223	.245	.245	.475
.550	.460	.423	.379	.310	.211	.194	.195	.207	.226	.237	.550
.650	.402		.322	.254		.185	.209	.193	.215	.234	.650
.750	.354	.371	.302	.236	.187	.184	.184	.213	.215	.226	.750
.800	.347									.223	.800
.850		.359	.295	.231	.218	.186	.188	.207	.207		.850
.900			.288	.226			.187	.203			.900
.950					.232	.185					.950

TABLE 5

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
					$\alpha = 0°$ $\beta = 0°$						
.0	.141	.057	-.011	-.020	-.002	.008	.002	.012		-.005	.0
22.5	.143	.058	-.011	-.025	.000	.009	-.002	.004	.002	.000	22.5
45.0	.143	.058	-.011	-.006	-.002	.007		.006	-.002	-.006	45.0
67.5	.140	.058	-.011	-.027	-.004	.006	-.002		.005	.004	67.5
90.0	.143	.058	-.011	-.028	-.006	-.011	.002	.002	-.005	.004	90.0
112.5	.143	.058		-.028	-.005	-.005	.000	.002	.001	.018	112.5
135.0	.143	.058	-.006	-.028	-.002	-.004	.000	.005	.005	.008	135.0
157.5	.145	.058	-.006	-.026	-.004	-.002	.002	.004	-.004	-.008	157.5
180.0	.145	.058	-.005	-.027	-.002	-.007	.002	.005	.002	-.001	180.0
202.5	.146	.058	-.002	-.026	-.002	-.007	.005	.007	.013	.001	202.5
225.0	.146	.058	-.002	-.026	-.002	.000	.005	.000	.004	.004	225.0
247.5	.147	.058	-.002	-.025	-.001	-.001	.012	.000	.006	.005	247.5
270.0		.058	-.002	-.022	-.001	.000	.005	.000	.006		270.0
292.5	.140	.058	.000	-.022	-.002	.001	.006	-.001	.004	-.006	292.5
315.0		.058	.014	-.023	-.002	.000	.006	.006	-.001	-.002	315.0
337.5		.058	.007	-.022	.000	-.001	.001	.011		.000	337.5
					$\alpha = 2.5°$ $\beta = 0°$						
.0	.196	.096	.018	-.009	-.005	.002	.013	.007		.000	.0
22.5	.186	.084	.012	-.015	-.008	-.004	-.002	.000	-.002	-.009	22.5
45.0	.186	.075	.006	-.011	-.015	-.006		-.004	-.014	-.014	45.0
67.5	.148	.062	-.002	-.042	-.022	-.015	-.005		-.006	-.007	67.5
90.0	.133	.042	-.027	-.047	-.030	-.028	-.005	-.004	-.015	-.009	90.0
112.5	.116	.034		-.057	-.027	-.012	-.007	-.005	-.006	.002	112.5
135.0	.102	.023	-.033	-.051	-.018	-.009	-.004	.001	-.004	.000	135.0
157.5	.094	.018	-.034	-.043	-.014	-.005	.001	.002	-.013	-.018	157.5
180.0	.094	.018	-.034	-.041	-.011	-.002	.006	.006	-.001	-.006	180.0
202.5	.094	.018	-.026	-.046	-.011	-.008	.001	.002	.007	-.005	202.5
225.0	.103	.018	-.026	-.050	-.018	-.012	.004	-.005	-.006	-.006	225.0
247.5	.110	.032	-.025	-.050	-.027	-.019	-.005	-.011	-.005	-.005	247.5
270.0	.127	.043	-.007	-.048	-.034	-.014	-.004	-.019	-.008		270.0
292.5	.143	.055	.030	-.036	-.018	-.009	-.004	-.011	-.008	-.015	292.5
315.0	.162	.074	.018	-.029	-.012	-.009	.000	.000	-.008	-.011	315.0
337.5	.173	.090	.021	-.022	.000	-.004	.002	.004		-.005	337.5
					$\alpha = 5.0°$ $\beta = 0°$						
.0	.225	.110	.031	.007	.009	.010	.014	.005		.000	.0
22.5	.213	.098	.023	.000	.005	.003	-.005	-.005	-.005	-.006	22.5
45.0	.193	.084	.005	-.006	-.014	-.010		-.019	-.022	-.014	45.0
67.5	.153	.056	-.021	-.042	-.029	-.031	-.028		-.024	-.019	67.5
90.0	.117	.027	-.044	-.057	-.044	-.051	-.026	-.026	-.028	-.024	90.0
112.5	.091	.011		-.070	-.038	-.028	-.021	-.019	-.014	-.010	112.5
135.0	.071	-.005	-.052	-.058	-.016	-.021	-.013	-.009	-.012	-.013	135.0
157.5	.063	-.006	-.052	-.041	-.014	-.010	-.010	-.009	-.023	-.027	157.5
180.0	.063	-.006	-.052	-.040	-.007	-.008	-.001	-.003	-.008	-.010	180.0
202.5	.066	-.006	-.039	-.048	-.012	-.012	-.009	-.008	.001	-.013	202.5
225.0	.075	-.006	-.039	-.056	-.021	-.017	-.007	-.017	-.012	-.009	225.0
247.5	.087	.006	-.044	-.063	-.037	-.034	-.019	-.023	-.016	-.015	247.5
270.0	.119	.023	-.025	-.056	-.047	-.041	-.026	-.033	-.023		270.0
292.5	.151	.050	-.007	-.041	-.026	-.028	-.026	-.027	-.022	-.028	292.5
315.0	.177	.078	.030	-.023	-.013	-.020	-.014	-.008	-.015	-.013	315.0
337.5	.198	.107	.043	-.006	.007	-.005	-.002	.001		-.003	337.5
					$\alpha = 7.5°$ $\beta = 0°$						
.0	.270	.160	.065	.033	.027	.028	.023	.017		.014	.0
22.5	.254	.140	.058	.023	.020	.015	.000	-.001	.001	.003	22.5
45.0	.215		.030	.005	-.012	-.014		-.026	-.026	-.014	45.0
67.5	.158	.058	-.008	-.048	-.044	-.057	-.049		-.041	-.034	67.5
90.0	.110	.014	-.057	-.078	-.076	-.082	-.050	-.042	-.038	-.034	90.0
112.5	.072	-.006		-.090	-.059	-.040	-.033	-.023	-.017	-.017	112.5
135.0	.051	-.019	-.064	-.068	-.027	-.026	-.022	-.020	-.017	-.016	135.0
157.5	.044	-.020	-.053	-.041	-.013	-.019	-.026	-.028	-.038	-.042	157.5
180.0	.044	-.020	-.046	-.037	-.002	-.005	.000	.001	-.008	-.016	180.0
202.5	.044	-.015	-.049	-.047	-.010	-.021	-.031	-.029	-.014	-.037	202.5
225.0	.051	-.015	-.060	-.064	-.027	-.026	-.019	-.024	-.019	-.020	225.0
247.5	.071	-.012	-.066	-.086	-.063	-.044	-.027	-.026	-.021	-.020	247.5
270.0	.108	.018	-.041	-.076	-.075	-.070	-.048	-.050	-.038		270.0
292.5	.164	.059	-.012	-.043	-.044	-.054	-.048	-.048	-.040	-.047	292.5
315.0	.210	.094	.043	-.010	-.013	-.024	-.028	-.012	-.020	-.015	315.0
337.5	.249	.146	.072	.013	.016	.001	-.001	.008		.005	337.5

TABLE 5, Continued

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
					$\alpha = 10.0°$		$\beta = 0°$				
.0	.325	.199	.102	.056	.051	.044	.048	.033		.023	.0
22.5	.301	.181	.090	.041	.035	.031	.015	.003	.008	.008	22.5
45.0	.247	.134	.044	.015	-.009	-.022		-.042	-.034	-.021	45.0
67.5	.160	.068	-.009	-.058	-.065	-.087	-.094		-.068	-.057	67.5
90.0	.101	.005	-.076	-.103	-.124	-.129	-.072	-.059	-.059	-.050	90.0
112.5	.049	-.032		-.131	-.086	-.051	-.033	-.028	-.030	-.024	112.5
135.0	.028	-.043	-.075	-.080	-.041	-.037	-.034	-.029	-.023	-.026	135.0
157.5	.028	-.025	-.058	-.050	-.033	-.045	-.058	-.051	-.048	-.042	157.5
180.0	.028	-.026	-.047	-.040	.000	.008	.002	-.008	-.026	-.024	180.0
202.5	.028	-.026	-.055	-.055	-.028	-.050	-.068	-.062	-.026	-.034	202.5
225.0	.028	-.046	-.075	-.077	-.037	-.037	-.031	-.038	-.030	-.034	225.0
247.5	.043	-.031	-.097	-.119	-.090	-.058	-.034	-.036	-.031	-.033	247.5
270.0	.097	.000	-.064	-.105	-.117	-.119	-.079	-.064	-.061		270.0
292.5	.172	.068	-.015	-.057	-.069	-.083	-.090	-.077	-.065	-.082	292.5
315.0	.239	.132	.062	-.001	-.013	-.028	-.035	-.022	-.031	-.015	315.0
337.5	.296	.185	.100	.037	.033	.019	.008	.012		.012	337.5
					$\alpha = 12.5°$		$\beta = 0°$				
.0	.382	.256	.146	.084	.077	.069	.069	.062		.037	.0
22.5	.351	.228	.126	.067	.054	.044	.035	.029	.015	.013	22.5
45.0	.276	.160	.063	.023	-.009	-.021		-.039	-.055	-.023	45.0
67.5	.174	.071	-.011	-.070	-.086	-.104	-.116		-.096	-.092	67.5
90.0	.087	-.012	-.101	-.135	-.165	-.177	-.109	-.056	-.082	-.085	90.0
112.5	.023	-.066		-.176	-.123	-.070	-.047	-.040	-.044	-.037	112.5
135.0	.002	-.066	-.098	-.104	-.069	-.065	-.069	-.046	-.040	-.033	135.0
157.5	.009	-.027	-.070	-.069	-.069	-.089	-.095	-.085	-.060	-.048	157.5
180.0	.023	-.025	-.059	-.044	-.001	.008	-.008	-.015	-.047	-.042	180.0
202.5	.014	-.033	-.062	-.070	-.063	-.096	-.120	-.109	-.030	-.047	202.5
225.0	.007	-.064	-.101	-.097	-.068	-.069	-.072	-.068	-.043	-.047	225.0
247.5	.014	-.064	-.129	-.170	-.126	-.077	-.048	-.049	-.058	-.053	247.5
270.0	.087	-.013	-.088	-.140	-.159	-.175	-.134	-.077	-.093		270.0
292.5	.177	.078	-.013	-.065	-.091	-.099	-.114	-.138	-.093	-.117	292.5
315.0	.273	.152	.075	.002	-.006	-.015	-.027	-.036	-.047	-.020	315.0
337.5	.350	.224	.133	.057	.053	.048	.032	.025		.015	337.5
					$\alpha = 15.0°$		$\beta = 0°$				
.0	.434	.304	.178	.112	.099	.093	.089	.083		.067	.0
22.5	.402	.268	.157	.088	.072	.067	.055	.046	.035	.030	22.5
45.0	.311	.184	.078	.032	-.006	-.015		-.039	-.050	-.044	45.0
67.5	.185	.070	-.012	-.078	-.097	-.113	-.134		-.156	-.158	67.5
90.0	.062	-.036	-.104	-.155	-.193	-.188	-.132	-.127	-.107	-.095	90.0
112.5	-.019	-.098		-.207	-.144	-.085	-.088	-.074	-.064	-.047	112.5
135.0	-.023	-.088	-.128	-.120	-.132	-.125	-.102	-.068	-.060	-.053	135.0
157.5	-.014	-.045	-.084	-.088	-.109	-.134	-.132	-.076	-.068	-.061	157.5
180.0	.005	-.044	-.058	-.055	-.006	-.006	-.044	-.067	-.068	-.067	180.0
202.5	-.007	-.044	-.068	-.088	-.104	-.161	-.141	-.077	-.047	-.071	202.5
225.0	-.021	-.077	-.111	-.111	-.133	-.128	-.126	-.074	-.067	-.067	225.0
247.5	-.021	-.096	-.158	-.211	-.140	-.091	-.070	-.086	-.074	-.070	247.5
270.0	.058	-.037	-.103	-.155	-.189	-.190	-.142	-.126	-.132		270.0
292.5	.175	.059	-.014	-.078	-.110	-.118	-.135	-.145	-.156	-.167	292.5
315.0	.296	.180	.088	.013	-.004	-.018	-.027	-.033	-.043	-.039	315.0
337.5	.389	.261	.162	.079	.067	.061	.046	.047		.035	337.5
					$\alpha = 0°$		$\beta = -5°$				
.0	.122	.039	-.026	-.064	-.040	-.024	-.010	-.015		-.023	.0
22.5	.154	.058	-.004	-.041	-.034	-.017	-.017	-.016	-.022	-.024	22.5
45.0	.180	.083	.015	-.008	-.015	-.008		-.006	-.013	-.020	45.0
67.5	.218	.103	.032	-.013	-.003	-.001	.002		.001	.001	67.5
90.0	.222	.113	.033	-.005	-.003	.005	.006	.008	.006	.002	90.0
112.5	.215	.113		-.009	-.003	.002	.001	.002	.000	.010	112.5
135.0	.196	.092	.021	-.023	-.015	-.012	-.007	-.017	-.008	-.002	135.0
157.5	.163	.070	.001	-.038	-.029	-.022	-.014	-.014	-.020	-.021	157.5
180.0	.132	.041	-.020	-.044	-.040	-.031	-.022	-.016	-.026	-.023	180.0
202.5	.112	.026	-.030	-.059	-.034	-.021	-.014	-.012	-.031	-.015	202.5
225.0	.092	.009	-.039	-.055	-.023	-.009	-.002	-.007	-.014	-.003	225.0
247.5	.083	.009	-.039	-.051	-.010	-.001	.001	-.006	-.012	-.008	247.5
270.0	.073	.009	-.039	-.047		.003	.005	.001	-.003		270.0
292.5	.078	.009	-.039	-.047	-.016	.002	.000	-.001	-.012	-.016	292.5
315.0	.085	.009	-.028	-.050	-.020	-.006	-.002	-.006	-.012	-.020	315.0
337.5	.096	.020	-.026	-.056	-.029	-.010	-.007	-.014	-.023	-.019	337.5

TABLE 5, Continued

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
					$\alpha = 2.5°$ $\beta = -5°$						
.0	.136	.054	-.015	-.049	-.037	-.027	-.019	-.024		-.027	.0
22.5	.173	.088	.015	-.030	-.022	-.015	-.015	-.015	-.013	-.013	22.5
45.0	.195	.109	.024	.003	-.006	.000		-.003	-.003	-.007	45.0
67.5	.226	.116	.037	-.006	.001	.003	.003		.007	.003	67.5
90.0	.215	.109	.022	-.012	-.013	-.003	-.005	-.001	-.001	-.003	90.0
112.5	.189	.084		-.022	-.023	-.020	-.017	-.013	-.017	-.006	112.5
135.0	.161	.061	-.002	-.043	-.036	-.030	-.029	-.047	-.022	-.021	135.0
157.5	.122	.040	-.021	-.049	-.048	-.036	-.027	-.029	-.029	-.029	157.5
180.0	.095	.012	-.037	-.059	-.038	-.029	-.022	-.017	-.016	-.017	180.0
202.5	.080	.002	-.044	-.059	-.029	-.020	-.015	-.014	-.027	-.010	202.5
225.0	.071	.002	-.056	-.052	-.019	-.008	-.007	-.010	-.013	-.022	225.0
247.5	.064	.002	-.056	-.050	-.014	-.005	-.001	-.001	-.003	-.007	247.5
270.0	.078	.002	-.056	-.050		-.009	-.008	-.010	-.015		270.0
292.5	.080	.002	-.056	-.057	-.035	-.015	-.013	-.010	-.015	-.017	292.5
315.0	.097	.014	-.031	-.065	-.043	-.028	-.022	-.017	-.019	-.024	315.0
337.5	.125	.028	-.022	-.055	-.044	-.033	-.030	-.029	-.003	-.029	337.5
					$\alpha = 5.0°$ $\beta = -5°$						
.0	.200	.100	.022	-.028	-.017	-.020	-.008	-.009		-.019	.0
22.5	.236	.130	.044	-.002	.003	.007	.005	.007	.006	-.001	22.5
45.0	.246	.139	.053	.023	.015	.019		.015	.010	.010	45.0
67.5	.241	.138	.053	.005	.008	.009	.009		.008	.006	67.5
90.0	.213	.112	.028	-.017	-.023	-.010	-.010	-.001	-.013	-.015	90.0
112.5	.169	.073		-.035	-.042	-.036	-.033	-.023	-.036	-.022	112.5
135.0	.130	.033	-.018	-.064	-.055	-.040	-.034	-.033	-.035	-.028	135.0
157.5	.090	.018	-.035	-.062	-.049	-.028	-.020	-.013	-.020	-.026	157.5
180.0	.066	-.001	-.041	-.058	-.031	-.016	-.013	-.010	-.013	-.022	180.0
202.5	.066	-.001	-.028	-.051	-.019	-.009	-.008	-.019	-.042	-.030	202.5
225.0	.065	-.001	-.037	-.044	-.009	.007	.009	.001	-.003	-.009	225.0
247.5	.059	-.001	-.046	-.050	-.014	-.002	-.003	-.016	-.022	-.029	247.5
270.0	.067	-.001	-.046	-.057		-.009	-.009	-.015	-.019		270.0
292.5	.082	-.001	-.046	-.065	-.051	-.026	-.015	-.015	-.017	-.017	292.5
315.0	.115	.035	-.026	-.066	-.065	-.047	-.034	-.036	-.031	-.029	315.0
337.5	.154	.064	-.002	-.048	-.045	-.040	-.034	-.038	.019	-.035	337.5
					$\alpha = 7.5°$ $\beta = -5°$						
.0	.236	.142	.044	-.010	-.005	-.009	-.005	-.007		-.013	.0
22.5	.272	.168	.078	.015	.016	.024	.019	.014	.017	.013	22.5
45.0	.272	.169	.078	.035	.028	.023		.014	.019	.017	45.0
67.5	.254	.149	.061	.005	.003	-.001	-.006		-.002	-.002	67.5
90.0	.200	.105	.022	-.028	-.040	-.038	-.047	-.037	-.038	-.042	90.0
112.5	.137	.046		-.062	-.078	-.081	-.066	-.049	-.059	-.043	112.5
135.0	.095	.012	-.050	-.093	-.084	-.058	-.049	-.044	-.040	-.037	135.0
157.5	.054	-.001	-.059	-.077	-.055	-.030	-.022	-.019	-.023	-.030	157.5
180.0	.046	-.021	-.059	-.065	-.035	-.027	-.024	-.024	-.022	-.028	180.0
202.5	.046	-.013	-.037	-.055	-.016	-.014	-.009	-.021	-.044	-.035	202.5
225.0	.046	-.014	-.040	-.049	-.014	-.005	.000	-.013	-.014	-.023	225.0
247.5	.046	-.014	-.054	-.065	-.027	-.023	-.028	-.034	-.034	-.029	247.5
270.0	.054	-.017	-.069	-.076		-.029	-.028	-.026	-.028		270.0
292.5	.083	-.005	-.054	-.086	-.093	-.058	-.043	-.034	-.034	-.031	292.5
315.0	.130	.035	-.027	-.078	-.088	-.086	-.069	-.057	-.051	-.047	315.0
337.5	.190	.083	.014	-.043	-.053	-.053	-.053	-.045	.016	-.040	337.5
					$\alpha = 10.0°$ $\beta = -5°$						
.0	.296	.192	.083	.017	.027	.015	.021	.002		.006	.0
22.5	.338	.221	.109	.044	.047	.047	.042	.031	.031	.037	22.5
45.0	.315	.207	.102	.054	.040	.035		.020	.016	.029	45.0
67.5	.276	.157	.064	.007	.001	-.009	-.022		-.021	-.019	67.5
90.0	.195	.099	.014	-.045	-.061	-.066	-.081	-.093	-.062	-.066	90.0
112.5	.118	.026		-.084	-.115	-.129	-.100	-.068	-.084	-.065	112.5
135.0	.065	-.014	-.077	-.119	-.115	-.064	-.042	-.043	-.045	-.038	135.0
157.5	.031	-.027	-.078	-.087	-.055	-.037	-.033	-.031	-.031	-.036	157.5
180.0	.031	-.027	-.062	-.069	-.044	-.047	-.058	-.047	-.029	-.031	180.0
202.5	.031	-.020	-.033	-.051	-.012	-.002	-.009	-.022	-.048	-.030	202.5
225.0	.031	-.020	-.045	-.056	-.033	-.029	-.040	-.048	-.041	-.028	225.0
247.5	.031	-.030	-.062	-.076	-.041	-.037	-.042	-.037	-.029	-.031	247.5
270.0	.040	-.030	-.082	-.100		-.049	-.035	-.031	-.034		270.0
292.5	.080	-.011	-.064	-.107	-.140	-.122	-.070	-.058	-.057	-.057	292.5
315.0	.150	.052	-.026	-.081	-.097	-.113	-.122	-.092	-.069	-.064	315.0
337.5	.234	.119	.040	-.026	-.045	-.049	-.057	-.061	.019	-.040	337.5

TABLE 5, Continued

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
					$\alpha = 12.5°$ $\beta = -5°$						
.0	.356	.242	.129	.055	.062	.051	.051	.041		.021	.0
22.5	.397	.267	.154	.086	.081	.074	.069	.060	.059	.052	22.5
45.0	.362	.244	.137	.077	.058	.047		.034	.024	.027	45.0
67.5	.285	.168	.076	.005	-.006	-.015	-.028		-.044	-.049	67.5
90.0	.183	.072	-.001	-.066	-.090	-.100	-.107	-.129	-.121	-.100	90.0
112.5	.078	-.017		-.117	-.159	-.173	-.120	-.066	-.092	-.084	112.5
135.0	.019	-.060	-.118	-.162	-.140	-.077	-.062	-.063	-.050	-.045	135.0
157.5	-.001	-.046	-.095	-.098	-.072	-.076	-.076	-.055	-.047	-.042	157.5
180.0	.011	-.040	-.066	-.074	-.079	-.092	-.102	-.088	-.055	-.049	180.0
202.5	.020	-.026	-.048	-.054	-.003	-.008	-.012	-.041	-.064	-.048	202.5
225.0	.012	-.026	-.058	-.072	-.072	-.094	-.106	-.086	-.047	-.044	225.0
247.5	-.002	-.058	-.089	-.100	-.071	-.081	-.079	-.043	-.037	-.040	247.5
270.0	.009	-.058	-.121	-.143		-.077	-.055	-.054	-.045		270.0
292.5	.070	-.017	-.080	-.127	-.169	-.174	-.094	-.072	-.098	-.099	292.5
315.0	.176	.066	-.012	-.074	-.093	-.111	-.126	-.135	-.112	-.090	315.0
337.5	.281	.155	.071	.005	-.022	-.027	-.042	-.034	.019	-.041	337.5
					$\alpha = 15.0°$ $\beta = -5°$						
.0	.410	.295	.182	.091	.095	.085	.088	.080		.051	.0
22.5	.442	.321	.203	.121	.112	.109	.102	.099	.086	.078	22.5
45.0	.395	.281	.168	.102	.076	.063		.053	.036	.042	45.0
67.5	.310	.175	.085	.009	-.007	-.021	-.024		-.053	-.064	67.5
90.0	.175	.072	-.012	-.079	-.108	-.117	-.117	-.138	-.152	-.159	90.0
112.5	.054	-.040		-.155	-.191	-.172	-.108	-.087	-.095	-.081	112.5
135.0	-.017	-.083	-.163	-.200	-.136	-.078	-.083	-.073	-.074	-.064	135.0
157.5	-.017	-.069	-.100	-.100	-.116	-.120	-.092	-.070	-.064	-.065	157.5
180.0	.002	-.041	-.060	-.083	-.099	-.108	-.114	-.078	-.071	-.073	180.0
202.5	.018	-.032	-.053	-.053	-.005	-.015	-.041	-.064	-.081	-.067	202.5
225.0	-.013	-.038	-.076	-.083	-.121	-.145	-.152	-.072	-.056	-.063	225.0
247.5	-.026	-.089	-.118	-.123	-.107	-.114	-.101	-.069	-.056	-.066	247.5
270.0	-.007	-.087	-.165	-.179		-.086	-.070	-.077	-.071		270.0
292.5	.071	-.020	-.104	-.140	-.183	-.179	-.122	-.156	-.150	-.117	292.5
315.0	.203	.090	.005	-.062	-.084	-.099	-.110	-.122	-.123	-.116	315.0
337.5	.338	.212	.112	.028	.003	.002	-.009	-.008	.014	-.031	337.5
					$\alpha = 0°$ $\beta = -10°$						
.0	.088	-.011	-.064	-.113	-.117	-.105	-.080	-.062	-.066	-.065	.0
22.5	.150	.051	-.018	-.070	-.080	-.090	-.094	-.075	-.068	-.068	22.5
45.0	.233	.117	.036	-.003	-.026	-.028		-.043	-.037	-.036	45.0
67.5	.288	.167	.084	.020	.019	.014	-.001		-.002	.008	67.5
90.0	.314	.190	.098	.037	.037	.027	.020	.014	.014	.021	90.0
112.5	.297	.174		.023	.021	.012	.001	-.005	.000	.016	112.5
135.0	.248	.131	.044	-.014	-.021	-.031	-.041	-.034	-.031	-.020	135.0
157.5	.172	.065	-.013	-.061	-.075	-.087	-.092	-.073	-.068	-.065	157.5
180.0	.091	.006	-.057	-.097	-.119	-.118	-.085	-.068	-.071	-.072	180.0
202.5	.046	-.034	-.083	-.114	-.097	-.058	-.038	-.035	-.051	-.042	202.5
225.0	.026	-.037	-.083	-.084	-.054	-.040	-.033	-.035	-.034	-.037	225.0
247.5	.026	-.037	-.063	-.065	-.037	-.050	-.062	-.065	-.047	-.040	247.5
270.0	.026	-.027	-.063	-.052	-.019	-.001	-.003	-.016	-.028		270.0
292.5	.026	-.027	-.063	-.058	-.031	-.045	-.059	-.063	-.052	-.048	292.5
315.0	.028	-.027	-.064	-.082	-.047	-.040	-.036	-.037	-.034	-.040	315.0
337.5	.039	-.027	-.076	-.110	-.078	-.045	-.027	-.034	-.030	-.043	337.5
					$\alpha = 2.5°$ $\beta = -10°$						
.0	.121	.026	-.043	-.092	-.093	-.099	-.092	-.065	-.072	-.073	.0
22.5	.201	.105	.015	-.038	-.045	-.049	-.061	-.049	-.049	-.048	22.5
45.0	.264	.159	.066	.027	.007	.009		-.012	-.010	-.007	45.0
67.5	.316	.199	.097	.041	.040	.031	.030		.020	.024	67.5
90.0	.317	.194	.097	.044	.040	.035	.024	.020	.015	.017	90.0
112.5	.279	.157		.015	.006	-.002	-.010	-.009	-.017	-.001	112.5
135.0	.208	.103	.019	-.036	-.043	-.050	-.068	-.051	-.049	-.044	135.0
157.5	.128	.045	-.031	-.069	-.087	-.105	-.086	-.068	-.069	-.064	157.5
180.0	.066	-.019	-.072	-.107	-.113	-.080	-.054	-.035	-.047	-.048	180.0
202.5	.022	-.032	-.082	-.102	-.066	-.036	-.026	-.034	-.049	-.040	202.5
225.0	.031	-.027	-.070	-.070	-.036	-.040	-.040	-.040	-.042	-.035	225.0
247.5	.031	-.026	-.057	-.052	-.019	-.009	-.014	-.029	-.031	-.041	247.5
270.0	.031	-.026	-.056	-.048	-.017	-.015	-.028	-.022	-.038		270.0
292.5	.031	-.026	-.065	-.062	-.034	-.037	-.035	-.030	-.033	-.038	292.5
315.0	.035	-.026	-.069	-.094	-.058	-.036	-.023	-.030	-.030	-.040	315.0
337.5	.076	-.017	-.066	-.112	-.095	-.065	-.036	-.049	-.092	-.049	337.5

TABLE 5, Continued

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION										θ, deg
	1	2	3	4	5	6	7	8	9	10	
					$\alpha = 5.0°$		$\beta = -10°$				
.0	.178	.063	-.025	-.066	-.061	-.082	-.085	-.089	-.070	-.056	.0
22.5	.264	.142	.039	-.008	-.014	-.014	-.027	-.033	-.029	-.016	22.5
45.0	.309	.192	.091	.051	.040	.037		.020	.006	.019	45.0
67.5	.329	.213	.107	.055	.055	.048	.040		.022	.036	67.5
90.0	.298	.191	.090	.040	.031	.021	.016	.006	-.003	.007	90.0
112.5	.239	.134		-.002	-.019	-.031	-.043	-.040	-.044	-.024	112.5
135.0	.162	.069	-.021	-.058	-.080	-.087	-.107	-.079	-.078	-.073	135.0
157.5	.084	.007	-.066	-.087	-.120	-.121	-.072	-.062	-.068	-.062	157.5
180.0	.025	-.043	-.096	-.113	-.089	-.057	-.042	-.030	-.041	-.044	180.0
202.5	.019	-.036	-.070	-.091	-.051	-.043	-.035	-.037	-.051	-.042	202.5
225.0	.019	-.038	-.059	-.061	-.034	-.043	-.047	-.055	-.049	-.043	225.0
247.5	.019	-.024	-.046	-.045	-.013	.000	-.002	-.020	-.023	-.037	247.5
270.0	.019	-.036	-.057	-.054	-.037	-.052	-.062	-.052	-.041		270.0
292.5	.019	-.043	-.069	-.073	-.050	-.043	-.031	-.033	-.037	-.035	292.5
315.0	.033	-.043	-.077	-.120	-.111	-.059	-.037	-.036	-.036	-.043	315.0
337.5	.099	-.005	-.054	-.106	-.114	-.134	-.072	-.076	-.105	-.069	337.5
					$\alpha = 7.5°$		$\beta = -10°$				
.0	.220	.111	.022	-.038	-.034	-.050	-.050	-.066	-.079	-.054	.0
22.5	.312	.195	.095	.029	.022	.022	.010	.001	-.010	.002	22.5
45.0	.354	.239	.127	.077	.064	.064		.047	.026	.027	45.0
67.5	.352	.232	.128	.064	.061	.050	.047		.020	.030	67.5
90.0	.298	.181	.084	.027	.014	.007	.001	-.013	-.033	-.019	90.0
112.5	.207	.106		-.034	-.056	-.066	-.081	-.080	-.079	-.051	112.5
135.0	.117	.030	-.047	-.098	-.123	-.131	-.141	-.077	-.106	-.095	135.0
157.5	.034	-.027	-.086	-.126	-.148	-.090	-.049	-.044	-.067	-.057	157.5
180.0	.004	-.054	-.098	-.121	-.069	-.064	-.055	-.048	-.049	-.047	180.0
202.5	.008	-.043	-.069	-.081	-.063	-.076	-.090	-.069	-.050	-.047	202.5
225.0	.020	-.030	-.052	-.059	-.020	-.005	-.010	-.035	-.044	-.049	225.0
247.5	.019	-.030	-.052	-.057	-.030	-.022	-.034	-.050	-.055	-.058	247.5
270.0	.007	-.044	-.073	-.072	-.057	-.059	-.057	-.051	-.045		270.0
292.5	.007	-.056	-.093	-.108	-.077	-.056	-.041	-.051	-.044	-.044	292.5
315.0	.048	-.037	-.090	-.141	-.166	-.141	-.057	-.055	-.051	-.050	315.0
337.5	.136	.024	-.035	-.100	-.122	-.135	-.150	-.141	-.109	-.091	337.5
					$\alpha = 10.0°$		$\beta = -10°$				
.0	.276	.153	.053	-.008	.000	-.020	-.016	-.036	-.056	-.047	.0
22.5	.368	.235	.123	.061	.049	.056	.044	.037	.026	.022	22.5
45.0	.393	.269	.149	.097	.088	.085		.068	.048	.048	45.0
67.5	.362	.246	.132	.068	.066	.054	.049		.021	.019	67.5
90.0	.291	.164	.071	.012	-.014	-.010	-.016	-.033	-.051	-.045	90.0
112.5	.172	.069		-.054	-.086	-.099	-.113	-.111	-.148	-.079	112.5
135.0	.083	-.005	-.090	-.128	-.157	-.158	-.122	-.086	-.107	-.088	135.0
157.5	-.005	-.056	-.124	-.152	-.140	-.079	-.059	-.066	-.061	-.049	157.5
180.0	-.009	-.073	-.118	-.111	-.094	-.088	-.072	-.051	-.051	-.052	180.0
202.5	.002	-.048	-.072	-.079	-.080	-.113	-.101	-.070	-.079	-.056	202.5
225.0	.007	-.037	-.059	-.057	-.005	.001	-.019	-.056	-.054	-.055	225.0
247.5	.000	-.047	-.059	-.073	-.064	-.079	-.088	-.068	-.061	-.061	247.5
270.0	-.014	-.073	-.098	-.087	-.090	-.093	-.098	-.056	-.056		270.0
292.5	-.001	-.084	-.117	-.144	-.125	-.073	-.059	-.073	-.056	-.051	292.5
315.0	.056	-.034	-.092	-.148	-.178	-.172	-.104	-.086	-.084	-.080	315.0
337.5	.162	.043	-.013	-.084	-.111	-.118	-.137	-.137	-.109	-.101	337.5
					$\alpha = 12.5°$		$\beta = -10°$				
.0	.342	.221	.111	.035	.043	.023	.026	.022	-.014	-.019	.0
22.5	.430	.302	.193	.105	.099	.094	.085	.073	.063	.061	22.5
45.0	.439	.318	.195	.134	.118	.108		.086	.071	.069	45.0
67.5	.401	.265	.144	.080	.071	.058	.047		.022	.016	67.5
90.0	.285	.170	.077	.001	-.019	-.030	-.031	-.054	-.076	-.084	90.0
112.5	.156	.053		-.080	-.116	-.135	-.150	-.142	-.180	-.136	112.5
135.0	.041	-.034	-.116	-.162	-.190	-.150	-.085	-.106	-.097	-.071	135.0
157.5	-.035	-.091	-.149	-.177	-.127	-.086	-.084	-.073	-.077	-.065	157.5
180.0	-.021	-.071	-.103	-.111	-.136	-.134	-.087	-.066	-.065	-.079	180.0
202.5	-.012	-.035	-.067	-.080	-.090	-.078	-.102	-.080	-.088	-.079	202.5
225.0	.012	-.037	-.054	-.059	-.015	-.014	-.062	-.072	-.067	-.072	225.0
247.5	-.015	-.054	-.072	-.091	-.109	-.149	-.100	-.071	-.066	-.079	247.5
270.0	-.030	-.091	-.127	-.116	-.127	-.118	-.093	-.070	-.076		270.0
292.5	-.011	-.091	-.137	-.171	-.155	-.094	-.086	-.093	-.077	-.065	292.5
315.0	.089	-.018	-.083	-.141	-.164	-.180	-.169	-.183	-.136	-.104	315.0
337.5	.209	.091	.023	-.054	-.081	-.084	-.109	-.114	-.098	-.120	337.5

TABLE 5, Continued

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
					α=15.0°		β=-10°				
.0	.402	.266	.160	.077	.079	.059	.066	.049	.027	.027	.0
22.5	.492	.349	.231	.142	.140	.130	.121	.113	.100	.099	22.5
45.0	.492	.354	.231	.156	.138	.129		.112	.098	.100	45.0
67.5	.409	.285	.160	.087	.071	.055	.051		.022	.017	67.5
90.0	.271	.157	.057	-.013	-.036	-.048	-.052	-.067	-.092	-.092	90.0
112.5	.115	.018		-.106	-.145	-.159	-.172	-.169	-.197	-.155	112.5
135.0	-.013	-.077	-.164	-.201	-.202	-.126	-.104	-.113	-.106	-.083	135.0
157.5	-.061	-.117	-.188	-.179	-.129	-.102	-.095	-.107	-.092	-.084	157.5
180.0	-.046	-.083	-.097	-.134	-.182	-.149	-.097	-.083	-.092	-.104	180.0
202.5	-.011	-.056	-.066	-.087	-.083	-.093	-.111	-.086	-.105	-.102	202.5
225.0	-.008	-.046	-.066	-.074	-.065	-.070	-.102	-.083	-.081	-.086	225.0
247.5	-.035	-.074	-.090	-.122	-.158	-.157	-.090	-.078	-.087	-.092	247.5
270.0	-.061	-.122	-.175	-.147	-.138	-.108	-.099	-.091	-.097		270.0
292.5	-.024	-.102	-.155	-.192	-.183	-.102	-.106	-.111	-.100	-.079	292.5
315.0	.098	.004	-.069	-.129	-.151	-.171	-.177	-.192	-.169	-.148	315.0
337.5	.252	.138	.057	-.026	-.059	-.061	-.083	-.077	-.087	-.092	337.5
					α= 0°		β=-15°				
.0	.038	-.058	-.119	-.165	-.177	-.153	-.071	-.070	-.099	-.092	.0
22.5	.154	.051	-.042	-.083	-.107	-.121	-.144	-.135	-.105	-.124	22.5
45.0	.281	.172	.063	.028	-.001	-.009		-.043	-.056	-.039	45.0
67.5		.244	.144	.079	.071	.065	.044		.027	.030	67.5
90.0	.423	.292	.173	.107	.102	.088	.076	.068	.057	.053	90.0
112.5	.397	.259		.088	.071	.062	.047	.037	.026	.041	112.5
135.0	.306	.177	.076	.018	.006	-.020	-.030	-.035	-.047	-.025	135.0
157.5	.179	.066	-.027	-.056	-.093	-.111	-.125	-.132	-.107	-.121	157.5
180.0	.053	-.038	-.110	-.147	-.182	-.144	-.069	-.082	-.103	-.095	180.0
202.5	-.021	-.103		-.205	-.144	-.091	-.081	-.074	-.076	-.063	202.5
225.0	-.033	-.098	-.123	-.118	-.137	-.141	-.137	-.075	-.060	-.060	225.0
247.5	-.017	-.059	-.085	-.096	-.109	-.145	-.132	-.092	-.062	-.060	247.5
270.0	.001	-.044	-.066	-.056	-.002	-.005	-.039	-.067	-.065		270.0
292.5	-.015	-.057	-.082	-.084	-.090	-.135	-.130	-.112	-.071	-.067	292.5
315.0	-.028	-.089	-.105	-.105	-.112	-.132	-.130	-.072	-.064	-.061	315.0
337.5	-.030	-.108	-.156	-.204	-.131	-.099	-.065	-.078	-.152	-.067	337.5
					α= 2.5°		β= -15°				
.0	.078	-.025	-.094	-.145	-.146	-.153	-.161	-.152	-.138	-.140	.0
22.5	.211	.102	.002	-.053	-.069	-.072	-.092	-.097	-.105	-.063	22.5
45.0	.333	.213	.098	.051	.032	.028		-.001	-.021	-.013	45.0
67.5	.411	.274	.164	.097	.095	.086	.070		.049	.050	67.5
90.0	.426	.288	.172	.110	.099	.088	.079	.064	.056	.050	90.0
112.5	.362	.234		.070	.049	.036	.022	.019	-.004	.012	112.5
135.0	.250	.142	.044	-.011	-.029	-.046	-.064	-.076	-.085	-.081	135.0
157.5	.117	.034	-.050	-.075	-.114	-.147	-.158	-.163	-.114	-.113	157.5
180.0	.013	-.070	-.141	-.166	-.193	-.125	-.081	-.072	-.071	-.062	180.0
202.5	-.032	-.111	-.165	-.195	-.130	-.086	-.090	-.079	-.075	-.060	202.5
225.0	-.027	-.083	-.115	-.107	-.131	-.165	-.083	-.077	-.068	-.056	225.0
247.5	-.013	-.050	-.083	-.083	-.082	-.078	-.084	-.081	-.076	-.067	247.5
270.0	-.007	-.044	-.072	-.060	-.027	-.013	-.063	-.063	-.062		270.0
292.5	-.017	-.057	-.087	-.088	-.111	-.189	-.139	-.054	-.056	-.054	292.5
315.0	-.039	-.098	-.124	-.141	-.123	-.114	-.140	-.075	-.051	-.056	315.0
337.5	-.006	-.096	-.145	-.201	-.153	-.090	-.050	-.086	-.145	-.060	337.5
					α= 5.0°		β= -15°				
.0	.141	.023	-.058	-.108	-.105	-.133	-.127	-.145	-.149	-.128	.0
22.5	.274	.153	.046	-.009	-.019	-.022	-.037	-.041	-.058	-.021	22.5
45.0	.382	.264	.139	.089	.075	.070		.044	.021	.030	45.0
67.5	.452	.307	.202	.121	.115	.107	.100		.075	.073	67.5
90.0	.428	.294	.185	.108	.098	.089	.086	.064	.056	.047	90.0
112.5	.341	.216		.048	.028	.019	.005	.003	-.030	-.016	112.5
135.0	.213	.107	.014	-.037	-.063	-.075	-.091	-.107	-.114	-.120	135.0
157.5	.076	.008	-.065	-.103	-.139	-.167	-.134	-.157	-.119	-.100	157.5
180.0	-.011	-.095	-.147	-.185	-.169	-.100	-.079	-.070	-.068	-.062	180.0
202.5	-.034	-.101	-.140	-.168	-.127	-.107	-.089	-.066	-.068	-.066	202.5
225.0	-.020	-.063	-.088	-.097	-.125	-.174	-.114	-.070	-.057	-.059	225.0
247.5	.001	-.039	-.063	-.065	-.024	-.022	-.057	-.065	-.058	-.056	247.5
270.0	-.011	-.039	-.063	-.065	-.075	-.079	-.117	-.070	-.064		270.0
292.5	-.019	-.066	-.087	-.093	-.131	-.154	-.082	-.062	-.055	-.058	292.5
315.0	-.034	-.103	-.141	-.187	-.131	-.082	-.089	-.066	-.055	-.051	315.0
337.5	.023	-.076	-.121	-.185	-.170	-.128	-.070	-.093	-.138	-.058	337.5

TABLE 5, Concluded

PRESSURE COEFFICIENT DATA FOR BODY ALONE

θ, deg	C_p AT BODY STATION 1	2	3	4	5	6	7	8	9	10	θ, deg
					$\alpha = 7.5°$		$\beta = -15°$				
.0	.192	.057	-.009	-.074	-.071	-.089	-.092	-.113	-.134	-.096	.0
22.5	.335	.206	.100	.029	.020	.021	.000	-.002	-.020	-.008	22.5
45.0	.427	.299	.178	.125	.109	.106		.079	.057	.067	45.0
67.5	.468	.328	.206	.137	.133	.120	.111		.086	.088	67.5
90.0	.418	.284	.171	.104	.088	.075	.072	.053	.039	.027	90.0
112.5	.305	.185		.027	.001	-.016	-.028	-.028	-.060	-.043	112.5
135.0	.169	.064	-.019	-.071	-.093	-.116	-.132	-.151	-.147	-.120	135.0
157.5	.027	-.039	-.107	-.131	-.167	-.159	-.120	-.135	-.118	-.081	157.5
180.0	-.044	-.119	-.176	-.197	-.148	-.095	-.088	-.077	-.079	-.076	180.0
202.5	-.044	-.101	-.134	-.146	-.139	-.148	-.092	-.071	-.083	-.081	202.5
225.0	-.034	-.065	-.085	-.091	-.113	-.146	-.125	-.082	-.077	-.076	225.0
247.5	.001	-.046	-.065	-.058	-.005	-.033	-.065	-.071	-.070	-.071	247.5
270.0	-.031	-.059	-.081	-.083	-.127	-.151	-.090	-.070	-.071		270.0
292.5	-.039	-.100	-.117	-.114	-.137	-.160	-.091	-.072	-.071	-.071	292.5
315.0	-.039	-.116	-.159	-.209	-.146	-.090	-.090	-.075	-.068	-.057	315.0
337.5	.057	-.055	-.097	-.165	-.180	-.174	-.132	-.161	-.142	-.048	337.5
					$\alpha = 10.0°$		$\beta = -15°$				
.0	.254	.109	.027	-.034	-.025	-.051	-.055	-.074	-.103	-.053	.0
22.5	.395	.261	.153	.072	.067	.065	.042	.036	.029	.030	22.5
45.0	.479	.342	.222	.156	.140	.139		.111	.086	.091	45.0
67.5	.491	.342	.219	.147	.141	.131	.114		.092	.090	67.5
90.0	.402	.275	.155	.088	.074	.054	.054	.033	.021	.011	90.0
112.5	.268	.141		-.001	-.033	-.055	-.076	-.067	-.096	-.072	112.5
135.0	.120	.023	-.066	-.111	-.132	-.153	-.173	-.191	-.186	-.074	135.0
157.5	-.017	-.087	-.151	-.166	-.181	-.120	-.116	-.132	-.116	-.061	157.5
180.0	-.072	-.155	-.210	-.191	-.142	-.111	-.105	-.107	-.107	-.088	180.0
202.5	-.053	-.089	-.125	-.153	-.168	-.174	-.096	-.085	-.118	-.106	202.5
225.0	-.015	-.060	-.087	-.091	-.077	-.117	-.121	-.093	-.089	-.100	225.0
247.5	-.015	-.060	-.074	-.082	-.051	-.078	-.096	-.085	-.090	-.083	247.5
270.0	-.045	-.076	-.101	-.118	-.176	-.162	-.092	-.085	-.093		270.0
292.5	-.072	-.141	-.158	-.147	-.132	-.118	-.105	-.100	-.089	-.035	292.5
315.0	-.037	-.122	-.161	-.215	-.191	-.113	-.113	-.113	-.085	-.046	315.0
337.5	.107	-.030	-.067	-.142	-.173	-.179	-.186	-.194	-.140	-.104	337.5
					$\alpha = 12.5°$		$\beta = -15°$				
.0	.324	.179	.076	.019	.020	-.004	-.002	-.021	-.042	-.026	.0
22.5	.474	.319	.205	.127	.121	.112	.097	.084	.078	.071	22.5
45.0	.548	.395	.267	.196	.175	.168		.146	.124	.128	45.0
67.5	.516	.368	.240	.168	.156	.140	.126		.102	.102	67.5
90.0	.402	.267	.154	.083	.058	.036	.042	.026	.005	.002	90.0
112.5	.231	.114		-.028	-.063	-.085	-.096	-.098	-.121	-.095	112.5
135.0	.069	-.020	-.102	-.141	-.167	-.183	-.200	-.207	-.189	-.100	135.0
157.5	-.065	-.126	-.185	-.197	-.163	-.111	-.127	-.128	-.107	-.077	157.5
180.0	-.084	-.161	-.210	-.184	-.147	-.140	-.112	-.118	-.110	-.093	180.0
202.5	-.056	-.087	-.108	-.163	-.198	-.177	-.098	-.103	-.128	-.107	202.5
225.0	-.018	-.056	-.072	-.079	-.048	-.109	-.109	-.090	-.099	-.096	225.0
247.5	-.057	-.087	-.089	-.112	-.126	-.140	-.105	-.098	-.110	.000	247.5
270.0	-.068	-.096	-.130	-.158	-.173	-.147	-.104	-.105	-.103		270.0
292.5	-.093	-.176	-.191	-.175	-.140	-.113	-.128	-.116	-.090	.002	292.5
315.0	-.014	-.103	-.153	-.203	-.210	-.170	-.162	-.131	-.098	-.019	315.0
337.5	.146	.020	-.029	-.103	-.147	-.149	-.159	-.156	-.127	-.124	337.5
					$\alpha = 15.0°$		$\beta = -15°$				
.0	.381	.235	.131	.062	.068	.040	.052	.031	.007	.012	.0
22.5	.533	.390	.271	.175	.159	.158	.149	.139	.126	.123	22.5
45.0	.591	.444	.298	.230	.210	.204		.182	.156	.163	45.0
67.5	.536	.393	.264	.176	.164	.148	.143		.113	.114	67.5
90.0	.396	.269	.149	.075	.044	.036	.044	.026	-.001	-.008	90.0
112.5	.212	.102		-.044	-.077	-.095	-.105	-.105	-.133	-.107	112.5
135.0	.040	-.043	-.125	-.159	-.184	-.182	-.194	-.189	-.175	-.068	135.0
157.5	-.102	-.164	-.201	-.199	-.157	-.112	-.127	-.127	-.115	-.088	157.5
180.0	-.082	-.161	-.201	-.182	-.158	-.149	-.112	-.121	-.112	-.102	180.0
202.5	-.070	-.101	-.114	-.176	-.205	-.170	-.102	-.116	-.141	-.122	202.5
225.0	-.014	-.059	-.066	-.081	-.077	-.101	-.100	-.094	-.105	-.092	225.0
247.5	-.089	-.101	-.104	-.150	-.162	-.144	-.102	-.109	-.119	-.061	247.5
270.0	-.077	-.122	-.157	-.172	-.165	-.136	-.109	-.112	-.106		270.0
292.5	-.098	-.177	-.211	-.179	-.135	-.114	-.122	-.112	-.094	-.057	292.5
315.0	.015	-.084	-.135	-.179	-.200	-.179	-.184	-.163	-.128	-.049	315.0
337.5	.196	.070	.008	-.077	-.108	-.105	-.123	-.113	-.127	-.116	337.5

TABLE 6

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
α = 0° β = 0°											
UPPER SURFACE											
.025		.163	.152	.160	.106	.120	.156	.166	.168		.025
.075		.143	.117	.099	.068	.081	.110	.123	.139	.122	.075
.125	.132	.106	.087	.078	.042	.046	.079	.100	.112	.106	.125
.175	.114	.087	.066	.062	.035	.047	.053	.065	.088	.091	.175
.225	.093	.067	.066	.033	.014	.013	.029	.048	.072	.081	.225
.275	.084	.067	.034	.012	.014	.009	.014	.038	.058	.064	.275
.325	.054	.035	.015	.002	.005	.004	.014	.019	.045	.060	.325
.375	.054	.028	.008	.007	-.006	.004	-.004	.019	.039	.050	.375
.425	.031	.020	.008	.000	.000	-.006	-.004	.004	.026	.035	.425
.475	.038	.008	-.004	-.015	-.009	-.009	-.012	-.013	.017	.023	.475
.550	.017	-.015	-.026	-.031	-.024	-.019	-.035	-.021	.007	.012	.550
.650	-.005	-.032	-.044	-.052	-.044	-.041	-.039	-.054	-.025	-.007	.650
.750	-.013	-.040	-.051	-.057	-.039	-.041	-.046	-.048	-.030	-.021	.750
.800	-.029									-.027	.800
.850		-.045	-.060	-.050	-.040	-.041	-.062	-.044	-.041		.850
.900			-.057	-.059			-.051	-.045			.900
.950					-.038	-.038					.950
LOWER SURFACE											
α = 2.5° β = 0°											
UPPER SURFACE											
.025			.040	.080	.031	.042	.062	.065	.067		.025
.075		.054	.039	.025	.007	.021	.041	.039	.047	.032	.075
.125	.055	.031	.020	.018	-.017	-.006	.021	.037	.030	.021	.125
.175	.039	.014	.005	.008	-.017	-.005	.002	.008	.015	.012	.175
.225	.028	-.001	-.008	-.017	-.031	-.034	-.022	-.007	.004	.005	.225
.275	.019	.012	-.021	-.039	-.031	-.034	-.035	-.007	-.007	-.011	.275
.325	-.017	-.018	-.038	-.052	-.037	-.033	-.031	-.034	-.015	-.003	.325
.375	-.001	-.027	-.045	-.039	-.037	-.029	-.047	-.026	-.015	-.012	.375
.425	-.033	-.038	-.037	-.050	-.038	-.042	-.047	-.034	-.026	-.025	.425
.475	-.019	-.045	-.054	-.065	-.047	-.048	-.055	-.057	-.033	-.033	.475
.550	-.034	-.065	-.079	-.077	-.057	-.058	-.073	-.057	-.042	-.040	.550
.650	-.050	-.078	-.083	-.088	-.078	-.070	-.079	-.092	-.071	-.063	.650
.750	-.059	-.085	-.091	-.097	-.070	-.079	-.077	-.088	-.070	-.069	.750
.800	-.068									-.067	.800
.850		-.085	-.100	-.078	-.071	-.065	-.090	-.080	-.081		.850
.900			-.079	-.086			-.083	-.075			.900
.950					-.064	-.051					.950
LOWER SURFACE											
.025		.279	.262	.262	.198	.219	.254	.271	.276		.025
.075		.237	.202	.183	.143	.162	.192	.208	.228	.211	.075
.125	.223	.185	.158	.156	.110	.112	.152	.177	.192	.189	.125
.175	.195	.159	.137	.132	.098	.105	.123	.137	.165	.170	.175
.225	.172	.136	.141	.107	.075	.071	.094	.116	.140	.158	.225
.275	.154	.133	.093	.078	.072	.064	.080	.105	.124	.137	.275
.325	.121	.099	.079	.065	.060	.054	.075	.079	.110	.129	.325
.375	.121	.091	.068	.065	.051	.054	.057	.079	.099	.116	.375
.425	.098	.079	.068	.058	.053	.044	.057	.066	.095	.099	.425
.475	.098	.067	.050	.039	.042	.032	.051	.050	.077	.088	.475
.550	.079	.037	.020	.026	.031	.019	.013	.039	.062	.072	.550
.650	.058	.024	.012	.002	.006	.002	.017	.007	.029	.050	.650
.750	.046	.012	.007	-.004	.006	-.005	.006	.001	.022	.033	.750
.800	.027									.023	.800
.850		.005	-.005	.004	.005	.005	-.009	.007	.007		.850
.900			.004	-.006			.000	.002			.900
.950					.008	.006					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
α = 5.0° β = 0°											
UPPER SURFACE											
.025		-.093	-.077	-.016	-.045	-.031	-.034	-.045	-.048		.025
.075		-.052	-.055	-.052	-.058	-.051	-.037	-.042	-.045	-.062	.075
.125	-.046	-.065	-.070	-.052	-.074	-.068	-.051	-.048	-.049	-.064	.125
.175	-.055	-.070	-.078	-.061	-.075	-.064	-.065	-.067	-.060	-.069	.175
.225	-.060	-.084	-.074	-.081	-.086	-.079	-.082	-.078	-.072	-.072	.225
.275	-.067	-.067	-.088	-.097	-.074	-.079	-.094	-.078	-.077	-.081	.275
.325	-.091	-.093	-.100	-.104	-.078	-.079	-.094	-.092	-.086	-.077	.325
.375	-.081	-.098	-.110	-.100	-.085	-.079	-.105	-.090	-.085	-.085	.375
.425	-.101	-.103	-.103	-.106	-.079	-.079	-.106	-.095	-.094	-.095	.425
.475	-.091	-.112	-.114	-.119	-.087	-.087	-.111	-.111	-.099	-.101	.475
.550	-.101	-.126		-.127	-.094	-.093	-.130	-.120	-.105	-.109	.550
.650	-.117	-.134	-.134	-.136	-.111	-.106	-.127	-.139	-.126	-.118	.650
.750	-.104	-.134	-.134	-.131	-.111	-.106	-.123	-.134	-.129	-.095	.750
.800	-.103									-.088	.800
.850		-.121	-.131	-.112	-.104	-.098	-.117	-.110	-.118		.850
.900			-.117	-.120			-.108	-.110			.900
.950					-.092	-.088					.950
LOWER SURFACE											
.025		.367	.353	.351	.277	.298	.342	.357	.363		.025
.075		.310	.277	.256	.210	.222	.261	.282	.301	.290	.075
.125	.296	.256	.226	.216	.172	.171	.218	.249	.258	.263	.125
.175	.266	.228	.197	.196	.158	.163	.178	.203	.230	.237	.175
.225	.237	.199	.184	.163	.127	.123	.151	.177	.205	.227	.225
.275	.224	.199	.159	.139	.123	.113	.134	.165	.182	.203	.275
.325	.192	.164	.140	.126	.108	.105	.132	.139	.167	.194	.325
.375	.189	.150	.126	.125	.096	.105	.112	.139	.156	.177	.375
.425	.160	.134	.126	.112	.099	.090	.108	.125	.144	.164	.425
.475	.160	.120	.099	.097	.080	.080	.099	.099	.129	.149	.475
.550	.130	.091	.077	.073	.070	.074	.066	.090	.115	.132	.550
.650	.113	.073	.066	.051	.045	.050	.066	.053	.076	.105	.650
.750	.091	.058	.059	.040	.045	.039	.047	.053	.070	.087	.750
.800	.080									.080	.800
.850		.059	.045	.045	.045	.050	.034	.056	.053		.850
.900			.055	.038			.040	.051			.900
.950					.045	.044					.950
α = 7.5° β = 0°											
UPPER SURFACE											
.025		-.149	-.132	-.072	-.098	-.080	-.099	-.117	-.125		.025
.075		-.106	-.103	-.108	-.110	-.091	-.098	-.104	-.116	-.135	.075
.125	-.105	-.118	-.114	-.099	-.117	-.113	-.098	-.104	-.116	-.132	.125
.175	-.108	-.118	-.114	-.100	-.116	-.099	-.119	-.117	-.119	-.131	.175
.225	-.110	-.124	-.114	-.120	-.118	-.118	-.126	-.126	-.124	-.130	.225
.275	-.112	-.104	-.126	-.134	-.110	-.118	-.133	-.126	-.129	-.133	.275
.325	-.133	-.133	-.140	-.146	-.114	-.118	-.133	-.140	-.131	-.124	.325
.375	-.123	-.133	-.140	-.132	-.114	-.112	-.143	-.127	-.128	-.129	.375
.425	-.140	-.139	-.140	-.144	-.114	-.114	-.143	-.136	-.135	-.137	.425
.475	-.127	-.144	-.147	-.153	-.114	-.116	-.143	-.152	-.137	-.143	.475
.550	-.138	-.156		-.167	-.114	-.118	-.169	-.152	-.141	-.145	.550
.650	-.138	-.169	-.165	-.167	-.123	-.127	-.164	-.174	-.162	-.138	.650
.750	-.124	-.160	-.152	-.158	-.130	-.127	-.145	-.150	-.155	-.126	.750
.800	-.130									-.124	.800
.850		-.147	-.151	-.143	-.126	-.127	-.149	-.137	-.143		.850
.900			-.144	-.153			-.139	-.136			.900
.950					-.125	-.126					.950
LOWER SURFACE											
.025		.446	.428	.413	.340	.372	.429	.440	.444		.025
.075		.385	.342	.318	.272	.282	.336	.357	.374	.369	.075
.125	.363	.328	.293	.275	.224	.219	.289	.313	.329	.337	.125
.175	.330	.291	.259	.249	.204	.208	.244	.271	.298	.309	.175
.225	.303	.265	.251	.216	.177	.183	.216	.241	.273	.295	.225
.275	.285	.255	.215	.190	.166	.176	.199	.229	.249	.271	.275
.325	.257	.225	.198	.173	.156	.163	.193	.204	.230	.255	.325
.375	.248	.212	.183	.172	.145	.157	.169	.192	.212	.239	.375
.425	.222	.193	.178	.155	.143	.144	.171	.180	.203	.219	.425
.475	.217	.179	.156	.140	.129	.136	.156	.160	.188	.203	.475
.550	.189	.144	.118	.120	.109	.127	.126	.145	.172	.183	.550
.650	.162	.125	.105	.092	.081	.105	.121	.110	.132	.162	.650
.750	.151	.113	.096	.077	.081	.096	.094	.100	.120	.141	.750
.800	.136									.134	.800
.850		.107	.085	.078	.081	.094	.081	.107	.104		.850
.900			.085	.078			.085	.100			.900
.950					.081	.094					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
					$\alpha = 10.0°$		$\beta = 0°$				
					UPPER SURFACE						
.025		-.205	-.182	-.114	-.141	-.128	-.158	-.176	-.188		.025
.075		-.161	-.149	-.150	-.141	-.128	-.152	-.165	-.173	-.192	.075
.125	-.156	-.167	-.154	-.137	-.153	-.145	-.147	-.151	-.168	-.186	.125
.175	-.156	-.167	-.156	-.137	-.141	-.134	-.156	-.160	-.170	-.181	.175
.225	-.156	-.167	-.154	-.154	-.152	-.153	-.171	-.169	-.170	-.180	.225
.275	-.156	-.147	-.164	-.169	-.140	-.153	-.176	-.167	-.171	-.184	.275
.325	-.176	-.169	-.172	-.176	-.149	-.153	-.167	-.173	-.173	-.167	.325
.375	-.164	-.169	-.171	-.164	-.149	-.151	-.185	-.173	-.165	-.170	.375
.425	-.183	-.176	-.164	-.164	-.149	-.158	-.178	-.173	-.171	-.178	.425
.475	-.166	-.176	-.179	-.180	-.149	-.163	-.178	-.187	-.175	-.179	.475
.550	-.166	-.187		-.189	-.145	-.163	-.205	-.184	-.174	-.162	.550
.650	-.163	-.183	-.187	-.189	-.145	-.150	-.198	-.194	-.180	-.164	.650
.750	-.152	-.173	-.171	-.184	-.145	-.154	-.185	-.170	-.168	-.160	.750
.800	-.159									-.159	.800
.850		-.172	-.180	-.169	-.147	-.153	-.185	-.159	-.169		.850
.900			-.164	-.176			-.176	-.159			.900
.950					-.145	-.134					.950
					LOWER SURFACE						
.025		.523	.509	.499	.435	.454	.502	.516	.518		.025
.075		.443	.415	.400	.342	.360	.404	.423	.450	.435	.075
.125	.420	.391	.363	.349	.292	.303	.354	.374	.401	.406	.125
.175	.395	.361	.323	.317	.266	.281	.308	.332	.371	.377	.175
.225	.356	.326	.303	.286	.246	.245	.278	.306	.341	.363	.225
.275	.349	.314	.279	.262	.235	.225	.262	.284	.315	.336	.275
.325	.318	.284	.255	.245	.220	.213	.246	.260	.295	.318	.325
.375	.305	.268	.245	.237	.206	.204	.227	.252	.280	.302	.375
.425	.278	.255	.233	.226	.206	.196	.216	.233	.265	.284	.425
.475	.269	.236	.212	.206	.198	.189	.204	.215	.249	.267	.475
.550	.242	.200	.166	.184	.179	.177	.182	.193	.227	.245	.550
.650	.219	.177	.170	.152	.141	.147	.170	.157	.187	.220	.650
.750	.198	.163	.152	.141	.144	.137	.137	.159	.176	.201	.750
.800	.191									.189	.800
.850		.163	.144	.137	.143	.137	.123	.161	.164		.850
.900			.144	.137			.127	.154			.900
.950					.138	.137					.950
					$\alpha = 12.5°$		$\beta = 0°$				
					UPPER SURFACE						
.025		-.250	-.236	-.154	-.186	-.176	-.206	-.232	-.232		.025
.075		-.212	-.197	-.198	-.187	-.176	-.199	-.212	-.222	-.234	.075
.125	-.210	-.225	-.207	-.177	-.191	-.186	-.185	-.193	-.215	-.229	.125
.175	-.207	-.217	-.207	-.178	-.177	-.173	-.192	-.210	-.213	-.223	.175
.225	-.205	-.217		-.194	-.187	-.190	-.204	-.210	-.212	-.218	.225
.275	-.204	-.191	-.209	-.207	-.177	-.190	-.212	-.210	-.211	-.213	.275
.325	-.210	-.216	-.209	-.212	-.185	-.190	-.199	-.218	-.210	-.190	.325
.375	-.194	-.216	-.214	-.198	-.192	-.181	-.212	-.201	-.201	-.192	.375
.425	-.209	-.216	-.201	-.198	-.180	-.194	-.212	-.209	-.206	-.196	.425
.475	-.196	-.216	-.216	-.217	-.193	-.194	-.212	-.225	-.203	-.195	.475
.550	-.196	-.216	-.229	-.225	-.179	-.193	-.232	-.219	-.194	-.190	.550
.650	-.196	-.210	-.206	-.225	-.180	-.177	-.222	-.220	-.198	-.193	.650
.750	-.187	-.210	-.198	-.214	-.178	-.177	-.206	-.198	-.190	-.193	.750
.800	-.194									-.192	.800
.850		-.210	-.210	-.196	-.176	-.177	-.214	-.187	-.195		.850
.900			-.197	-.209			-.201	-.185			.900
.950					-.176	-.168					.950
					LOWER SURFACE						
.025		.578	.572	.559	.499	.517	.569	.580	.581		.025
.075		.510	.476	.459	.407	.417	.466	.489	.516	.499	.075
.125	.479	.457	.425	.411	.346	.358	.412	.440	.468	.467	.125
.175	.456	.419	.388	.373	.318	.332	.366	.395	.432	.441	.175
.225	.423	.391	.361	.341	.294	.301	.335	.367	.402	.424	.225
.275	.407	.371	.339	.313	.287	.282	.314	.345	.376	.399	.275
.325	.378	.343	.319	.297	.273	.269	.299	.318	.356	.379	.325
.375	.364	.329	.306	.288	.262	.258	.284	.310	.337	.359	.375
.425	.335	.310	.289	.280	.258	.248	.268	.292	.320	.343	.425
.475	.325	.289	.270	.260	.244	.235	.256	.268	.304	.324	.475
.550	.295	.255	.223	.227	.224	.220	.225	.248	.281	.302	.550
.650	.271	.234	.212	.198	.185	.190	.214	.202	.237	.273	.650
.750	.258	.222	.196	.181	.185	.181	.181	.212	.230	.255	.750
.800	.248									.242	.800
.850		.214	.189	.178	.185	.181	.169	.206	.212		.850
.900			.188	.178			.172	.198			.900
.950					.181	.181					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 15.0°$ $\beta = 0°$											
UPPER SURFACE											
.025		-.283	-.269	-.189	-.225	-.228	-.256	-.279	-.276		.025
.075		-.246	-.225	-.245	-.224	-.216	-.241	-.264	-.266	-.246	.075
.125	-.236	-.259	-.244	-.224	-.223	-.224	-.223	-.244	-.260	-.246	.125
.175	-.236	-.250	-.239	-.219	-.217	-.209	-.236	-.254	-.257	-.246	.175
.225	-.236	-.250		-.231	-.225	-.226	-.248	-.252	-.253	-.244	.225
.275	-.230	-.224	-.236	-.243	-.210	-.222	-.244	-.252	-.252	-.244	.275
.325	-.244	-.243	-.243	-.243	-.224	-.222	-.237	-.268	-.246	-.230	.325
.375	-.228	-.242	-.242	-.236	-.224	-.218	-.249	-.246	-.233	-.235	.375
.425	-.245	-.241	-.224	-.236	-.222	-.230	-.243	-.258	-.231	-.243	.425
.475	-.225	-.241	-.243	-.251	-.231	-.230	-.243	-.258	-.230	-.245	.475
.550	-.225	-.239	-.225	-.250	-.231	-.230	-.257	-.241	-.222	-.240	.550
.650	-.233	-.235	-.231	-.246	-.218	-.228	-.249	-.241	-.230	-.251	.650
.750	-.233	-.232	-.230	-.243	-.190	-.205	-.239	-.228	-.224	-.247	.750
.800	-.242									-.244	.800
.850		-.232	-.248	-.231	-.190	-.193	-.245	-.222	-.230		.850
.900			-.232	-.231			-.232	-.222			.900
.950					-.190	-.193					.950
LOWER SURFACE											
.025		.627	.631	.627	.559	.623	.628	.631	.635		.025
.075		.570	.544	.525	.470	.495	.535	.555	.571	.553	.075
.125	.542	.519	.492	.472	.408	.420	.479	.498	.532	.525	.125
.175	.513	.487	.449	.421	.378	.382	.431	.459	.493	.496	.175
.225	.474	.459	.436	.395	.364	.360	.401	.430	.441	.481	.225
.275	.461	.430	.408	.371	.345	.342	.378	.404	.434	.453	.275
.325	.439	.412	.381	.355	.328	.329	.358	.379	.411	.433	.325
.375	.412	.394	.369	.341	.319	.318	.342	.358	.392	.414	.375
.425	.395	.373	.346	.328	.309	.308	.321	.335	.378	.395	.425
.475	.374	.353	.329	.312	.302	.294	.309	.319	.359	.378	.475
.550	.345	.313	.274	.277	.269	.276	.277	.299	.331	.353	.550
.650	.322	.284	.249	.242	.225	.243	.268	.248	.288	.326	.650
.750	.310	.277	.228	.225	.229	.234	.222	.265	.282	.308	.750
.800	.309									.299	.800
.850		.268	.223	.225	.229	.231	.222	.253	.264		.850
.900			.223	.225			.222	.244			.900
.950					.233	.224					.950
$\alpha = 0°$ $\beta = -5°$											
UPPER SURFACE											
.025		.174	.165	.195	.132	.074	.087	.088	.101		.025
.075		.135	.133	.127	.090	.052	.063	.069	.074	.063	.075
.125	.105	.105	.097	.088	.048	.027	.034	.039	.058	.052	.125
.175	.081	.081	.068	.062	.026	.011	.017	.020	.036	.039	.175
.225	.057	.061	.053	.040	.013	.006	-.001	.000	.021	.034	.225
.275	.048	.044	.036	.024	.005	.006	-.012	-.008	.009	.018	.275
.325	.034	.034	.024	.010	.000	-.006	-.015	-.020	.001	.011	.325
.375	.017	.024	.006	-.001	-.010	-.005	-.020	-.029	-.011	-.005	.375
.425	.008	.012	-.010	-.013	-.017	-.008	-.029	-.032	-.019	-.013	.425
.475	.001	.005	-.008	-.023	-.024	-.018	-.029	-.040	-.027	-.022	.475
.550	-.015	-.019	-.027	-.043	-.038	-.031	-.048	-.052	-.039	-.034	.550
.650	-.031	-.040	-.050	-.063	-.059	-.048	-.048	-.068	-.057	-.048	.650
.750	-.050	-.055	-.064	-.071	-.059	-.048	-.064	-.060	-.066	-.058	.750
.800	-.050									-.063	.800
.850		-.055	-.064	-.080	-.059	-.043	-.064	-.058	-.069		.850
.900			-.068	-.083			-.064	-.060			.900
.950					-.059	-.021					.950
LOWER SURFACE											

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 2.5°$ $\beta = -5°$											
UPPER SURFACE											
.025		.058	.072	.107	.064	-.014	-.009	-.019	-.013		.025
.075		.064	.062	.055	.028	-.013	-.018	-.017	-.018	-.037	.075
.125	.052	.038	.034	.036	-.008	-.023	-.027	-.017	-.026	-.042	.125
.175	.036	.019	.013	.026	-.018	-.014	-.044	-.040	-.037	-.047	.175
.225	.021	.000	.000	-.001	-.043	-.036	-.057	-.052	-.044	-.049	.225
.275	.008	.009	-.014	-.026	-.036	-.036	-.057	-.052	-.051	-.058	.275
.325	-.021	-.019	-.025	-.037	-.046	-.036	-.049	-.074	-.058	-.051	.325
.375	-.021	-.026	-.037	-.037	-.049	-.036	-.062	-.064	-.054	-.055	.375
.425	-.043	-.036	-.030	-.044	-.049	-.050	-.057	-.074	-.066	-.070	.425
.475	-.025	-.040	-.050	-.064	-.059	-.050	-.059	-.085	-.071	-.074	.475
.550	-.047	-.071		-.082	-.068	-.059	-.082	-.088	-.075	-.079	.550
.650	-.063	-.081	-.079	-.098	-.085	-.076	-.082	-.111	-.102	-.097	.650
.750	-.071	-.088	-.085	-.108	-.077	-.076	-.082	-.097	-.101	-.101	.750
.800	-.079									-.090	.800
.850		-.084	-.100	-.091	-.078	-.068	-.078	-.082	-.101		.850
.900			-.083	-.100			-.072	-.079			.900
.950					-.071	-.055					.950
LOWER SURFACE											
.025		.310	.314	.292	.228	.166	.192	.206	.208		.025
.075		.257	.243	.216	.165	.126	.145	.154	.166	.154	.075
.125	.224	.210	.198	.173	.115	.085	.112	.126	.139	.134	.125
.175	.196	.179	.166	.142	.095	.078	.084	.095	.111	.118	.175
.225	.171	.160	.142	.114	.070	.057	.068	.076	.097	.110	.225
.275	.158	.146	.127	.094	.070	.057	.059	.068	.081	.092	.275
.325	.134	.123	.106	.074	.051	.049	.051	.057	.068	.079	.325
.375	.122	.108	.096	.068	.051	.049	.044	.047	.057	.071	.375
.425	.106	.096	.088	.056	.040	.036	.036	.036	.046	.056	.425
.475	.097	.087	.068	.045	.031	.036	.027	.027	.038	.043	.475
.550	.079	.057		.019	.021	.024	.013	.015	.026	.031	.550
.650	.053	.037	.009	-.009	-.006	.006	.009	-.014	-.001	.014	.650
.750	.039	.026	.001	-.030	-.006	.005	-.005	-.009	-.010	.001	.750
.800	.025									-.004	.800
.850		.018	-.009	-.008	-.006	.005	-.009	-.006	-.016		.850
.900			-.009	-.019			-.005	-.009			.900
.950					-.005	.005					.950
$\alpha = 5.0°$ $\beta = -5°$											
UPPER SURFACE											
.025		-.040	-.024	.023	-.006	-.087	-.096	-.114	-.124		.025
.075		-.017	-.017	-.020	-.032	-.084	-.093	-.104	-.114	-.135	.075
.125	-.020	-.031	-.034	-.034	-.058	-.084	-.093	-.099	-.110	-.135	.125
.175	-.032	-.046	-.047	-.045	-.072	-.068	-.107	-.108	-.116	-.131	.175
.225	-.045	-.059	-.062	-.064	-.094	-.074	-.114	-.115	-.120	-.130	.225
.275	-.053	-.059	-.070	-.077	-.091	-.074	-.114	-.115	-.124	-.135	.275
.325	-.074	-.074	-.081	-.089	-.091	-.074	-.107	-.127	-.125	-.129	.325
.375	-.074	-.081	-.088	-.089	-.091	-.074	-.113	-.127	-.123	-.132	.375
.425	-.090	-.088	-.085	-.096	-.091	-.083	-.113	-.127	-.130	-.140	.425
.475	-.090	-.096	-.102	-.109	-.096	-.083	-.112	-.139	-.136	-.147	.475
.550	-.101	-.112		-.120	-.104	-.087	-.119	-.141	-.140	-.149	.550
.650	-.109	-.126	-.127	-.142	-.118		-.115	-.152	-.154	-.156	.650
.750	-.099	-.126	-.127	-.135	-.118	-.104	-.115	-.135	-.151	-.141	.750
.800	-.099									-.131	.800
.850		-.104	-.114	-.118	-.114	-.102	-.110	-.125	-.143		.850
.900			-.103	-.118			-.101	-.126			.900
.950					-.104	-.077					.950
LOWER SURFACE											
.025		.432	.413	.403	.317	.240	.267	.281	.290		.025
.075		.356	.329	.295	.238	.184	.206	.218	.238	.228	.075
.125	.319	.297	.273	.244	.179	.147	.173	.191	.206	.207	.125
.175	.285	.260	.233	.216	.159	.134	.144	.155	.180	.188	.175
.225	.252	.225	.210	.176	.131	.114	.121	.134	.160	.174	.225
.275	.237	.221	.187	.151	.123	.104	.113	.123	.143	.157	.275
.325	.206	.192	.165	.139	.107	.093	.104	.106	.130	.147	.325
.375	.197	.172	.148	.129	.097	.098	.091	.101	.120	.135	.375
.425	.174	.160	.148	.116	.097	.083	.087	.090	.107	.118	.425
.475	.174	.147	.125	.094	.082	.076	.076	.076	.096	.108	.475
.550	.144	.108	.120	.071	.068	.066	.057	.064	.082	.093	.550
.650	.119	.091	.065	.051	.036	.044	.057	.036	.051	.072	.650
.750	.101	.076	.053	.031	.046	.042	.036	.043	.046	.059	.750
.800	.084									.054	.800
.850		.070	.042	.044	.043	.042	.026	.045	.032		.850
.900			.046	.036			.036	.040			.900
.950					.044	.042					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/C	C_p AT WING STATION										x/C
	1	2	3	4	5	6	7	8	9	10	
	$\alpha = 7.5°$ $\beta = -5°$										
UPPER SURFACE											
.025		-.095	-.085	-.031	-.063	-.136	-.158	-.171	-.189		.025
.075		-.063	-.064	-.072	-.078	-.135	-.147	-.160	-.175	-.190	.075
.125	-.068	-.078	-.082	-.072	-.097	-.123	-.147	-.147	-.171	-.189	.125
.175	-.076	-.085	-.089	-.079	-.102	-.098	-.153	-.161	-.169	-.183	.175
.225	-.084	-.096	-.098	-.098	-.123	-.113	-.159	-.166	-.168	-.182	.225
.275	-.084	-.087	-.108	-.119	-.122	-.109	-.159	-.160	-.168	-.182	.275
.325	-.109	-.110	-.117	-.123	-.122	-.109	-.146	-.173	-.169	-.167	.325
.375	-.109	-.119	-.123	-.123	-.122	-.102	-.154	-.164	-.163	-.168	.375
.425	-.122	-.120	-.117	-.127	-.122	-.102	-.142	-.172	-.171	-.175	.425
.475	-.114	-.129	-.133	-.140	-.126	-.107	-.142	-.183	-.172	-.179	.475
.550	-.123	-.146	-.123	-.149	-.133	-.113	-.159	-.179	-.171	-.168	.550
.650	-.123	-.153	-.152	-.164	-.149	-.125	-.159	-.185	-.176	-.168	.650
.750	-.114	-.139	-.139	-.155	-.138	-.130	-.152	-.162	-.165	-.164	.750
.800	-.111									-.161	.800
.850		-.126	-.136	-.138	-.145	-.127	-.159	-.151	-.165		.850
.900			-.128	-.145			-.144	-.155			.900
.950					-.136	-.125					.950
LOWER SURFACE											
.025		.526	.514	.477	.406	.313	.353	.357	.361		.025
.075		.442	.402	.371	.305	.247	.278	.297	.305	.298	.075
.125	.404	.379	.343	.315	.245	.202	.242	.255	.272	.272	.125
.175	.366	.336	.304	.278	.218	.196	.208	.223	.244	.250	.175
.225	.329	.304	.286	.241	.192	.166	.180	.199	.222	.238	.225
.275	.315	.285	.250	.215	.179	.157	.171	.190	.202	.217	.275
.325	.281	.255	.227	.196	.166	.148	.163	.166	.185	.204	.325
.375	.267	.241	.211	.186	.153	.148	.144	.161	.176	.191	.375
.425	.243	.224	.203	.169	.153	.133	.144	.147	.162	.174	.425
.475	.234	.205	.179	.153	.138	.126	.138	.131	.147	.163	.475
.550	.207	.167	.159	.128	.121	.114	.110	.119	.132	.146	.550
.650	.179	.144	.116	.100	.085	.091	.112	.089	.099	.126	.650
.750	.159	.128	.104	.084	.091	.091	.082	.088	.096	.113	.750
.800	.140									.103	.800
.850		.118	.093	.089	.090	.087	.071	.088	.087		.850
.900			.101	.083			.080	.079			.900
.950					.085	.087					.950
	$\alpha = 10.0°$ $\beta = -5°$										
UPPER SURFACE											
.025		-.165	-.149	-.091	-.110	-.184	-.193	-.231	-.246		.025
.075		-.129	-.117	-.125	-.122	-.174	-.197	-.211	-.233	-.241	.075
.125	-.129	-.129	-.127	-.123	-.141	-.165	-.189	-.198	-.227	-.241	.125
.175	-.129	-.138	-.135	-.123	-.141	-.149	-.198	-.205	-.224	-.234	.175
.225	-.129	-.142	-.142	-.142	-.166	-.159	-.204	-.210	-.221	-.232	.225
.275	-.138	-.133	-.148	-.158	-.148	-.154	-.197	-.209	-.219	-.226	.275
.325	-.153	-.152	-.154	-.168	-.160	-.160	-.186	-.209	-.216	-.208	.325
.375	-.147	-.155	-.163	-.155	-.160	-.125	-.192	-.209	-.207	-.207	.375
.425	-.163	-.161		-.170	-.160	-.119	-.184	-.209	-.213	-.207	.425
.475	-.157	-.161	-.165	-.178	-.159	-.126	-.184	-.199	-.212	-.207	.475
.550	-.163	-.177	-.178	-.193	-.167	-.135	-.195	-.210	-.202	-.201	.550
.650	-.151	-.183	-.189	-.201	-.182	-.146	-.195	-.210	-.202	-.202	.650
.750	-.145	-.164	-.168	-.180	-.172	-.151	-.195	-.195	-.196	-.199	.750
.800	-.145									-.198	.800
.850		-.155	-.168	-.165	-.184	-.151	-.199	-.188	-.199		.850
.900			-.159	-.177			-.182	-.189			.900
.950					-.177	-.151					.950
LOWER SURFACE											
.025		.625	.604	.566	.476	.381	.427	.431	.427		.025
.075		.523	.487	.446	.376	.305	.347	.360	.372	.361	.075
.125	.484	.462	.421	.379	.301	.260	.303	.319	.339	.334	.125
.175	.447	.409	.382	.347	.275	.249	.269	.288	.311	.313	.175
.225	.411	.382	.353	.310	.247	.230	.245	.266	.285	.300	.225
.275	.394	.357	.325	.274	.230	.217	.229	.246	.263	.279	.275
.325	.358	.330	.298	.254	.217	.208	.221	.230	.246	.266	.325
.375	.344	.307	.279	.243	.205	.203	.208	.218	.235	.248	.375
.425	.319	.293	.265	.227	.205	.193	.197	.203	.219	.234	.425
.475	.305	.271	.245	.211	.191	.183	.191	.186	.208	.219	.475
.550	.274	.229	.211	.188	.176	.172	.164	.172	.188	.202	.550
.650	.246	.203	.170	.155	.136	.144	.158	.139	.158	.184	.650
.750	.224	.190	.154	.139	.134	.136	.128	.140	.152	.169	.750
.800	.205									.160	.800
.850		.180	.144	.139	.134	.136	.122	.135	.135		.850
.900			.144	.129			.127	.130			.900
.950					.126	.139					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 12.5^\circ$ $\beta = -5^\circ$											
UPPER SURFACE											
.025		-.222	-.201	-.132	-.159	-.222	-.248	-.278	-.277		.025
.075		-.186	-.163	-.177	-.160	-.211	-.242	-.260	-.266	-.256	.075
.125	-.186	-.185	-.170	-.164	-.174	-.204	-.223	-.241	-.258	-.255	.125
.175	-.182	-.185	-.178	-.164	-.174	-.184	-.241	-.247	-.255	-.252	.175
.225	-.182	-.190	-.184	-.182	-.197	-.203	-.240	-.247	-.252	-.249	.225
.275	-.182	-.177	-.186	-.193	-.183	-.203	-.235	-.243	-.250	-.247	.275
.325	-.198	-.196	-.193	-.195	-.189	-.198	-.221	-.250	-.242	-.231	.325
.375	-.186	-.197	-.193	-.193	-.189	-.165	-.231	-.243	-.229	-.231	.375
.425	-.202	-.201	-.189	-.193	-.189	-.152	-.210	-.243	-.235	-.237	.425
.475	-.191	-.201	-.201	-.207	-.189	-.152	-.220	-.252	-.232	-.238	.475
.550	-.188	-.209		-.218	-.196	-.159	-.233	-.248	-.224	-.233	.550
.650	-.180	-.199	-.203	-.227	-.207	-.176	-.233	-.246	-.233	-.246	.650
.750	-.172	-.190	-.191	-.204	-.202	-.176	-.233	-.221	-.228	-.242	.750
.800	-.183									-.241	.800
.850		-.190	-.196	-.196	-.214	-.176	-.239	-.216	-.236		.850
.900			-.189	-.207			-.224	-.214			.900
.950					-.209	-.176					.950
LOWER SURFACE											
.025		.689	.689	.648	.567	.467	.478	.485	.481		.025
.075		.597	.562	.527	.451	.380	.407	.426	.432	.417	.075
.125	.561	.533	.497	.458	.384	.332	.367	.382	.396	.394	.125
.175	.512	.489	.455	.418	.346	.316	.327	.345	.368	.373	.175
.225	.483	.451	.426	.381	.324	.274	.306	.324	.344	.359	.225
.275	.466	.426	.393	.349	.301	.273	.289	.310	.323	.337	.275
.325	.433	.397	.362	.327	.287	.261	.280	.287	.306	.325	.325
.375	.408	.376	.348	.316	.275	.253	.266	.275	.288	.307	.375
.425	.381	.357	.334	.294	.275	.240	.255	.262	.279	.292	.425
.475	.370	.337	.310	.285	.256	.230	.243	.244	.263	.277	.475
.550	.337	.294	.273	.251	.236	.218	.212	.222	.244	.256	.550
.650	.310	.262	.237	.218	.191	.184	.206	.190	.206	.229	.650
.750	.280	.247	.215	.197	.199	.184	.172	.188	.202	.221	.750
.800	.270									.213	.800
.850		.241	.205	.200	.193	.184	.172	.183	.190		.850
.900			.208	.192			.172	.181			.900
.950					.187	.184					.950
$\alpha = 15.0^\circ$ $\beta = -5^\circ$											
UPPER SURFACE											
.025		-.261	-.240	-.157	-.193	-.250	-.273	-.301	-.290		.025
.075		-.223	-.202	-.205	-.196	-.235	-.271	-.285	-.281	-.260	.075
.125	-.216	-.229	-.215	-.189	-.208	-.229	-.247	-.263	-.276	-.261	.125
.175	-.215	-.223	-.216	-.185	-.199	-.197	-.259	-.273	-.276	-.259	.175
.225	-.211	-.229	-.217	-.203	-.217	-.228	-.261	-.274	-.269	-.257	.225
.275	-.214	-.209	-.218	-.212	-.199	-.233	-.255	-.265	-.266	-.257	.275
.325	-.227	-.227	-.225	-.220	-.212	-.227	-.241	-.278	-.263	-.243	.325
.375	-.211	-.227	-.227	-.209	-.209	-.199	-.248	-.259	-.251	-.246	.375
.425	-.223	-.228	-.214	-.215	-.201	-.186	-.237	-.266	-.252	-.249	.425
.475	-.208	-.222	-.228	-.227	-.211	-.173	-.240	-.271	-.254	-.249	.475
.550	-.204	-.221		-.235	-.216	-.177	-.256	-.254	-.245	-.246	.550
.650	-.203	-.218	-.217	-.233	-.225	-.186	-.247	-.266	-.255	-.255	.650
.750	-.199	-.211	-.209	-.218	-.211	-.192	-.248	-.243	-.249	-.254	.750
.800	-.208									-.251	.800
.850		-.212	-.215	-.209	-.227	-.187	-.259	-.233	-.255		.850
.900			-.209	-.221			-.241	-.233			.900
.950					-.224	-.187					.950
LOWER SURFACE											
.025		.746	.748	.713	.637	.521	.554	.542	.532		.025
.075		.656	.625	.597	.520	.442	.489	.491	.494	.475	.075
.125	.625	.599	.567	.523	.452	.394	.440	.447	.460	.453	.125
.175	.585	.548	.511	.471	.412	.362	.403	.413	.431	.430	.175
.225	.542	.514	.487	.441	.385	.353	.380	.391	.404	.419	.225
.275	.530	.480	.457	.409	.365	.340	.361	.371	.384	.395	.275
.325	.495	.459	.427	.390	.343	.334	.343	.352	.365	.378	.325
.375	.471	.441	.414	.369	.331	.317	.334	.337	.349	.368	.375
.425	.444	.421	.387	.356	.320	.308	.313	.320	.336	.349	.425
.475	.418	.396	.375	.345	.316	.298	.304	.301	.321	.332	.475
.550	.385	.353	.310	.307	.286	.285	.273	.276	.296	.306	.550
.650	.357	.324	.291	.268	.238	.256	.267	.246	.262	.291	.650
.750	.334	.316	.263	.248	.238	.241	.231	.241	.257	.284	.750
.800	.329									.280	.800
.850		.302	.263	.247	.238	.241	.231	.237	.245		.850
.900			.259	.247			.231	.232			.900
.950					.238	.241					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 0°$ $\beta = -10°$											
UPPER SURFACE											
.025		.186	.216	.210	.183	.067	.067	.064	.073		.025
.075		.159	.154	.156	.126	.053	.052	.042	.056	.042	.075
.125	.129	.126	.118	.121	.088	.037	.037	.028	.042	.027	.125
.175	.107	.098	.095	.101	.062	.037	.024	.012	.023	.020	.175
.225	.084	.077	.080	.077	.033	.022	.005	.000	.015	.015	.225
.275	.072	.077	.065	.053	.026	.024	.005	.001	.007	.005	.275
.325	.048	.050	.052	.041	.014	.017	.006	-.018	-.001	.003	.325
.375	.048	.041	.038	.038	.005	.017	-.002	-.018	-.001	-.008	.375
.425	.020	.029		.029	.005	.006	-.002	-.017	-.012	-.018	.425
.475	.027	.022	.021	.013	-.008	.002	-.011	-.032	-.018	-.025	.475
.550	.013	-.006	.011	-.008	-.018	-.007	-.027	-.037	-.023	-.032	.550
.650	-.008	-.018	-.020	-.034	-.041	-.021	-.027	-.053	-.043	-.045	.650
.750	-.019	-.025	-.029	-.058	-.044	-.032	-.038	-.035	-.042	-.053	.750
.800	-.034									-.054	.800
.850		-.026	-.038	-.051	-.044	-.032	-.048	-.034	-.048		.850
.900			-.033	-.051			-.041	-.035			.900
.950					-.044	-.019					.950
LOWER SURFACE											
$\alpha = 2.5°$ $\beta = -10°$											
UPPER SURFACE											
.025		.115	.131	.132	.112	-.013	-.038	-.045	-.064		.025
.075		.099	.089	.084	.069	-.007	-.030	-.038	-.059	-.083	.075
.125	.078	.061	.064	.067	.030	-.013	-.019	-.030	-.062	-.087	.125
.175	.054	.045	.041	.054	.019	-.005	-.031	-.045	-.065	-.080	.175
.225	.043	.024	.026	.028	-.017	-.024	-.045	-.051	-.068	-.085	.225
.275	.026	.031	.013	-.002	-.017	-.018	-.038	-.043	-.065	-.090	.275
.325	.001	.005	-.002	-.011	-.033	-.026	-.027	-.060	-.068	-.068	.325
.375	.001	-.007	-.014	-.012	-.045	-.019	-.041	-.047	-.062	-.073	.375
.425	-.018	-.015		-.024	-.040	-.034	-.041	-.054	-.068	-.085	.425
.475	-.007	-.024	-.025	-.040	-.052	-.034	-.041	-.066	-.074	-.090	.475
.550	-.026	-.047	-.027	-.054	-.065	-.045	-.060	-.066	-.075	-.090	.550
.650	-.047	-.059	-.057	-.079	-.080	-.059	-.060	-.082	-.095	-.109	.650
.750	-.054	-.067	-.064	-.089	-.080	-.059	-.061	-.079	-.090	-.106	.750
.800	-.065									-.100	.800
.850		-.059	-.079	-.084	-.080	-.059	-.069	-.070	-.098		.850
.900			-.061	-.078			-.058	-.068			.900
.950					-.073	-.054					.950
LOWER SURFACE											
.025		.310	.329	.321	.277	.149	.163	.171	.168		.025
.075		.269	.252	.249	.208	.121	.123	.129	.133	.124	.075
.125	.225	.215	.212	.214	.152	.093	.104	.111	.109	.105	.125
.175	.197	.186	.177	.188	.132	.100	.083	.084	.089	.089	.175
.225	.171	.157	.154	.152	.099	.071	.065	.072	.076	.083	.225
.275	.162	.162	.139	.122	.086	.063	.057	.071	.064	.068	.275
.325	.130	.131	.122	.109	.073	.063	.063	.054	.055	.068	.325
.375	.130	.116	.111	.098	.063	.063	.045	.048	.051	.058	.375
.425	.100	.106		.086	.057	.048	.051	.047	.043	.044	.425
.475	.111	.096	.092	.072	.050	.041	.046	.032	.034	.037	.475
.550	.092	.065	.077	.046	.034	.033	.026	.026	.027	.023	.550
.650	.064	.051	.043	.014	.006	.019	.025	.000	.000	.008	.650
.750	.046	.041	.034	.001	.004	.011	.009	.004	-.001	-.002	.750
.800	.028									-.005	.800
.850		.034	.015	.009	.004	.014	.001	.009	-.011		.850
.900			.026	.008			.007	.002			.900
.950					.004	.025					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 5.0°$ $\beta = -10°$											
UPPER SURFACE											
.025		.024	.035	.046	.035	-.108	-.135	-.152	-.178		.025
.075		.024	.015	.009	.000	-.073	-.118	-.134	-.164	-.180	.075
.125	.007	-.006	-.007	-.004	-.033	-.060	-.099	-.113	-.157	-.178	.125
.175	-.008	-.024	-.025	-.015	-.035	-.048	-.098	-.134	-.156	-.174	.175
.225	-.025	-.040	-.039	-.041	-.065	-.058	-.104	-.134	-.153	-.172	.225
.275	-.032	-.031	-.051	-.059	-.071	-.058	-.095	-.112	-.151	-.174	.275
.325	-.058	-.058	-.059	-.071	-.085	-.058	-.082	-.135	-.150	-.149	.325
.375	-.058	-.066	-.070	-.071	-.095	-.058	-.095	-.109	-.136	-.153	.375
.425	-.078	-.076		-.071	-.087	-.065	-.082	-.117	-.143	-.164	.425
.475	-.065	-.078	-.080	-.090	-.102	-.066	-.082	-.128	-.145	-.161	.475
.550	-.077	-.103	-.080	-.104	-.113	-.074	-.099	-.119	-.143	-.158	.550
.650	-.091	-.112	-.108	-.123	-.125	-.092	-.089	-.135	-.156	-.165	.650
.750	-.087	-.108	-.108	-.128	-.125	-.092	-.097	-.123	-.142	-.161	.750
.800	-.087									-.155	.800
.850		-.089	-.108	-.104	-.108	-.092	-.106	-.110	-.147		.850
.900			-.091	-.104			-.095	-.106			.900
.950					-.108	-.074					.950
LOWER SURFACE											
.025		.431	.443	.420	.363	.215	.228	.233	.241		.025
.075		.364	.344	.327	.272	.169	.174	.181	.201	.194	.075
.125	.314	.306	.291	.279	.212	.136	.150	.157	.173	.171	.125
.175	.285	.268	.254	.245	.181	.139	.126	.129	.151	.156	.175
.225	.252	.236	.226	.209	.149	.105	.105	.112	.136	.146	.225
.275	.241	.235	.203	.176	.137	.100	.096	.108	.123	.130	.275
.325	.205	.201	.184	.156	.117	.093	.102	.089	.110	.125	.325
.375	.205	.183	.169	.150	.105	.093	.084	.093	.107	.115	.375
.425	.175	.169		.139	.105	.080	.084	.080	.096	.103	.425
.475	.181	.160	.145	.113	.089	.074	.082	.065	.087	.088	.475
.550	.152	.124	.124	.089	.073	.066	.054	.059	.076	.079	.550
.650	.121	.100	.090	.057	.039	.047	.054	.033	.048	.062	.650
.750	.098	.091	.079	.039	.039	.043	.034	.048	.050	.052	.750
.800	.082									.045	.800
.850		.083	.060	.047	.037	.041	.026	.047	.035		.850
.900			.067	.045			.037	.040			.900
.950					.041	.046					.950
$\alpha = 7.5°$ $\beta = -10°$											
UPPER SURFACE											
.025		-.041	-.026	-.015	-.024	-.154	-.183	-.206	-.227		.025
.075		-.027	-.034	-.041	-.046	-.124	-.169	-.184	-.212	-.219	.075
.125	-.030	-.051	-.054	-.044	-.069	-.118	-.142	-.162	-.208	-.219	.125
.175	-.045	-.067	-.067	-.052	-.082	-.085	-.143	-.176	-.205	-.216	.175
.225	-.059	-.078	-.079	-.076	-.104	-.070	-.155	-.175	-.199	-.214	.225
.275	-.065	-.066	-.090	-.097	-.104	-.079	-.148	-.162	-.197	-.214	.275
.325	-.093	-.096	-.098	-.108	-.113	-.079	-.129	-.174	-.196	-.188	.325
.375	-.087	-.105	-.108	-.093	-.125	-.079	-.145	-.154	-.178	-.190	.375
.425	-.109	-.109		-.106	-.115	-.090	-.129	-.154	-.187	-.199	.425
.475	-.097	-.117	-.116	-.124	-.131	-.090	-.112	-.171	-.185	-.196	.475
.550	-.109	-.136	-.116	-.135	-.142	-.096	-.111	-.157	-.182	-.187	.550
.650	-.118	-.139	-.135	-.154	-.149	-.110	-.111	-.171	-.200	-.195	.650
.750	-.100	-.122	-.124	-.139	-.141	-.110	-.121	-.170	-.183	-.191	.750
.800	-.103									-.189	.800
.850		-.111	-.124	-.123	-.137	-.110	-.129	-.152	-.189		.850
.900			-.111	-.121			-.117	-.143			.900
.950					-.132	-.110					.950
LOWER SURFACE											
.025		.561	.553	.510	.444	.269	.291	.292	.303		.025
.075		.463	.432	.402	.336	.215	.232	.238	.259	.251	.075
.125	.414	.393	.367	.344	.269	.183	.204	.212	.231	.229	.125
.175	.376	.349	.324	.304	.236	.183	.178	.182	.209	.212	.175
.225	.341	.316	.296	.269	.201	.154	.156	.165	.189	.203	.225
.275	.326	.305	.270	.233	.188	.144	.147	.156	.173	.186	.275
.325	.291	.269	.245	.214	.168	.139	.146	.140	.165	.176	.325
.375	.278	.252	.230	.204	.155	.137	.131	.137	.156	.169	.375
.425	.255	.238		.191	.149	.127	.125	.128	.143	.152	.425
.475	.247	.221	.195	.165	.134	.118	.121	.110	.136	.141	.475
.550	.219	.180	.174	.136	.117	.109	.097	.101	.121	.126	.550
.650	.179	.157	.133	.101	.078	.086	.097	.071	.092	.107	.650
.750	.157	.146	.121	.086	.080	.083	.072	.084	.088	.097	.750
.800	.136									.093	.800
.850		.133	.104	.090	.083	.078	.065	.082	.078		.850
.900			.109	.086			.073	.079			.900
.950					.079	.088					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 10.0°$ $\beta = -10°$											
UPPER SURFACE											
.025		-.123	-.103	-.079	-.076	-.208	-.240	-.265	-.260		.025
.075		-.091	-.092	-.103	-.095	-.171	-.227	-.246	-.252	-.257	.075
.125	-.086	-.106	-.108	-.100	-.118		-.190		-.250	-.257	.125
.175	-.096	-.116	-.118	-.100	-.118	-.128	-.195	-.230	-.246	-.254	.175
.225	-.105	-.128	-.128	-.125	-.142	-.099	-.204	-.232	-.242	-.252	.225
.275	-.113	-.112	-.136	-.142	-.142	-.111	-.201	-.210	-.241	-.252	.275
.325	-.136	-.141	-.142	-.154	-.157	-.111	-.178	-.221	-.237	-.227	.325
.375	-.130	-.143	-.150	-.138	-.157	-.111	-.201	-.200	-.220	-.227	.375
.425	-.147	-.151		-.145	-.157	-.118	-.181	-.206	-.228	-.234	.425
.475	-.135	-.151	-.157	-.158	-.165	-.118	-.180	-.217	-.227	-.227	.475
.550	-.148	-.168	-.157	-.171	-.171	-.126	-.177	-.206	-.220	-.221	.550
.650	-.143	-.168	-.165	-.182	-.181	-.136	-.137	-.222	-.237	-.230	.650
.750	-.130	-.149	-.149	-.164	-.164	-.143	-.145	-.204	-.221	-.230	.750
.800	-.141									-.228	.800
.850		-.147	-.156	-.151	-.165	-.142	-.155	-.188	-.228		.850
.900			-.147	-.151			-.139	-.183			.900
.950					-.167	-.142					.950
LOWER SURFACE											
.025		.672	.664	.613	.541	.347	.360	.357	.351		.025
.075		.565	.526	.497	.419	.286	.299	.305	.313	.302	.075
.125	.525	.490	.455	.428	.344	.250	.269	.277	.288	.278	.125
.175	.481	.441	.410	.386	.310	.244	.245	.247	.261	.261	.175
.225	.440	.406	.377	.344	.274	.213	.222	.226	.242	.253	.225
.275	.420	.389	.347	.311	.257	.203	.208	.221	.225	.235	.275
.325	.387	.351	.317	.287	.237	.196	.203	.200	.214	.226	.325
.375	.369	.330	.300	.277	.222	.194	.190	.198	.205	.214	.375
.425	.339	.311		.259	.220	.179	.186	.185	.189	.197	.425
.475	.332	.295	.264	.235	.201	.170	.177	.168	.181	.188	.475
.550	.295	.245	.240	.203	.179	.157	.148	.156	.167	.169	.550
.650	.258	.224	.196	.162	.140	.138	.144	.123	.134	.150	.650
.750	.224	.208	.180	.149	.144	.137	.127	.126	.133	.139	.750
.800	.202									.136	.800
.850		.200	.170	.159	.144	.134	.121	.127	.121		.850
.900			.173	.151			.125	.124			.900
.950					.141	.134					.950
$\alpha = 12.5°$ $\beta = -10°$											
UPPER SURFACE											
.025		-.182	-.157	-.139	-.134	-.246	-.270	-.307	-.284		.025
.075		-.144	-.140	-.153	-.146	-.223	-.270	-.291	-.281	-.276	.075
.125	-.131	-.155	-.153	-.146	-.162		-.241	-.263	-.278	-.277	.125
.175	-.139	-.159	-.160	-.146	-.168	-.181	-.249	-.276	-.276	-.273	.175
.225	-.150	-.166	-.166	-.163	-.189		-.263	-.276	-.272	-.273	.225
.275	-.150	-.149	-.172	-.183	-.180	-.135	-.255	-.257	-.270	-.272	.275
.325	-.170	-.173	-.178	-.192	-.195	-.140	-.237	-.275	-.269	-.249	.325
.375	-.165	-.179	-.185	-.180	-.195	-.120	-.249	-.249	-.250	-.253	.375
.425	-.179	-.182		-.183	-.194	-.152	-.223	-.257	-.256	-.257	.425
.475	-.169	-.182	-.188	-.199	-.202	-.153	-.217	-.265	-.257	-.256	.475
.550	-.168	-.197	-.188	-.212	-.202	-.154	-.213	-.250	-.248	-.251	.550
.650	-.167	-.193	-.195	-.212	-.212	-.167	-.188	-.266	-.260	-.264	.650
.750	-.160	-.181	-.181	-.201	-.206	-.167	-.188	-.242	-.249	-.262	.750
.800	-.166									-.259	.800
.850		-.179	-.192	-.186	-.206	-.150	-.195	-.217	-.257		.850
.900			-.178	-.185			-.182	-.197			.900
.950					-.206	-.170					.950
LOWER SURFACE											
.025		.768	.757	.701	.627	.404	.418	.411	.403		.025
.075		.655	.612	.573	.491	.345	.363	.363	.373	.354	.075
.125	.615	.579	.543	.498	.411	.307	.326	.336	.348	.336	.125
.175	.570	.526	.489	.450	.370	.292	.300	.305	.323	.319	.175
.225	.525	.488	.454	.413	.337	.271	.280	.286	.303	.310	.225
.275	.504	.463	.420	.377	.317	.260	.265	.277	.288	.293	.275
.325	.466	.435	.391	.353	.299	.252	.262	.257	.274	.283	.325
.375	.448	.407	.370	.336	.284	.248	.247	.252	.260	.269	.375
.425	.416	.385		.317	.273	.236	.238	.236	.248	.256	.425
.475	.403	.368	.332	.294	.255	.225	.226	.219	.238	.240	.475
.550	.366	.319	.298	.265	.238	.210	.199	.201	.219	.221	.550
.650	.327	.286	.249	.221	.190	.192	.194	.171	.186	.206	.650
.750	.285	.275	.235	.202	.193	.187	.169	.179	.186	.202	.750
.800	.271									.199	.800
.850		.264	.222	.206	.192	.184	.169	.183	.175		.850
.900			.226	.197			.173	.178			.900
.950					.188	.182					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 15.0°$ $\beta = -10°$											
UPPER SURFACE											
.025		-.229	-.202	-.177	-.168	-.265	-.280	-.311	-.292		.025
.075		-.190	-.177	-.192	-.180	-.253	-.286	-.304	-.291	-.279	.075
.125	-.186	-.201	-.188	-.173	-.194	-.252	-.253	-.291	-.288	-.283	.125
.175	-.184	-.201	-.192	-.174	-.194	-.213	-.272	-.300	-.286	-.280	.175
.225	-.183	-.201	-.199	-.190	-.208	-.223	-.289	-.299	-.286	-.281	.225
.275	-.182	-.182	-.199	-.206	-.202	-.195	-.280	-.285	-.283	-.280	.275
.325	-.205	-.202	-.208	-.215	-.214	-.175	-.265	-.292	-.280	-.263	.325
.375	-.192	-.207	-.208	-.205	-.214	-.160	-.272	-.285	-.266	-.263	.375
.425	-.214	-.215		-.203	-.214	-.170	-.246	-.285	-.272	-.269	.425
.475	-.194	-.215	-.212	-.220	-.220	-.170	-.247	-.290	-.269	-.271	.475
.550	-.194	-.222	-.213	-.232	-.221	-.170	-.255	-.271	-.264	-.265	.550
.650	-.189	-.212	-.208	-.223	-.221	-.182	-.233	-.293	-.276	-.276	.650
.750	-.183	-.202	-.201	-.212	-.221	-.187	-.223	-.266	-.267	-.277	.750
.800	-.193									-.275	.800
.850		-.202	-.210	-.203	-.221	-.187	-.223	-.244	-.274		.850
.900			-.195	-.205			-.206	-.229			.900
.950					-.221	-.187					.950
LOWER SURFACE											
.025		.832	.823	.775	.701	.460	.469	.451	.444		.025
.075		.721	.683	.644	.558	.395	.413	.413	.420	.399	.075
.125	.680	.650	.608	.565	.478	.365	.386	.380	.398	.382	.125
.175	.631	.598	.555	.504	.431	.335	.354	.356	.377	.368	.175
.225	.587	.556	.517	.470	.400	.326	.335	.337	.356	.360	.225
.275	.568	.516	.483	.438	.375	.314	.323	.323	.341	.343	.275
.325	.536	.494	.451	.413	.352	.305	.308	.310	.323	.328	.325
.375	.507	.472	.430	.391	.341	.296	.295	.300	.312	.317	.375
.425	.479	.453		.372	.324	.288	.278	.281	.298	.302	.425
.475	.456	.425	.385	.354	.321	.275	.270	.265	.286	.287	.475
.550	.418	.375	.346	.323	.292	.258	.245	.240	.264	.268	.550
.650	.374	.343	.300	.275	.240	.237	.240	.218	.240	.261	.650
.750	.334	.324	.275	.256	.238	.232	.214	.231	.237	.253	.750
.800	.322									.252	.800
.850		.313	.278	.244	.240	.231	.212	.231	.227		.850
.900			.270	.246			.212	.224			.900
.950					.237	.225					.950
$\alpha = 0°$ $\beta = 15°$											
UPPER SURFACE											
.025		.205	.212	.220	.207	.050	.044	.038	.035		.025
.075		.160	.153	.155	.147	.041	.030	.025	.024	.002	.075
.125	.128	.116	.115	.129	.100	.027	.021	.021	.015	-.006	.125
.175	.108	.090	.090	.111	.077	.037	.009	.002	.004	-.010	.175
.225	.087	.069	.071	.082	.046	.018	.005	-.004	-.001	-.013	.225
.275	.075	.073	.056	.056	.034	.012	-.001	.000	-.007	-.021	.275
.325	.047	.041	.044	.045	.018	.005	.008	-.013	-.012	-.013	.325
.375	.049	.033	.030	.047	.002	.008	-.007	-.008	-.010	-.018	.375
.425	.023	.023		.036	.005	.000	-.004	-.018	-.015	-.027	.425
.475	.027	.013	.015	.015	-.014	-.007	-.006	-.028	-.022	-.035	.475
.550	.007	-.008	.009	-.005	-.030	-.018	-.031	-.030	-.026	-.038	.550
.650	-.017	-.023	-.017	-.028	-.058	-.033	-.032	-.051	-.044	-.052	.650
.750	-.026	-.032	-.027	-.044	-.062	-.033	-.037	-.037	-.043	-.057	.750
.800	-.039									-.058	.800
.850		-.037	-.040	-.040	-.057	-.033	-.045	-.037	-.052		.850
.900			-.030	-.050			-.037	-.035			.900
.950					-.044	-.027					.950
LOWER SURFACE											

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 2.5°$ $\beta = -15°$											
UPPER SURFACE											
.025		.114	.124	.128	.122	-.038	-.081	-.114	-.132		.025
.075		.083	.083	.076	.068	-.013	-.053	-.091	-.126	-.148	.075
.125	.059	.045	.045	.056	.033	-.021	-.044	-.064	-.122	-.146	.125
.175	.037	.021	.021	.041	.017	-.013	-.046	-.076	-.118	-.142	.175
.225	.021	.004	.005	.015	-.009	-.026	-.047	-.088	-.109	-.138	.225
.275	.011	.008	-.008	-.005	-.020	-.031	-.049	-.062	-.091	-.141	.275
.325	-.014	-.017	-.020	-.019	-.037	-.034	-.039	-.069	-.070	-.113	.325
.375	-.012	-.027	-.030	-.019	-.049	-.032	-.052	-.053	-.056	-.117	.375
.425	-.038	-.038		-.026	-.047	-.038	-.050	-.059	-.063	-.124	.425
.475	-.028	-.045	-.044	-.047	-.062	-.045	-.052	-.072	-.067	-.126	.475
.550	-.046	-.064	-.046	-.059	-.083	-.053	-.069	-.072	-.068	-.120	.550
.650	-.064	-.079	-.071	-.083	-.103	-.069	-.070	-.089	-.086	-.124	.650
.750	-.072	-.085	-.079	-.095	-.104	-.072	-.075	-.073	-.082	-.115	.750
.800	-.078									-.104	.800
.850		-.071	-.090	-.084	-.085	-.071	-.081	-.069	-.091		.850
.900			-.066	-.082			-.071	-.068			.900
.950					-.079	-.063					.950
LOWER SURFACE											
.025		.297	.310	.314	.294	.122	.126	.126	.134		.025
.075		.240	.236	.243	.225	.097	.091	.091	.105	.098	.075
.125	.204	.192	.194	.204	.167	.079	.073	.078	.087	.082	.125
.175	.176	.162	.165	.181	.140	.081	.060	.058	.071	.068	.175
.225	.154	.139	.144	.152	.107	.057	.049	.046	.062	.065	.225
.275	.143	.137	.127	.127	.092	.052	.043	.045	.056	.051	.275
.325	.114	.107	.109	.111	.069	.045	.045	.033	.046	.050	.325
.375	.107	.097	.098	.107	.059	.044	.033	.037	.043	.044	.375
.425	.082	.086		.096	.056	.036	.032	.024	.032	.032	.425
.475	.085	.077	.086	.076	.037	.030	.031	.013	.026	.023	.475
.550	.062	.050	.064	.052	.022	.021	.007	.006	.018	.016	.550
.650	.031	.028	.039	.018	-.012	.001	.012	-.012	-.001	.001	.650
.750	.021	.024	.028	.000	-.017	.001	-.004	.007	-.001	-.004	.750
.800	.005									-.009	.800
.850		.020	.015	.006	-.015	.000	-.009	.005	-.007		.850
.900			.021	-.001			-.001	.003			.900
.950					-.013	.004					.950
$\alpha = 5.0°$ $\beta = -15°$											
UPPER SURFACE											
.025		.045	.056	.068	.062	-.102	-.153	-.188	-.207		.025
.075		.028	.031	.020	.015	-.086	-.149	-.179	-.201	-.215	.075
.125	.011	-.004	-.005	.002	-.015	-.045	-.139	-.148	-.202	-.214	.125
.175	-.007	-.025	-.026	-.005	-.026	-.038	-.118	-.169	-.195	-.207	.175
.225	-.023	-.043	-.039	-.031	-.051	-.058	-.078	-.166	-.190	-.209	.225
.275	-.033	-.034	-.053	-.052	-.059	-.058	-.073	-.139	-.184	-.208	.275
.325	-.058	-.059	-.066	-.062	-.073	-.059	-.068	-.133	-.174	-.180	.325
.375	-.053	-.069	-.075	-.060	-.083	-.058	-.082	-.082	-.153	-.182	.375
.425	-.078	-.078		-.066	-.082	-.065	-.076	-.091	-.157	-.190	.425
.475	-.070	-.084	-.081	-.084	-.101	-.072	-.079	-.105	-.156	-.187	.475
.550	-.084	-.099	-.086	-.101	-.110	-.079	-.096	-.103	-.140	-.179	.550
.650	-.098	-.113	-.105	-.116	-.134	-.094	-.094	-.118	-.118	-.189	.650
.750	-.096	-.109	-.108	-.126	-.126	-.096	-.102	-.099	-.109	-.184	.750
.800	-.092									-.180	.800
.850		-.089	-.099	-.099	-.111	-.098	-.111	-.095	-.117		.850
.900			-.088	-.098			-.095	-.092			.900
.950					-.108	-.091					.950
LOWER SURFACE											
.025		.402	.421	.417	.390	.181	.192	.188	.204		.025
.075		.331	.327	.329	.295	.146	.149	.152	.167	.157	.075
.125	.288	.278	.278	.286	.233	.122	.132	.133	.149	.140	.125
.175	.257	.244	.245	.251	.194	.123	.113	.113	.133	.129	.175
.225	.229	.217	.223	.222	.164	.103	.097	.100	.118	.121	.225
.275	.211	.211	.200	.190	.149	.096	.090	.092	.109	.108	.275
.325	.184	.184	.181	.175	.124	.088	.091	.079	.099	.103	.325
.375	.175	.168	.167	.164	.107	.088	.079	.085	.095	.095	.375
.425	.150	.156		.149	.103	.077	.078	.073	.086	.084	.425
.475	.150	.147	.146	.128	.088	.070	.072	.058	.077	.075	.475
.550	.121	.116	.127	.101	.065	.059	.050	.049	.067	.062	.550
.650	.086	.091	.094	.063	.033	.044	.050	.023	.043	.049	.650
.750	.070	.083	.077	.046	.027	.043	.037	.042	.043	.040	.750
.800	.060									.036	.800
.850		.077	.065	.047	.027	.038	.030	.047	.034		.850
.900			.072	.040			.036	.045			.900
.950					.028	.040					.950

TABLE 6, Continued

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	C_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 7.5°$ $\beta = -15°$											
UPPER SURFACE											
.025		-.018	-.011	-.002	.000	-.161	-.209	-.248	-.240		.025
.075		-.025	-.023	-.037	-.038	-.160	-.217	-.248	-.240	-.251	.075
.125	-.036	-.053	-.055	-.047	-.065	-.092	-.199	-.222	-.240	-.251	.125
.175	-.055	-.070	-.071	-.057	-.073	-.070	-.216	-.235	-.240	-.247	.175
.225	-.065	-.085	-.084	-.081	-.097	-.088	-.173	-.244	-.235	-.248	.225
.275	-.073	-.075	-.094	-.101	-.098	-.088	-.143	-.206	-.234	-.247	.275
.325	-.097	-.101	-.104	-.111	-.115	-.089	-.107	-.212	-.230	-.221	.325
.375	-.096	-.110	-.113	-.103	-.122	-.088	-.109	-.174	-.207	-.219	.375
.425	-.116	-.116		-.110	-.120	-.096	-.103	-.164	-.213	-.226	.425
.475	-.107	-.122	-.120	-.126	-.136	-.100	-.104	-.152	-.213	-.226	.475
.550	-.117	-.136	-.127	-.141	-.146	-.104	-.123	-.140	-.201	-.216	.550
.650	-.127	-.146	-.136	-.155	-.165	-.118	-.117	-.155	-.212	-.228	.650
.750	-.117	-.124	-.130	-.142	-.146	-.118	-.126	-.131	-.193	-.228	.750
.800	-.114									-.225	.800
.850		-.115	-.127	-.127	-.143	-.120	-.132	-.124	-.193		.850
.900			-.117	-.128			-.122	-.122			.900
.950					-.140	-.120					.950
LOWER SURFACE											
.025		.542	.558	.535	.485	.237	.247	.243	.255		.025
.075		.450	.436	.425	.373	.197	.206	.205	.223	.207	.075
.125	.396	.388	.378	.367	.297	.171	.179	.186	.204	.190	.125
.175	.357	.341	.335	.327	.256	.173	.160	.159	.184	.177	.175
.225	.325	.311	.305	.288	.218	.151	.148	.152	.168	.170	.225
.275	.308	.302	.284	.261	.204	.141	.137	.142	.157	.154	.275
.325	.271	.271	.260	.236	.183	.134	.141	.132	.147	.149	.325
.375	.264	.254	.247	.224	.167	.133	.127	.121	.140	.140	.375
.425	.236	.239		.210	.159	.124	.121	.115	.128	.128	.425
.475	.232	.224	.216	.186	.142	.114	.114	.103	.120	.118	.475
.550	.201	.187	.191	.156	.123	.105	.090	.089	.107	.104	.550
.650	.160	.164	.149	.119	.079	.089	.090	.066	.081	.087	.650
.750	.135	.152	.133	.095	.073	.085	.077	.082	.082	.082	.750
.800	.123									.080	.800
.850		.148	.116	.098	.076	.083	.068	.088	.074		.850
.900			.123	.090			.075	.084			.900
.950					.076	.081					.950
$\alpha = 10.0°$ $\beta = -15°$											
UPPER SURFACE											
.025		-.085	-.076	-.062	-.057	-.209	-.248	-.281	-.263		.025
.075		-.075	-.066	-.088	-.088	-.223	-.263	-.287	-.262	-.260	.075
.125	-.081	-.096	-.103	-.095	-.111	-.161	-.238	-.268	-.262	-.263	.125
.175	-.092	-.113	-.114	-.101	-.116	-.108	-.263	-.277	-.262	-.260	.175
.225	-.104	-.122	-.121	-.123	-.141	-.120	-.258	-.290	-.259	-.260	.225
.275	-.113	-.114	-.133	-.141	-.137	-.116	-.223	-.261	-.260	-.258	.275
.325	-.136	-.139	-.142	-.148	-.154	-.119	-.200	-.250	-.256	-.240	.325
.375	-.128	-.145	-.152	-.143	-.159	-.116	-.184	-.261	-.238	-.240	.375
.425	-.149	-.152		-.146	-.153	-.126	-.146	-.254	-.244	-.245	.425
.475	-.141	-.156	-.159	-.164	-.169	-.129	-.137	-.233	-.246	-.246	.475
.550	-.152	-.171	-.160	-.175	-.180	-.133	-.147	-.192	-.235	-.240	.550
.650	-.146	-.166	-.166	-.185	-.187	-.143	-.147	-.200	-.249	-.252	.650
.750	-.134	-.149	-.152	-.165	-.172	-.147	-.152	-.169	-.231	-.249	.750
.800	-.137									-.249	.800
.850		-.145	-.160	-.151	-.171	-.149	-.158	-.168	-.221		.850
.900			-.148	-.159			-.147	-.170			.900
.950					-.167	-.145					.950
LOWER SURFACE											
.025		.689	.692	.647	.578	.292	.300	.290	.299		.025
.075		.569	.544	.523	.449	.252	.260	.258	.274	.259	.075
.125	.511	.495	.471	.453	.369	.226	.239	.237	.256	.242	.125
.175	.468	.441	.423	.402	.322	.222	.213	.216	.237	.231	.175
.225	.425	.411	.389	.361	.290	.200	.199	.203	.221	.222	.225
.275	.408	.391	.361	.331	.268	.197	.192	.196	.208	.210	.275
.325	.375	.358	.335	.307	.248	.188	.191	.185	.197	.202	.325
.375	.360	.339	.316	.292	.229	.183	.174	.175	.190	.192	.375
.425	.334	.322		.270	.219	.174	.167	.166	.177	.179	.425
.475	.318	.303	.284	.247	.203	.165	.165	.149	.170	.168	.475
.550	.276	.262	.252	.218	.179	.149	.136	.134	.154	.154	.550
.650	.231	.232	.207	.172	.136	.134	.135	.114	.131	.141	.650
.750	.199	.218	.188	.151	.126	.133	.123	.132	.131	.137	.750
.800	.188									.135	.800
.850		.210	.180	.153	.122	.133	.115	.133	.125		.850
.900			.179	.143			.117	.133			.900
.950					.126	.127					.950

TABLE 6. Concluded

PRESSURE COEFFICIENT DATA FOR WING ALONE

x/c	c_p AT WING STATION										x/c
	1	2	3	4	5	6	7	8	9	10	
$\alpha = 12.5°$ $\beta = -15°$											
UPPER SURFACE											
.025		-.147	-.136	-.108	-.104	-.235	-.266	-.294	-.274		.025
.075		-.119	-.110	-.128	-.124	-.248	-.286	-.296	-.277	-.257	.075
.125	-.115	-.134	-.136	-.128	-.142	-.228	-.257	-.287	-.277	-.261	.125
.175	-.127	-.142	-.148	-.130	-.149	-.165	-.283	-.292	-.274	-.258	.175
.225	-.134	-.154	-.158	-.151	-.171	-.130	-.280	-.305	-.274	-.254	.225
.275	-.142	-.143	-.165	-.168	-.162	-.138	-.268	-.282	-.273	-.255	.275
.325	-.164	-.168	-.173	-.175	-.180	-.140	-.250	-.261	-.269	-.243	.325
.375	-.157	-.173	-.179	-.170	-.185	-.139	-.236	-.274	-.255	-.245	.375
.425	-.177	-.178		-.171	-.177	-.146	-.200	-.270	-.262	-.248	.425
.475	-.167	-.181	-.183	-.189	-.192	-.147	-.209	-.275	-.264	-.248	.475
.550	-.173	-.196	-.184	-.202	-.204	-.153	-.206	-.237	-.253	-.243	.550
.650	-.161	-.183	-.185	-.204	-.199	-.162	-.178	-.261	-.265	-.247	.650
.750	-.153	-.170	-.171	-.181	-.191	-.166	-.164	-.228	-.252	-.248	.750
.800	-.160									-.246	.800
.850		-.16	-.181	-.171	-.190	-.171	-.164	-.210	-.248		.850
.900			-.170	-.178			-.158	-.206			.900
.950					-.186	-.167					.950
LOWER SURFACE											
.025		.817	.805	.751	.672	.348	.348	.337	.330		.025
.075		.680	.644	.609	.524	.306	.320	.308	.314	.292	.075
.125	.627	.600	.563	.531	.439	.284	.292	.290	.298	.277	.125
.175	.577	.544	.508	.475	.391	.267	.274	.268	.281	.270	.175
.225	.531	.508	.474	.436	.356	.256	.254	.257	.264	.263	.225
.275	.511	.474	.441	.401	.335	.249	.251	.245	.254	.249	.275
.325	.478	.443	.414	.375	.311	.241	.245	.237	.243	.244	.325
.375	.455	.422	.392	.353	.293	.235	.230	.232	.235	.230	.375
.425	.427	.401		.337	.279	.224	.220	.216	.221	.223	.425
.475	.402	.378	.351	.313	.260	.215	.213	.205	.212	.210	.475
.550	.357	.331	.314	.275	.242	.204	.190	.185	.194	.199	.550
.650	.302	.299	.263	.230	.191	.187	.191	.162	.175	.190	.650
.750	.267	.288	.244	.206	.191	.181	.175	.179	.175	.191	.750
.800	.255									.186	.800
.850		.273	.236	.209	.188	.181	.166	.180	.168		.850
.900			.233	.201			.166	.174			.900
.950					.185	.180					.950
$\alpha = 15.0°$ $\beta = -15°$											
UPPER SURFACE											
.025		-.215	-.199	-.171	-.153	-.273	-.289	-.308	-.284		.025
.075		-.180	-.154	-.174	-.167	-.275	-.304	-.308	-.285	-.258	.075
.125	-.171	-.186	-.181	-.174	-.186	-.266	-.277	-.305	-.288	-.261	.125
.175	-.173	-.187	-.191	-.174	-.186	-.223	-.301	-.306	-.288	-.260	.175
.225	-.180	-.196	-.196	-.190	-.204	-.213	-.287	-.306	-.287	-.258	.225
.275	-.179	-.179	-.199	-.206	-.194	-.159	-.299	-.302	-.286	-.258	.275
.325	-.198	-.202	-.209	-.209	-.216	-.157	-.289	-.272	-.286	-.248	.325
.375	-.191	-.206	-.213	-.206	-.217	-.158	-.281	-.285	-.277	-.249	.375
.425	-.207	-.209		-.207	-.215	-.166	-.247	-.277	-.279	-.254	.425
.475	-.199	-.213	-.216	-.223	-.224	-.171	-.263	-.300	-.281	-.251	.475
.550	-.196	-.223	-.215	-.231	-.232	-.172	-.264	-.279	-.274	-.249	.550
.650	-.186	-.210	-.212	-.228	-.228	-.183	-.236	-.304	-.280	-.258	.650
.750	-.183	-.202	-.204	-.216	-.224	-.184	-.191	-.269	-.272	-.258	.750
.800	-.187									-.254	.800
.850		-.198	-.212	-.206	-.223	-.184	-.181	-.255	-.271		.850
.900			-.202	-.209			-.178	-.252			.900
.950					-.221	-.184					.950
LOWER SURFACE											
.025		.914	.901	.843	.762	.401	.396	.373	.364		.025
.075		.780	.730	.696	.606	.366	.371	.354	.359	.336	.075
.125	.729	.696	.652	.608	.514	.341	.348	.337	.346	.326	.125
.175	.677	.638	.587	.546	.460	.324	.326	.321	.328	.316	.175
.225	.627	.595	.558	.504	.427	.315	.311	.306	.316	.312	.225
.275	.608	.557	.517	.475	.402	.305	.297	.296	.304	.299	.275
.325	.571	.529	.488	.444	.382	.300	.289	.290	.291	.291	.325
.375	.538	.507	.465	.417	.358	.293	.282	.276	.279	.279	.375
.425	.505	.482		.398	.340	.280	.271	.262	.274	.271	.425
.475	.476	.460	.418	.376	.328	.270	.263	.248	.264	.258	.475
.550	.431	.407	.375	.339	.307	.260	.242	.225	.244	.246	.550
.650	.372	.372	.322	.290	.251	.243	.245	.215	.230	.241	.650
.750	.332	.353	.299	.273	.251	.236	.215	.231	.229	.239	.750
.800	.320									.237	.800
.850		.343	.301	.270	.252	.233	.216	.230	.224		.850
.900			.295	.268			.212	.227			.900
.950					.247	.233					.950

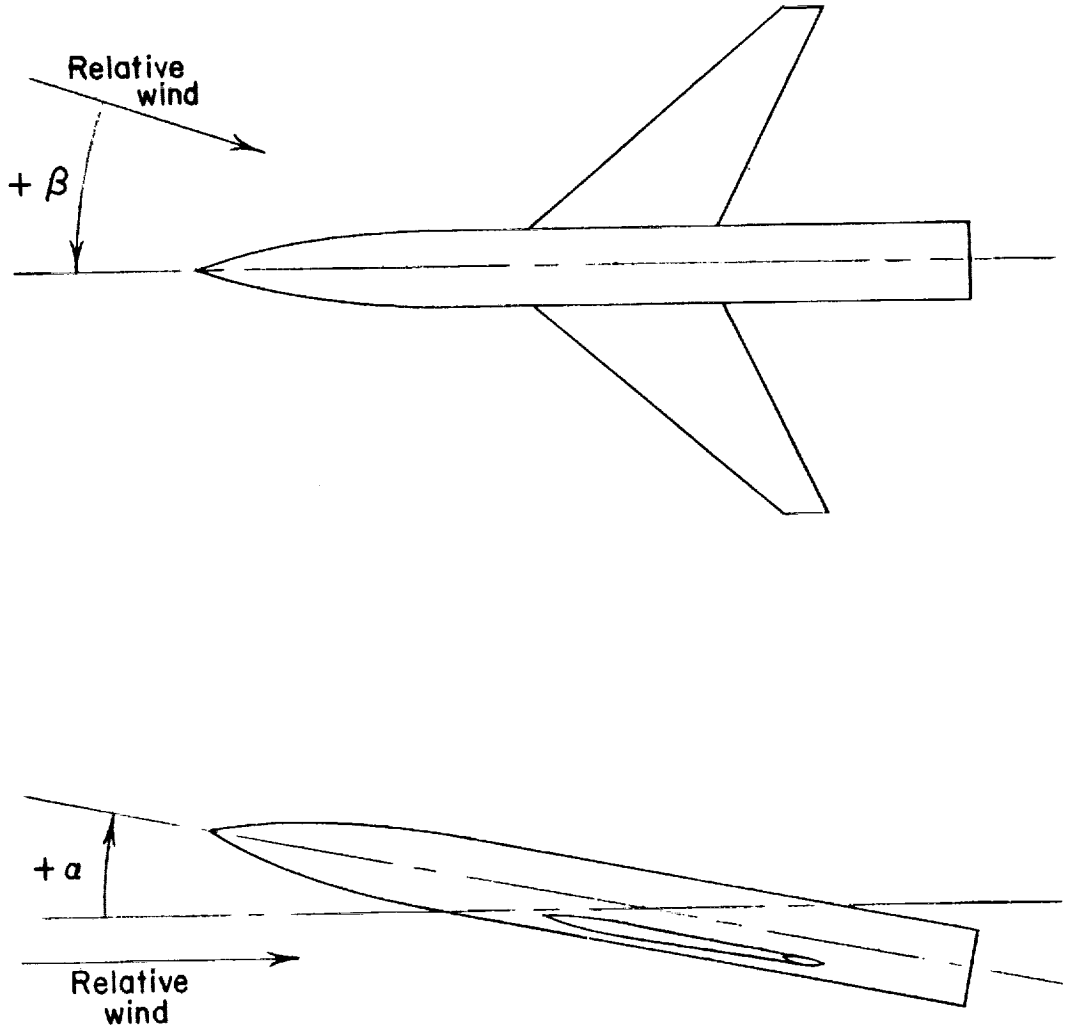

Figure 1.- Identification of positive directions in angle of attack α and angle of sideslip β.

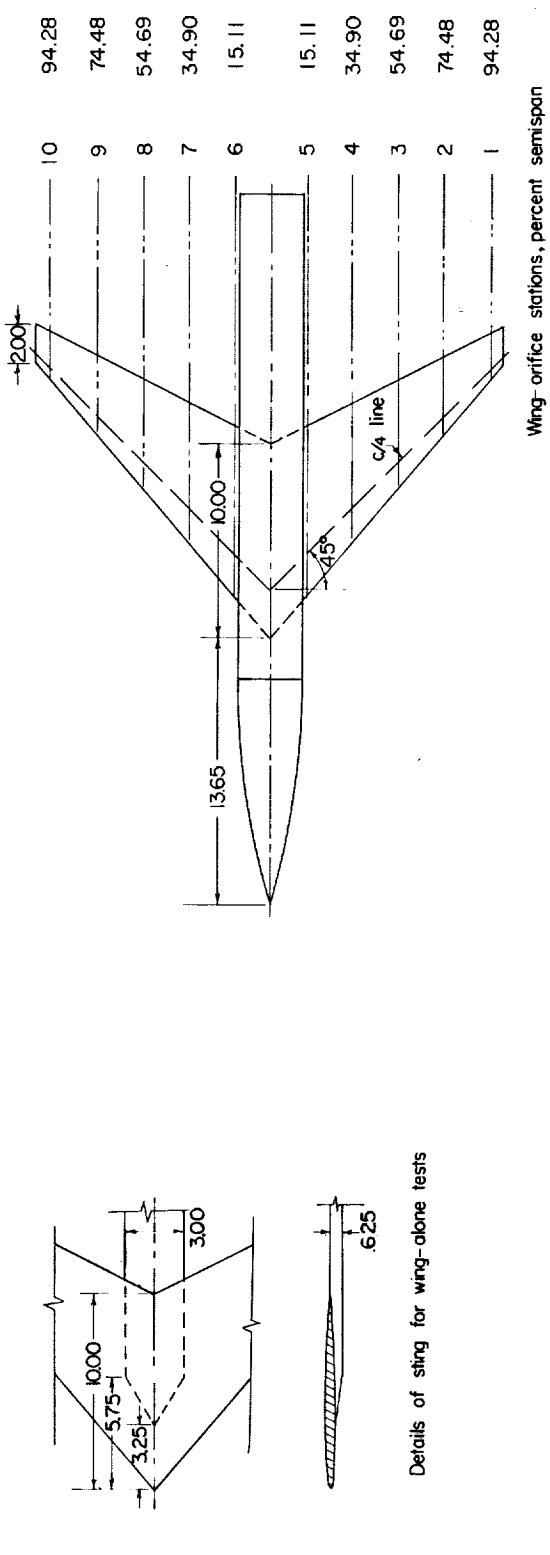

Figure 2.- Details of model configuration. All dimensions are in inches unless otherwise noted.

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